



APOLLO TRAINING

BLOCK II CSM
PROPULSION SUBSYSTEM
CSM 106 AND SUBS

10 MARCH 1969

FOR TRAINING PURPOSES ONLY



The following corrections are to be made to the CSM Propulsion Subsystem handout package dated 10 March 1969.

Page 4 SM-2A-1353N

Change note as follows: from ** CSM 111 and subs: to ** CSM 108 and subs.

Add note as follows to SP0061T and SP0062T,

*** Caution and warning light on MDC-2 will illuminate on CSM 106 and 107 but is ignored. Light on MDC-2 is disconnected and covered with a decal on CSM 108 and subs.

Page 5 SM-2A-2028

Delete MDC-2 and from SP0061T and SP0062T.

Page 7 SM-2A-1243C

Disconnect lead to SPS PUSNSR caution and warning light on MDC-2 thus rendering the light inoperative from the PUG system on CSM 106 and subs.

Page 10 P-5096-1A

Delete page in its entirety due to SPS PUSNSR caution and warning light being disconnected (page 7).

Page 68

Change SPS TEMP FUEL ENGINE FEEDLINE (SP0048T) and TEMP OX ENGINE FEEDLINE (SP0049T) operating range as follows: +45 to +75°F. Redline mark at +27 and +100°F for SP0048T.

Delete all presently stated operating range voltages and temperatures for SP0049T and add as follows:

L 1250 vdc = 45°F

L 95 vdc = 75°F

Delete, parenthesis and CH RCS 1 and parenthesis and CH RCS 2

Add 1 after CROOOLP HEL TANK PRESS
 1 after CROO3T HEL TANK TEMP
 2 after CROO2T HEL TANK PRESS
 2 after CROO4T HEL TANK TEMP

P-0043

CSM Propulsion Power Distribution CSM 106 and subs: Delete Panel 2 SPS PUSNR caution and warning light. Add statement to SPO61T and SPO62T, light on Panel 2 will illuminate on CSM 106 and 107 but is ignored, on CSM 108 and subs light is disconnected and covered with a decal.

P-0044

Control and display panel MDC-2. The SPS PUSNR Light is covered with a decal on CSM 106 and subs.

Questions Relative To The Contents of This Document

Should Be Directed To:

W. F. Green

F. M. Matthews

NR SD

**Downey, California
Extension 4325, or 6**

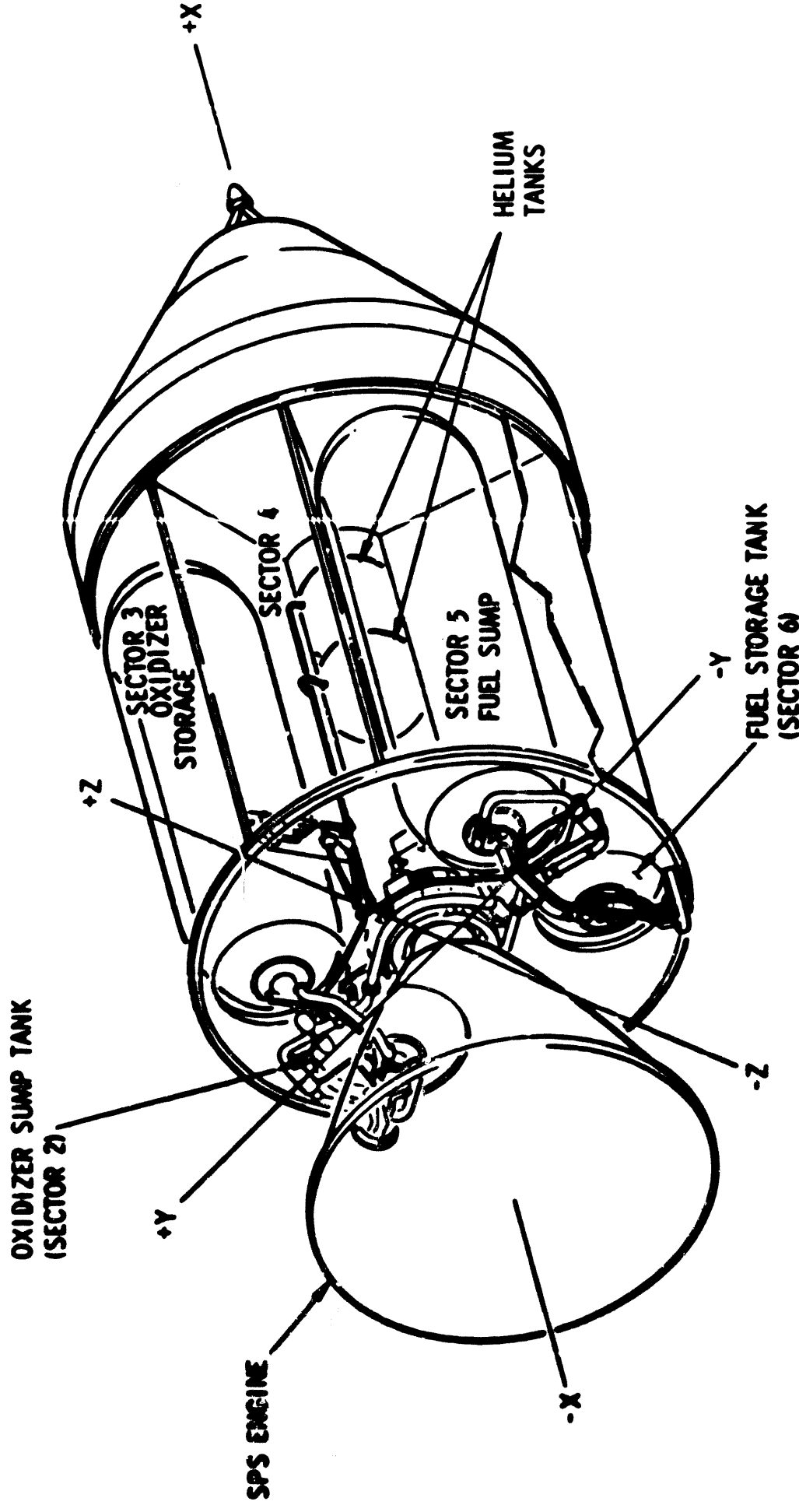
TABLE OF CONTENTS

	Page
Service Propulsion Subsystem	1
CSM Reaction Control System Interface Control	28
Service Module Reaction Control Subsystem	29
Command Module Reaction Control Subsystem	44
SPS Data	55
SPS Measurement	62
SMRCS Data	71
SMRCS Measurements	74
CMRCS Data	79
CMRCS Measurements	81
List of Abbreviations	85
Service Module Drawing Systems	87
Command Module Drawing Systems	88
Power Sources (MDC-229 to MDC-8)	89
Control and Display Panel	92
Propulsion Power Distribution	93

The following pages contained within this handout package have been revised or added since the 12 December 1968 Block II CSM Propulsion Subsystem Handout Package.

<u>Page Number</u>	<u>Handout Number</u>	<u>Page Number</u>	<u>Handout Number</u>
2	P-5108F	75	P-6060G
3	P-5003E	76	P-6059F
4	SM-2A-1353N	77	SMRCS Measurements
5	SM-2A-2028	78	SMRCS Measurements
6	P-5048B	79	CMRCS Data
7	SM-2A-1243G	80	CMRCS Data
12	P-5020E	81	P-7027G
23	SM-2A-1352J	82	P-7030
26	P-5100N	83	CMRCS Measurements
27	P-5117	84	CMRCS Measurements
30	P-6003A	89	P-0039F
47	SM-2A-1241R	90	P-0040F
54	SM-2A-1240T	92	Control and Display Panel
55	SPS Data	93	Propulsion Power Distribution
56	SPS Data		
57	SPS Data		
58	SPS Data		
62	P-5099C		
63	P-5101B		
64	P-5098C		
65	P-5007E		
68	SPS Measurements		
69	SPS Measurements		
70	SPS Measurements		
71	SMRCS Data		
72	SMRCS Data		
73	SMRCS Data		
74	P-6061C		

SERVICE PROPULSION SYSTEM



TYPE OF SYSTEM

NON-THROTTLEABLE, PRESSURE FEED SYSTEM, HYPERGOLIC PROPELLANTS AND ABLATIVE THRUST CHAMBER WITH A RADIANT COOLED NOZZLE EXTENSION.

PURPOSE:

PROVIDE THRUST FOR X-AXIS VELOCITY CHANGES AFTER SEPARATION FROM THE S-IVB BOOSTER UP UNTIL CM/SM SEPARATION. IN ADDITION SUPPORTS THE SPS ABORT CAPABILITY DURING THE BOOST PHASE AFTER THE LAUNCH ESCAPE TOWER HAS BEEN JETTISONED.

SYSTEM DESCRIPTION:

ONE SYSTEM CONSISTING OF A PROPELLANT FEED SYSTEM AND ONE ENGINE

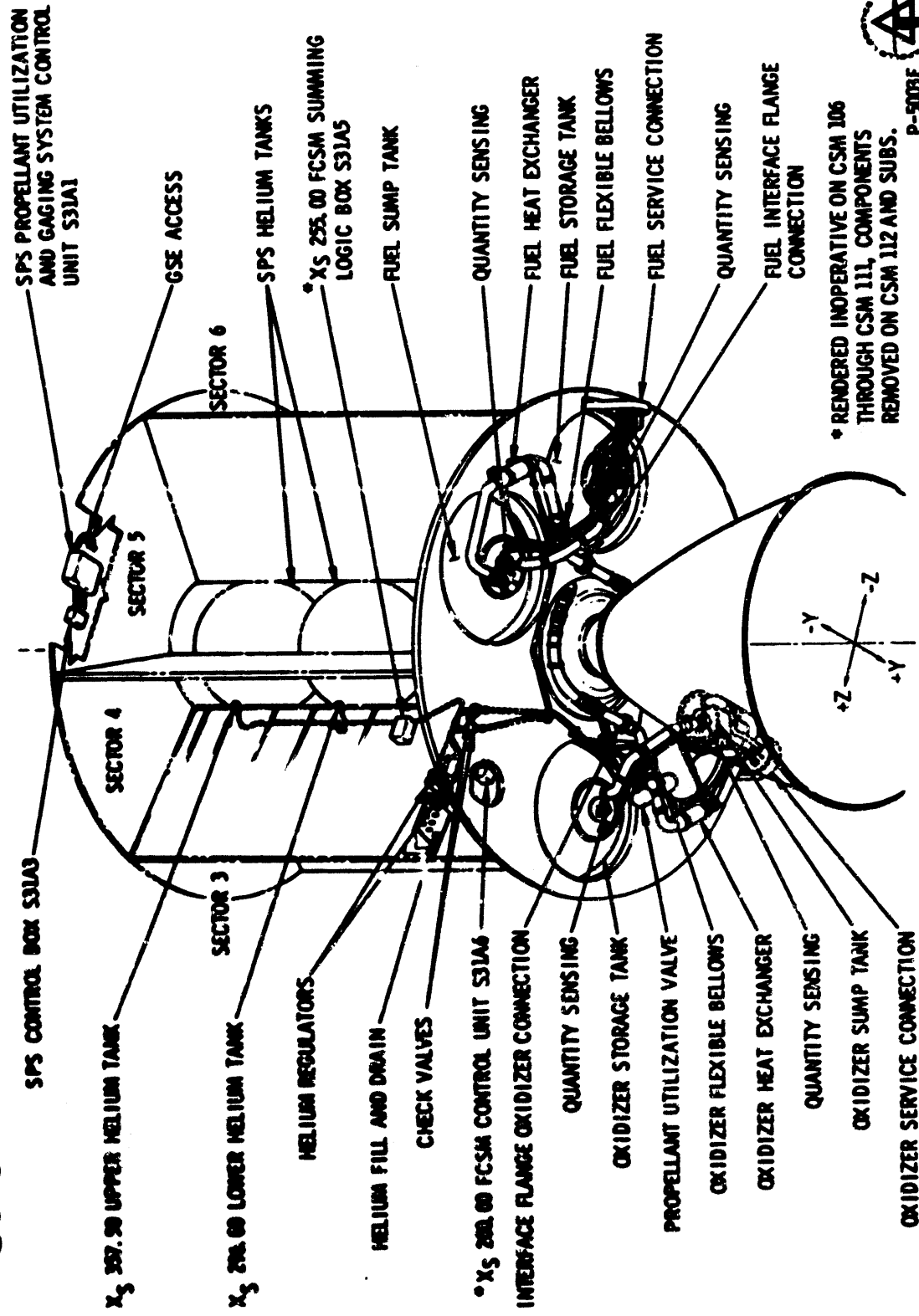
PROPELLANTS - INHIBITED NITROGEN TETROXIDE AND BLENDED HYDRAZINE (APPROX 50% UDMH & 50% HYDRAZINE)

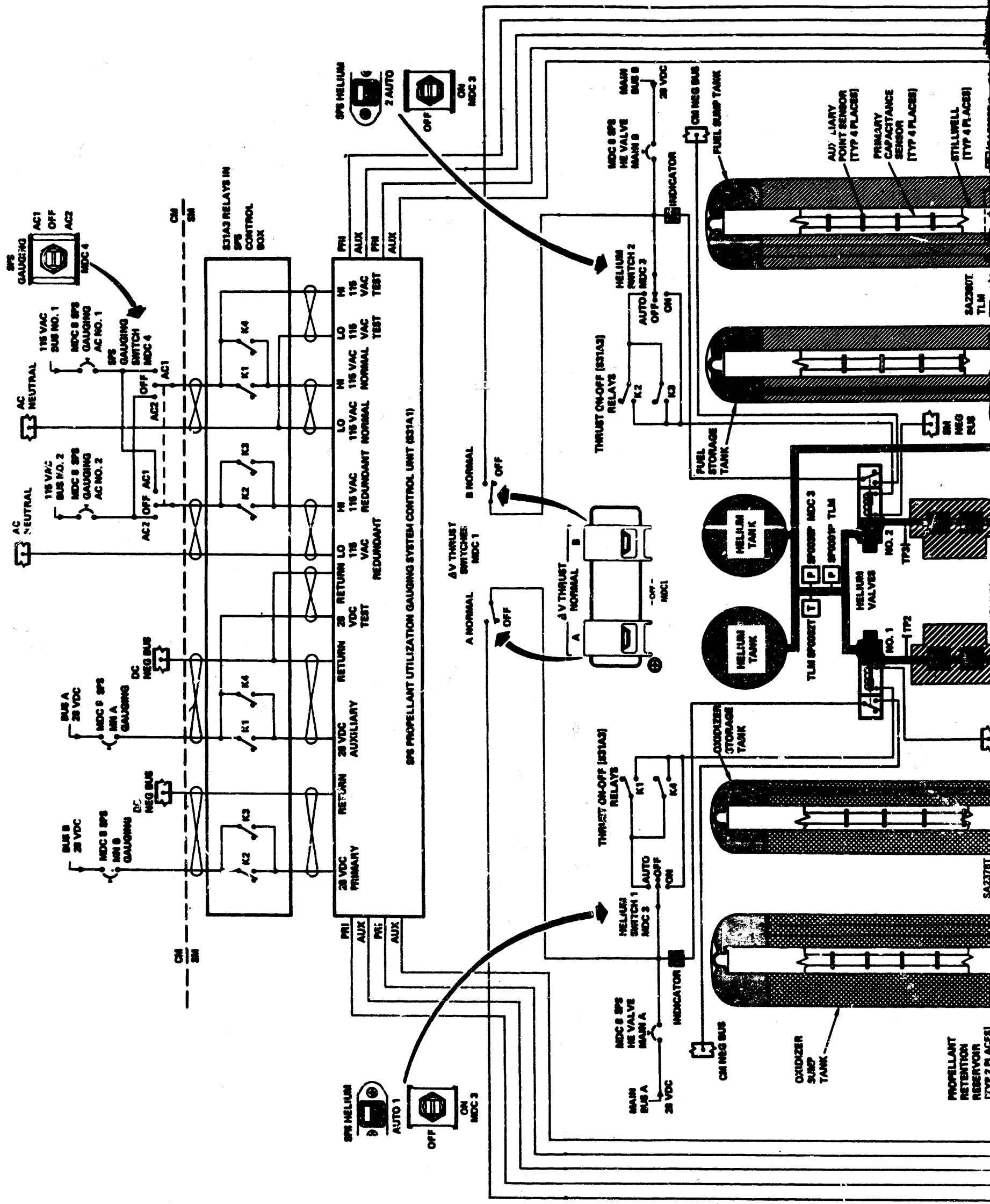
SYSTEM CONTROLLED BY CMC OR SCS WITH A DIRECT CONTROL BACK-UP CAPABILITY

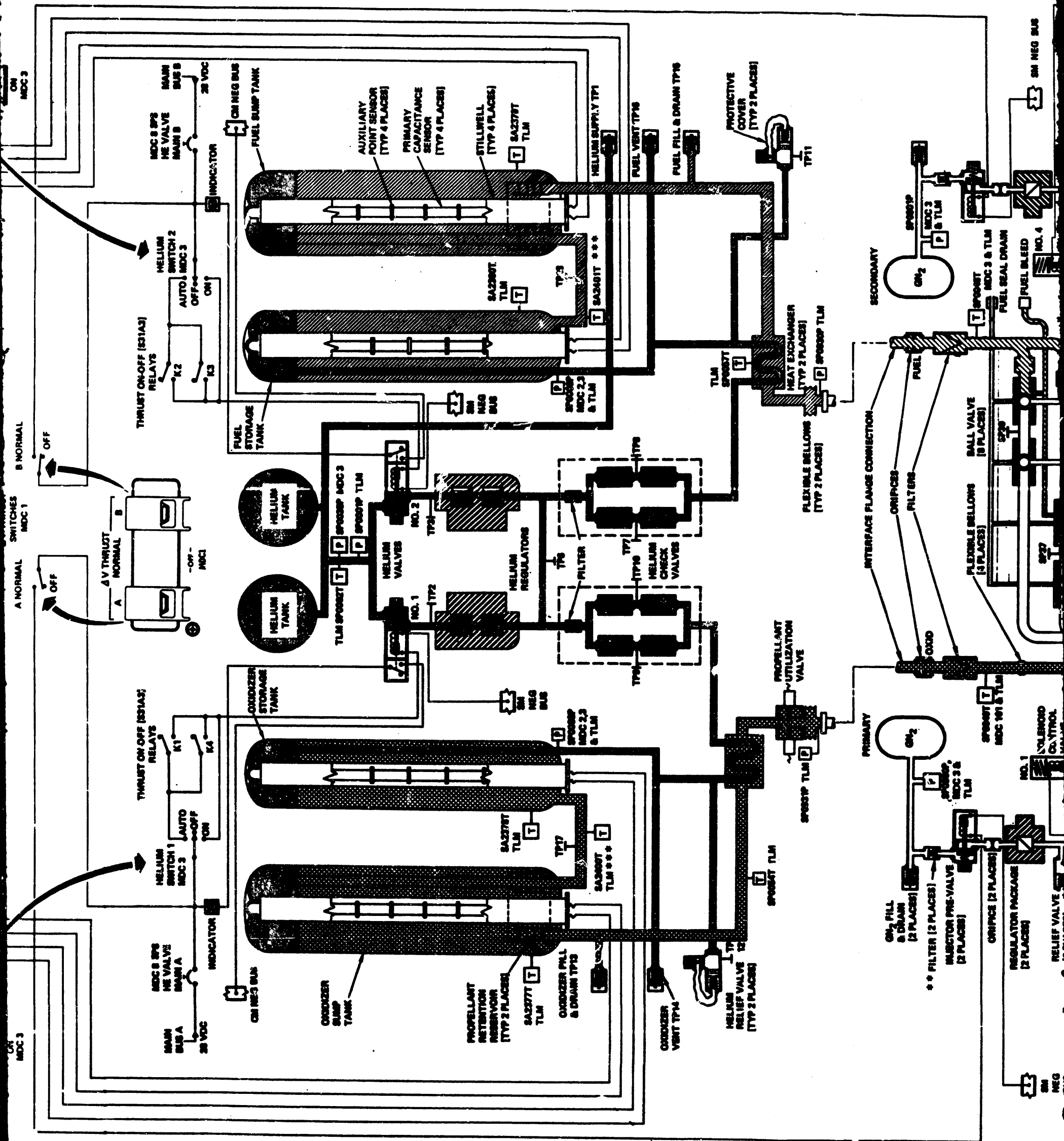
[illegible]

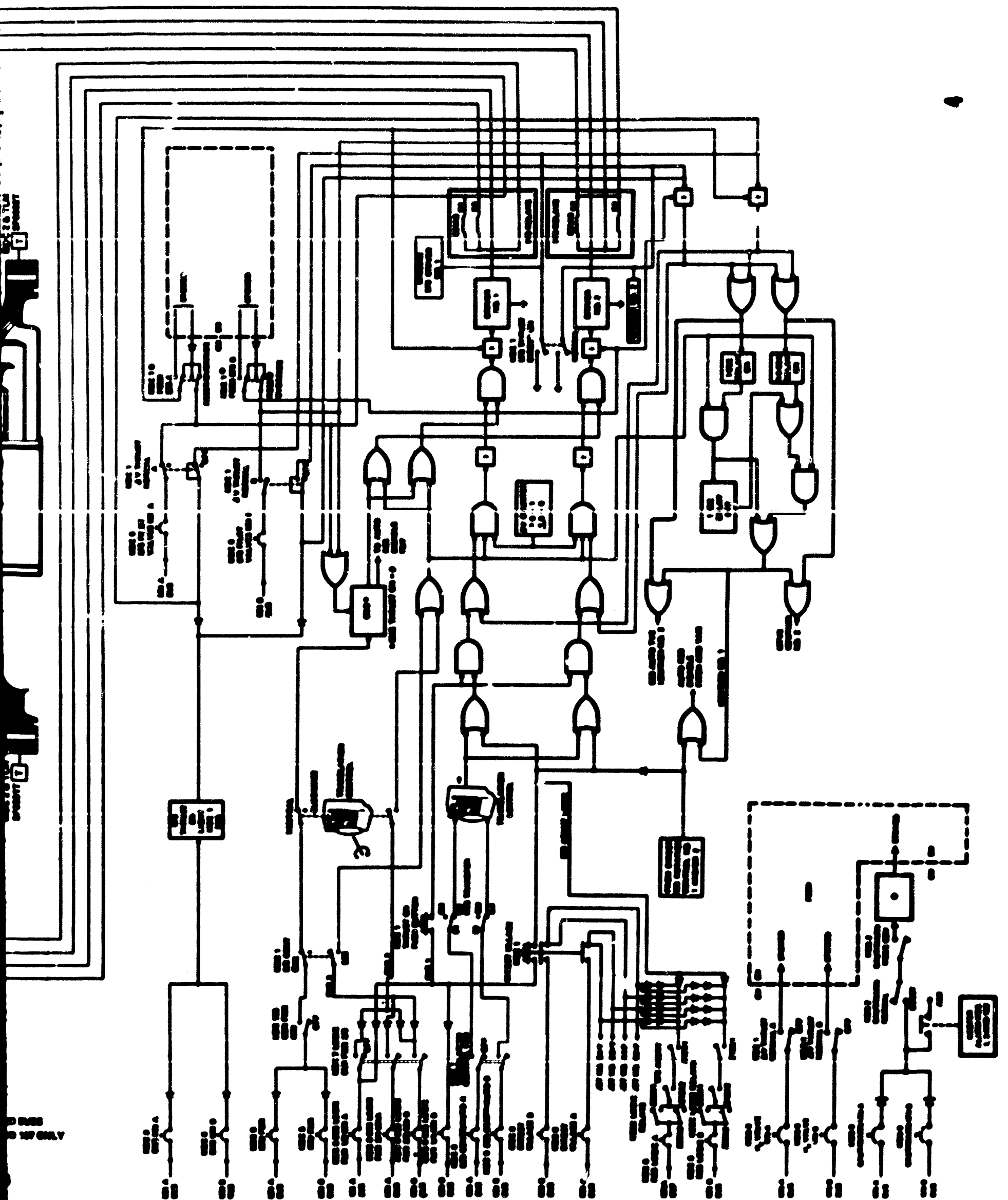
NOTE:
FLIGHT COMBUSTION STABILITY MONITOR SWITCHES
SPS A AND SPS B ARE POSITIONED AND LOCKED
TO RESET / OVERRIDE RENDERING FCSM INOPERATIVE.
CSM 112 AND SUBS, FCSM SPS A AND SPS B
SWITCHES ARE REMOVED.

SPS COMPONENT LOCATIONS AFT VIEW

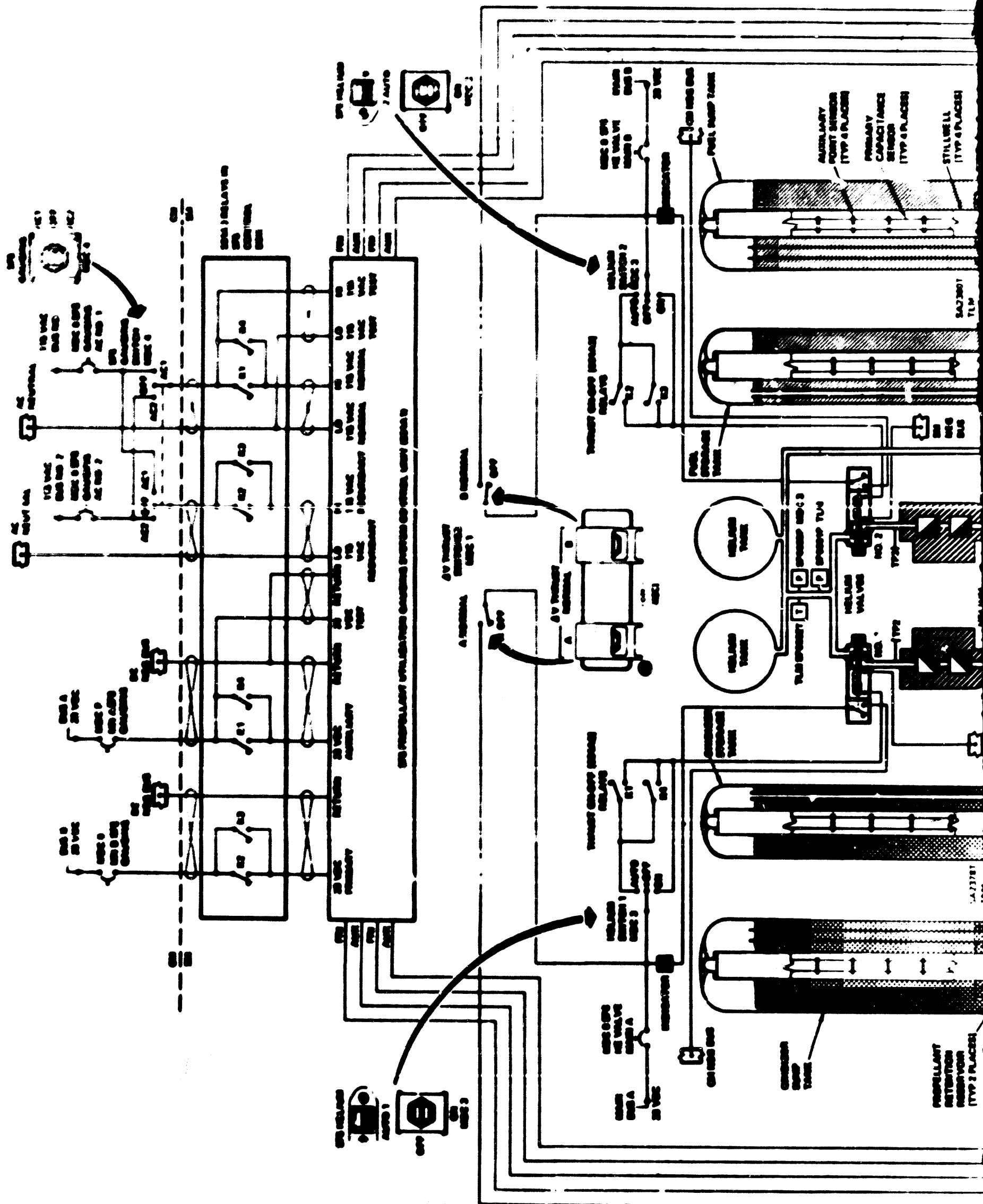


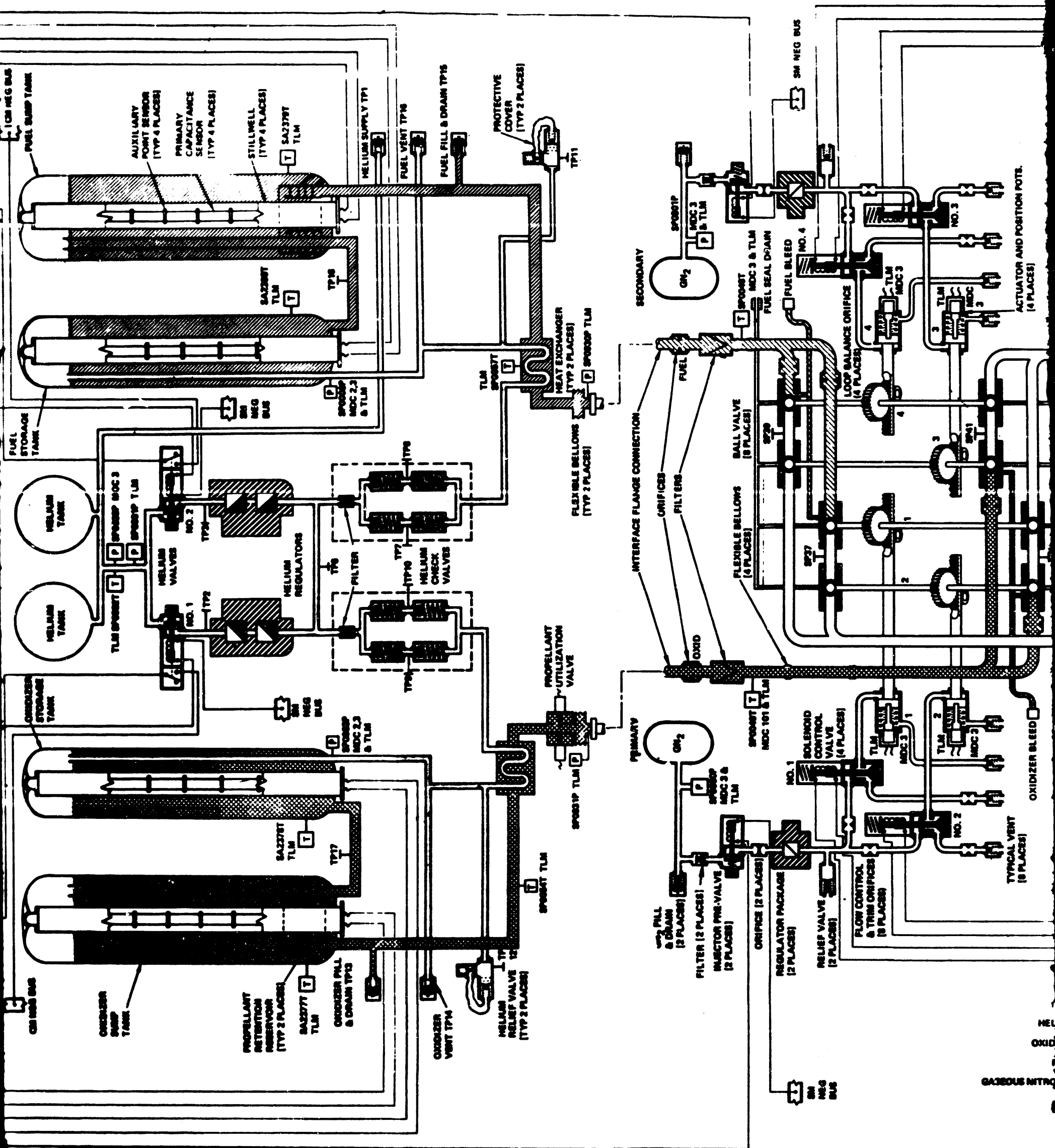






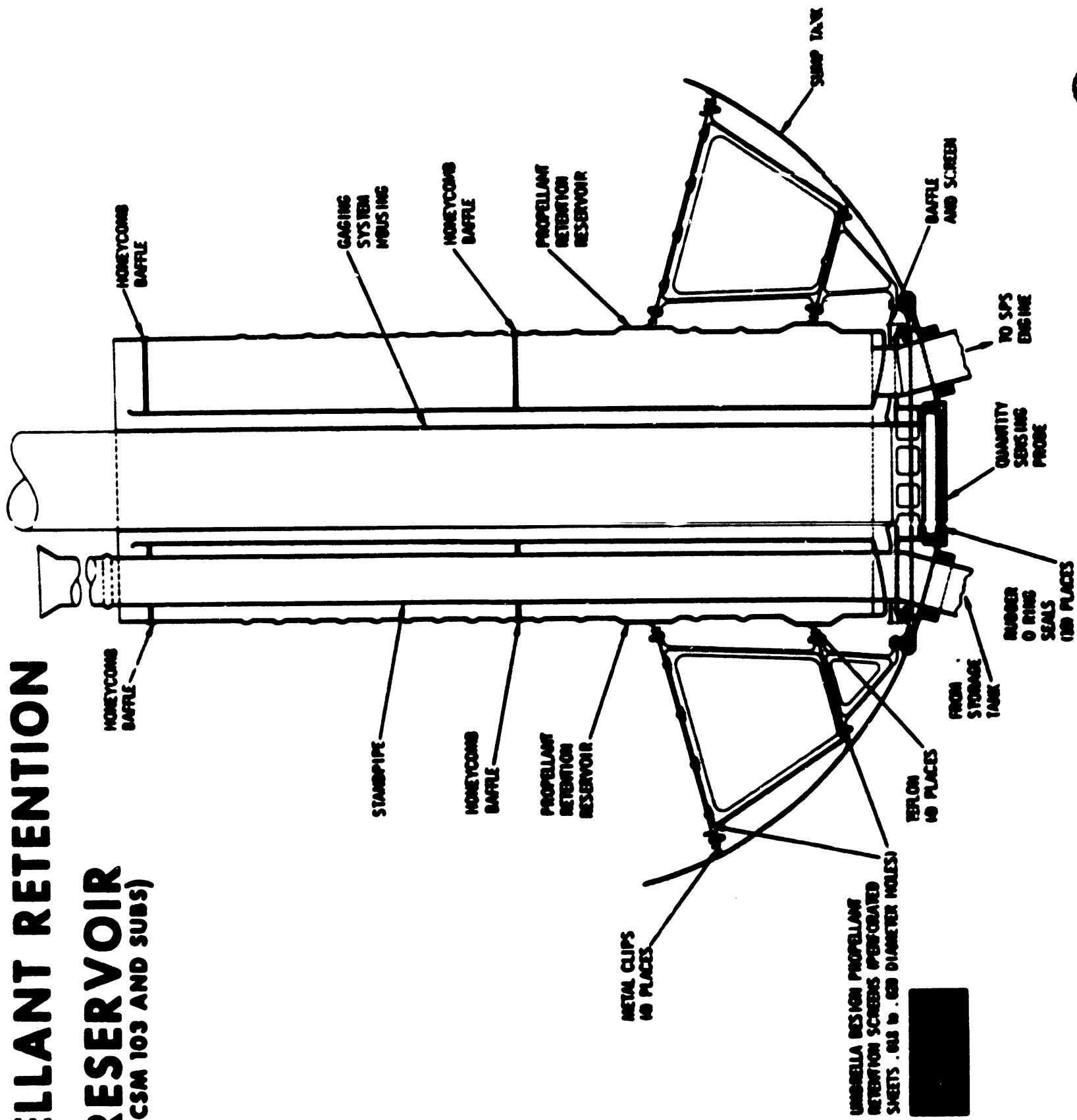
JPS FUNCTIONAL FLOW CSM 106 THROUGH CSM 111

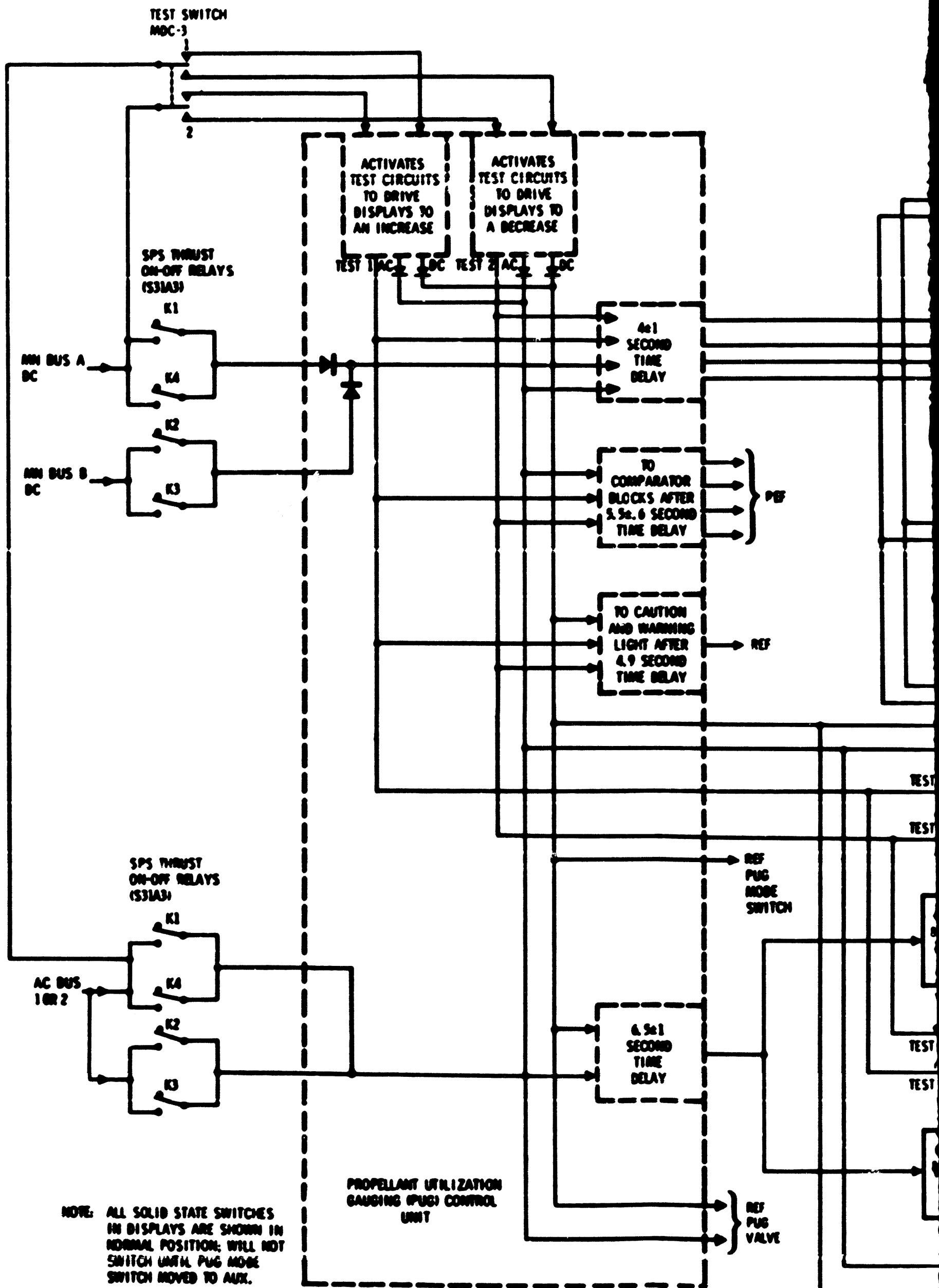


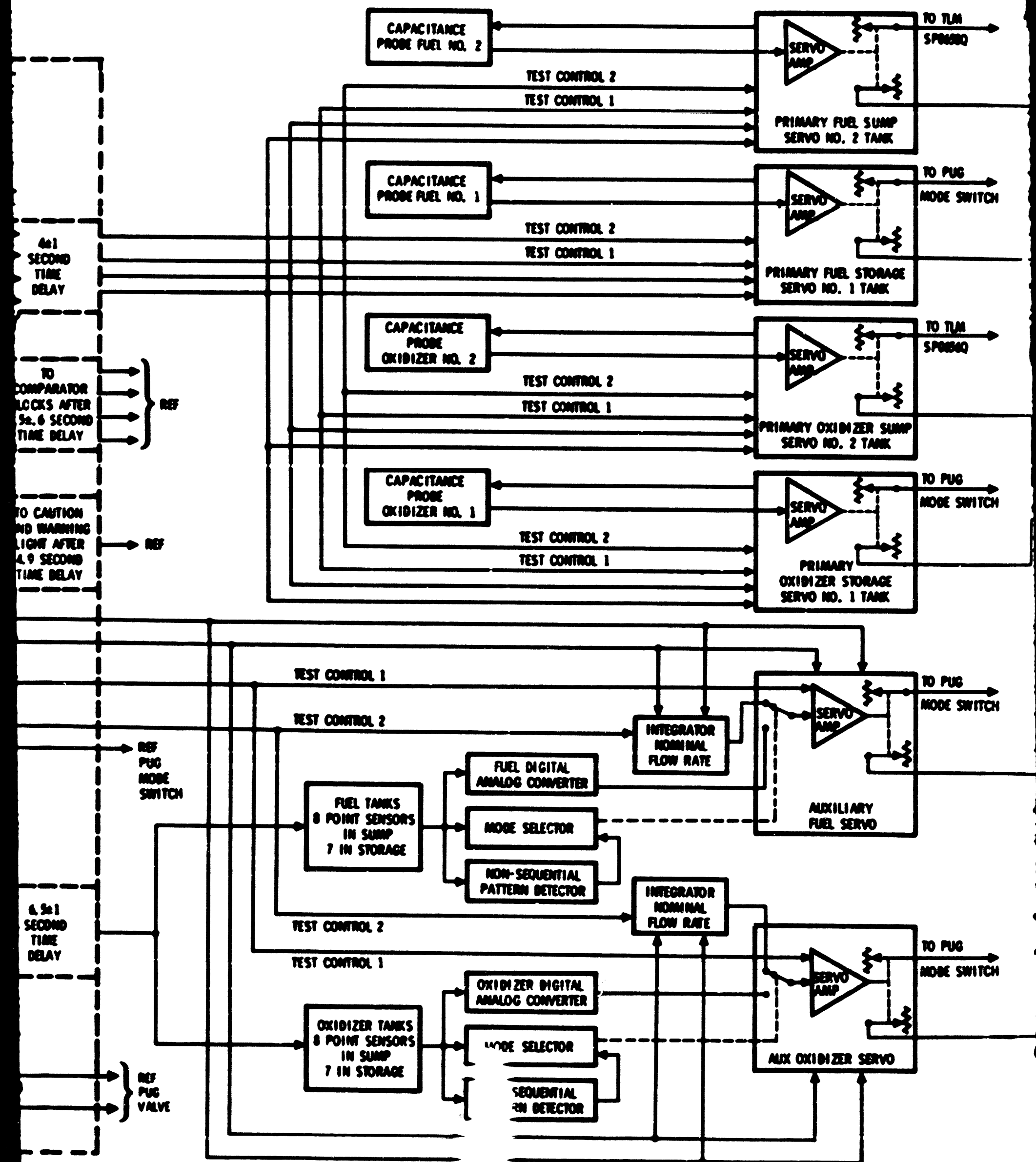


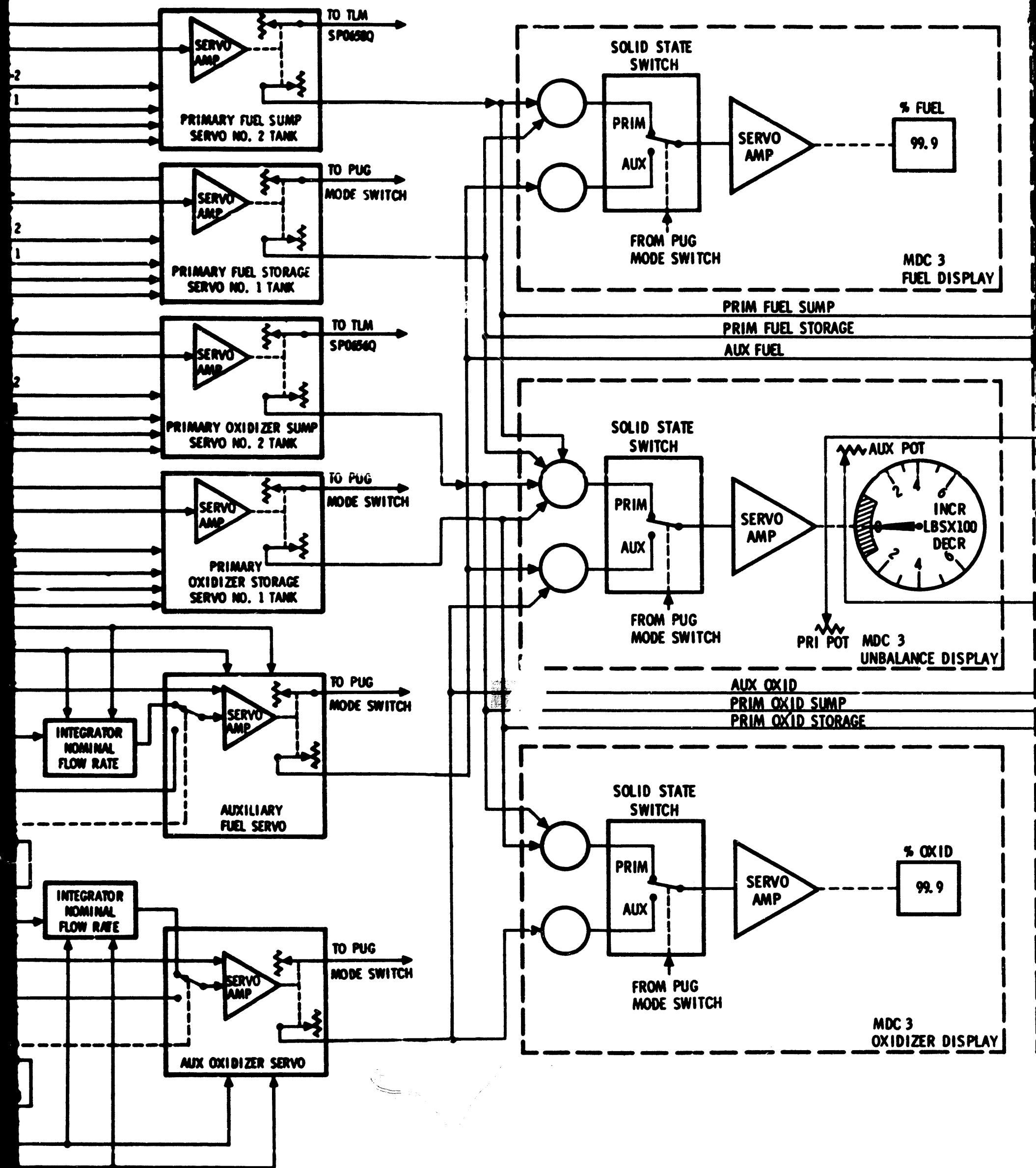
SPS PROPELLANT RETENTION

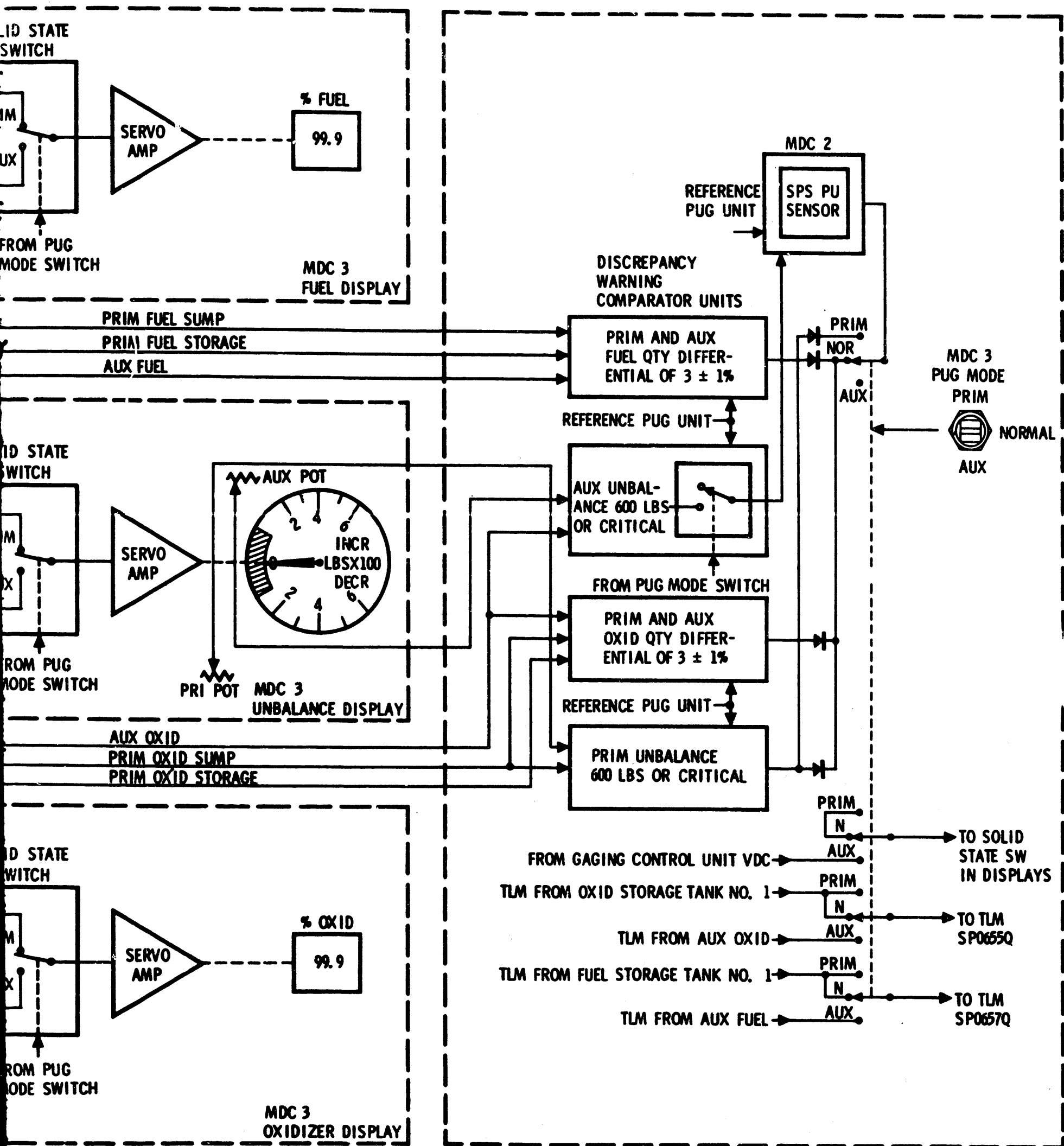
RESERVOIR (CSM 103 AND SUBS)







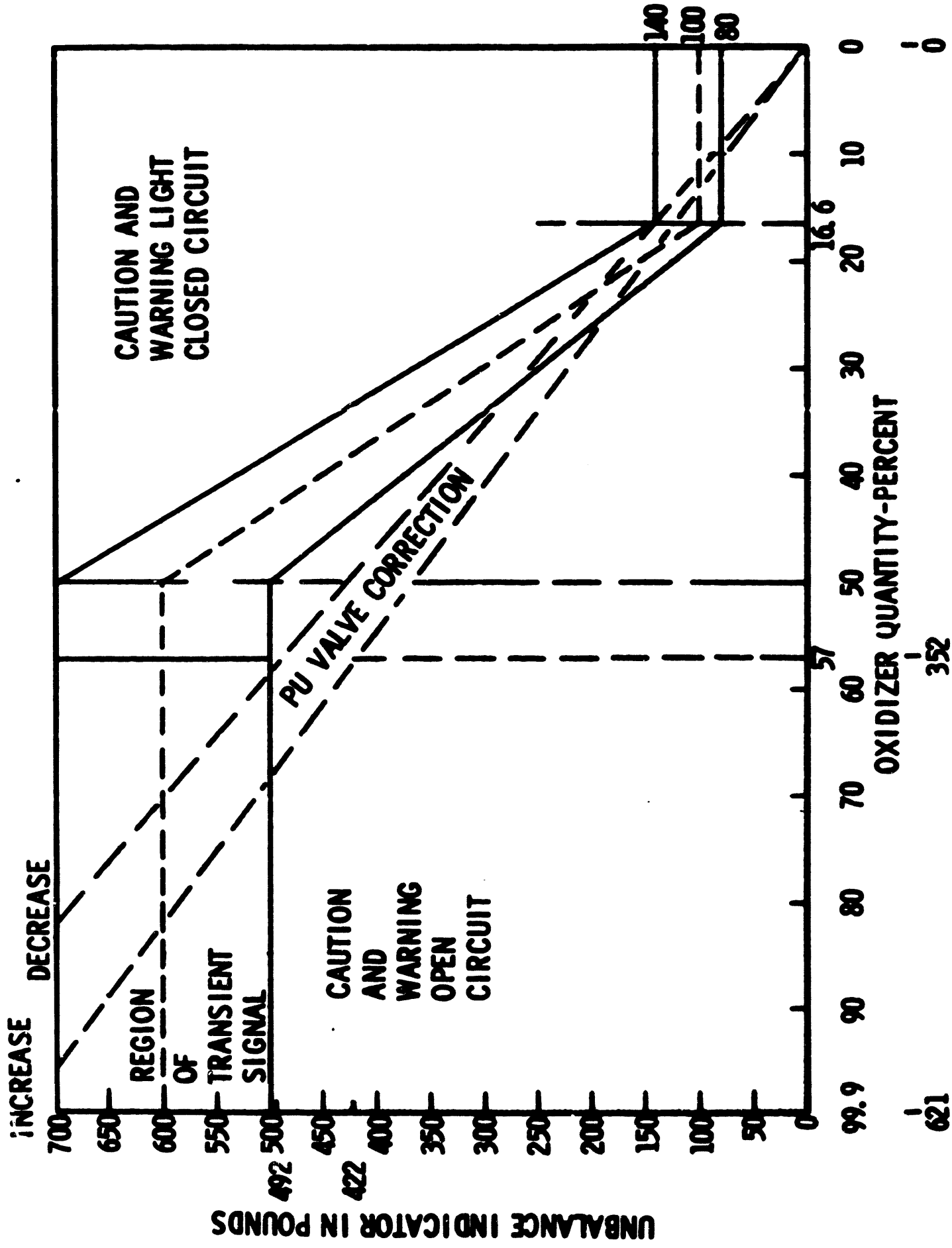




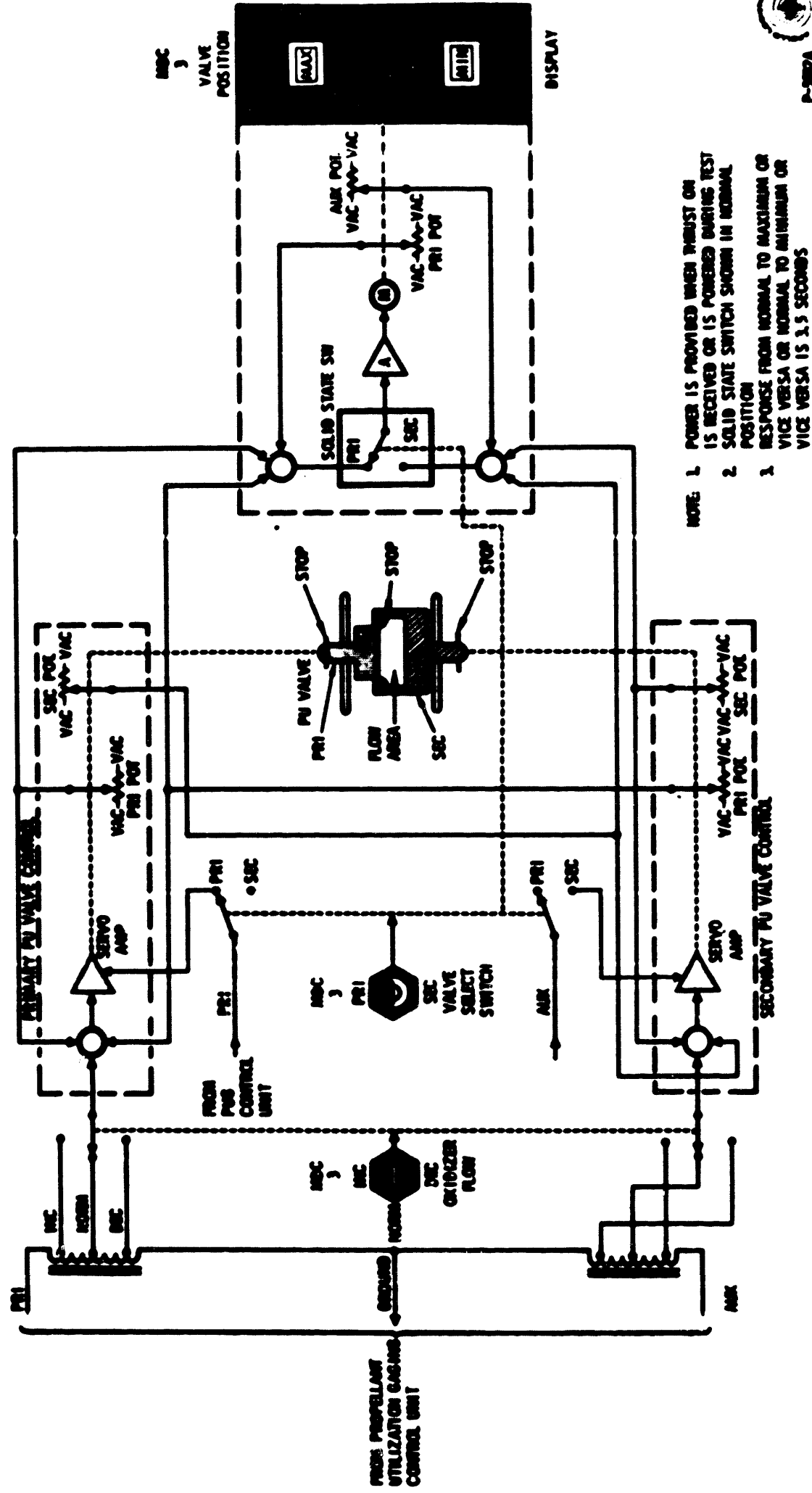
OXIDIZER POINT SENSOR LOCATIONS					
NUMBER	WEIGHT POUNDS		PERCENT MAXIMUM		INCHES
	TEMP	70°F	70°F		
	PRESS.	176.60 PSIA	176.60 PSIA		
15	1,586.01	OXIDIZER SUMP TANK NO. 2 OXIDIZER STORAGE TANK NO. 1	6.5	25.36	
14	3,172.02		13.0	41.34	
13	4,758.03		19.5	57.25	
12	6,344.04		26.0	72.07	
11	7,930.05		32.5	86.89	
10	9,516.06		39.0	101.72	
9	11,102.07		45.5	116.54	
8	12,688.08		52.0	131.60	
7	14,274.09		57.8	10.50	
6	15,860.10		64.3	30.57	
5	17,446.11		70.8	49.55	
4	19,032.13		77.3	68.54	
3	20,618.14		83.8	87.52	
2	22,204.15		90.3	106.51	
1	23,790.16		96.8	125.49	

FUEL POINT SENSOR LOCATIONS				
NUMBER	WEIGHT POUNDS		PERCENT MAXIMUM	
	70°F		70°F	
	PRESS. 176.60 PSIA		176.60 PSIA	
15	992.08	6.5	FUEL SUMP TANK NO. 2	25.36
14	1984.16	13.0		41.34
13	2976.24	19.5		57.25
12	3968.32	26.0		72.07
11	4960.40	32.5		86.89
10	5952.48	39.0		101.72
9	6944.56	45.5		116.54
8	7936.64	52.0		131.60
7	8928.72	57.8	FUEL STORAGE TANK NO. 1	10.48
6	9920.80	64.3		30.55
5	10912.88	70.8		49.54
4	11904.96	77.3		68.52
3	12897.04	83.8		87.51
2	13889.12	90.3		106.49
1	14881.20	96.8		125.48

CAUTION AND WARNING CRITICAL PROPELLANT UNBALANCE COMPARATOR

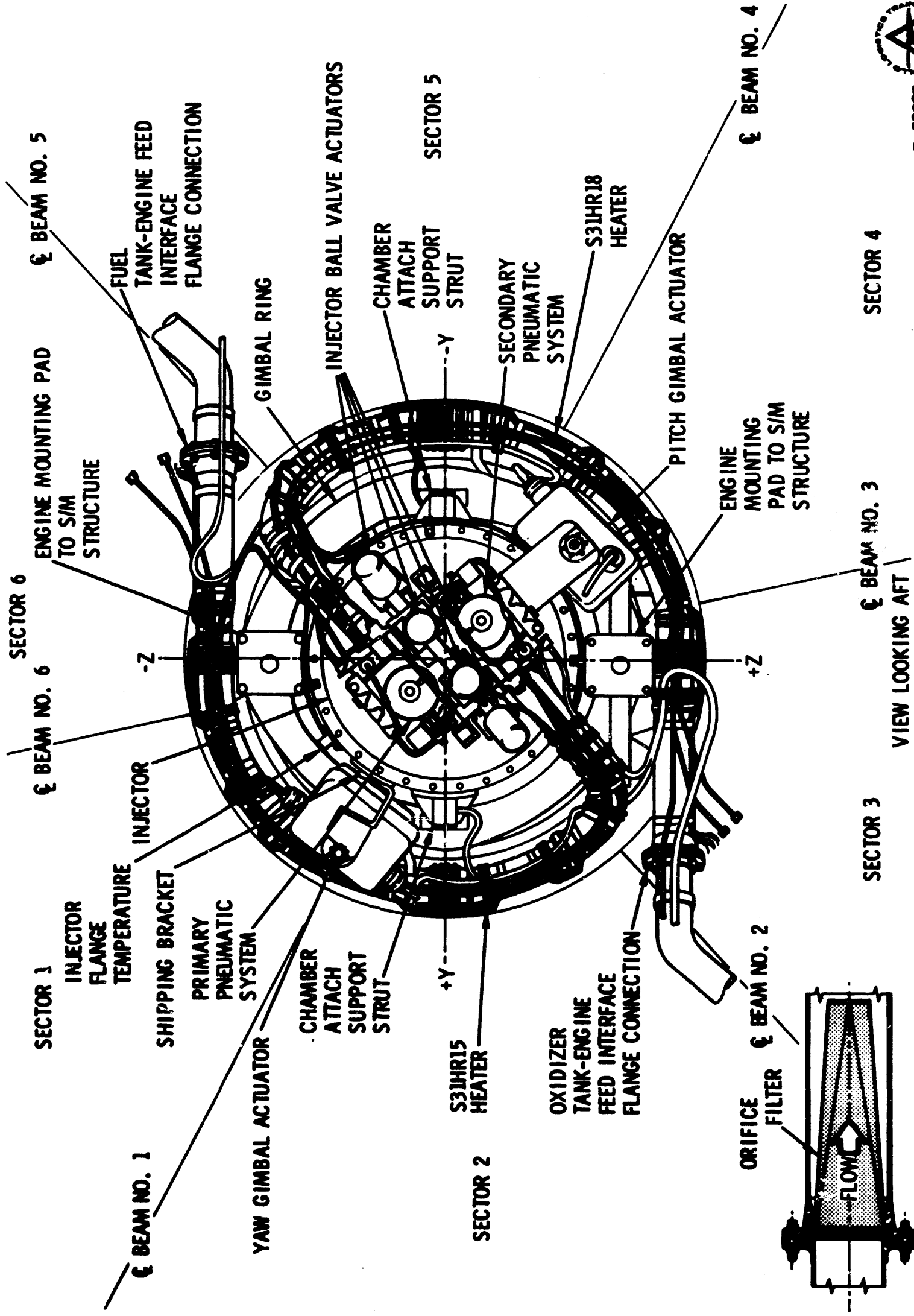


PROPELLANT UTILIZATION VALVE CONTROL AND FLAG DISPLAY

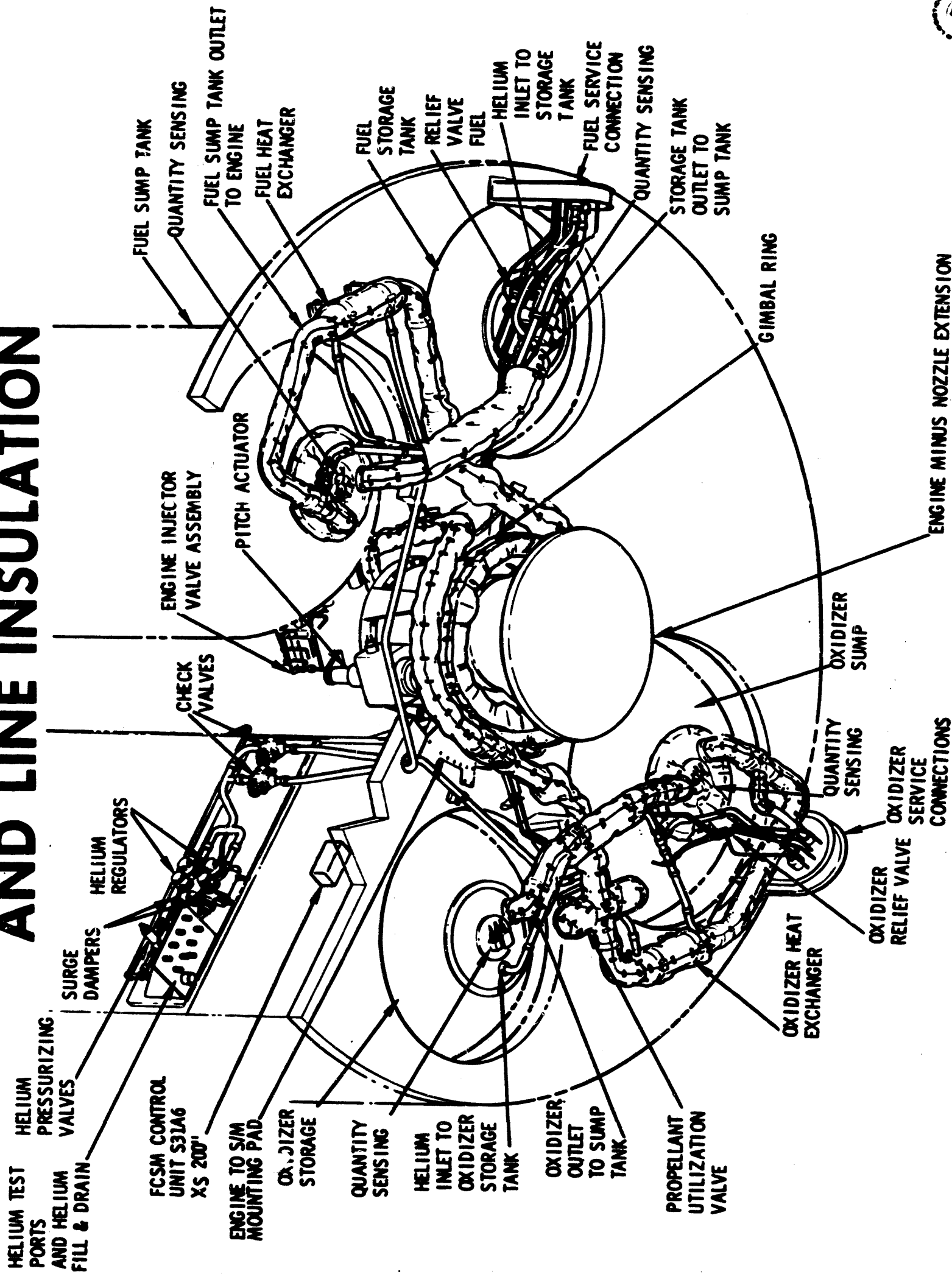


TOP VIEW OF SPS ENGINE

(CSM 106 AND SUBS)



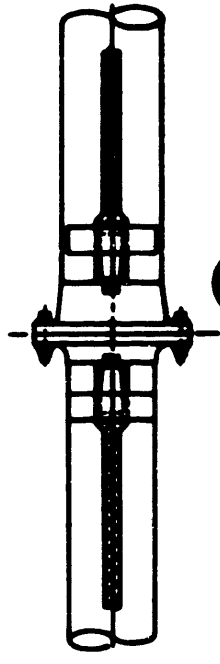
SPS COMPONENT LOCATIONS AFT VIEW AND LINE INSULATION



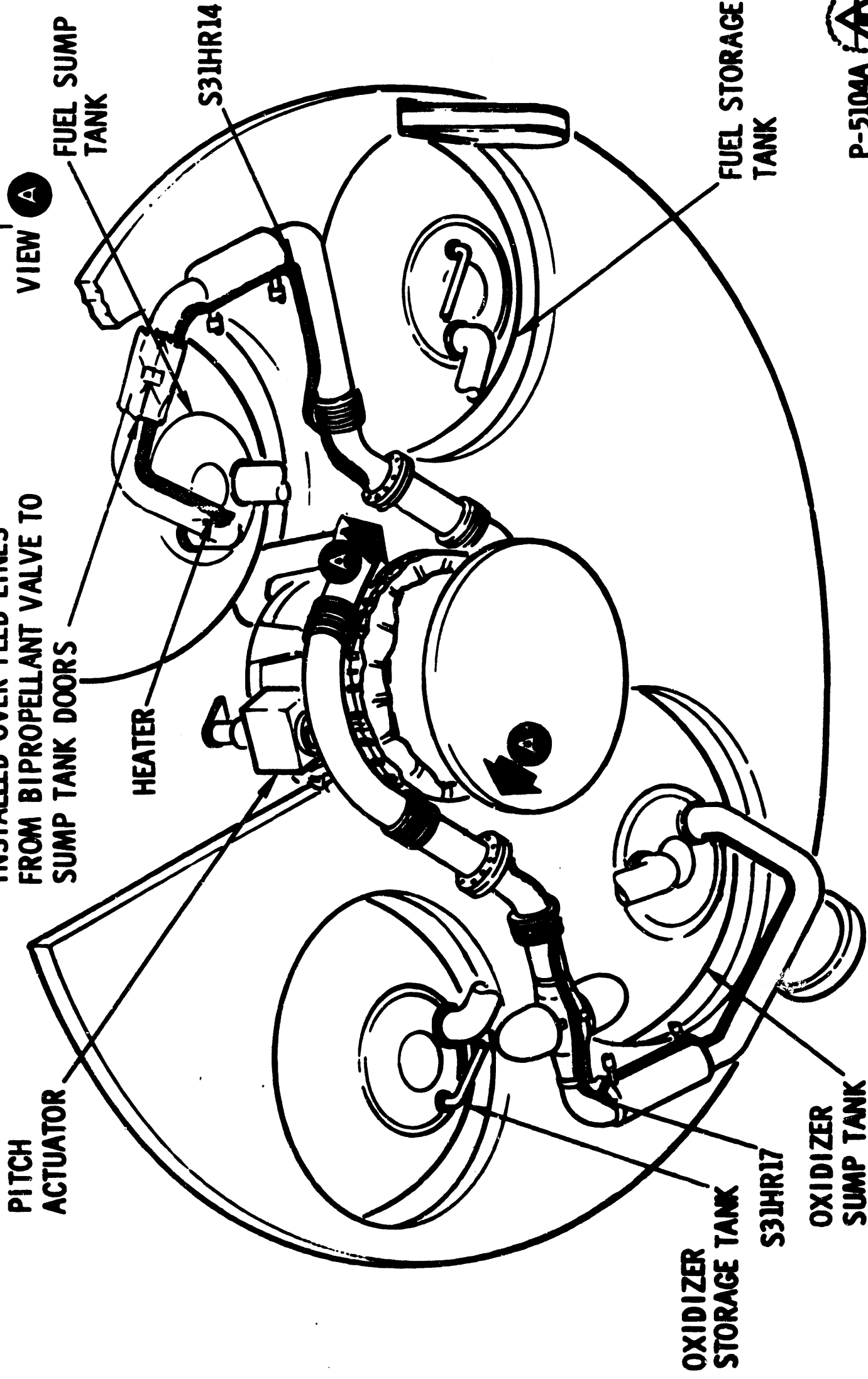
SPS HEATER INSTALLATION, TANK FEED LINES

CSM 106 & SUBS

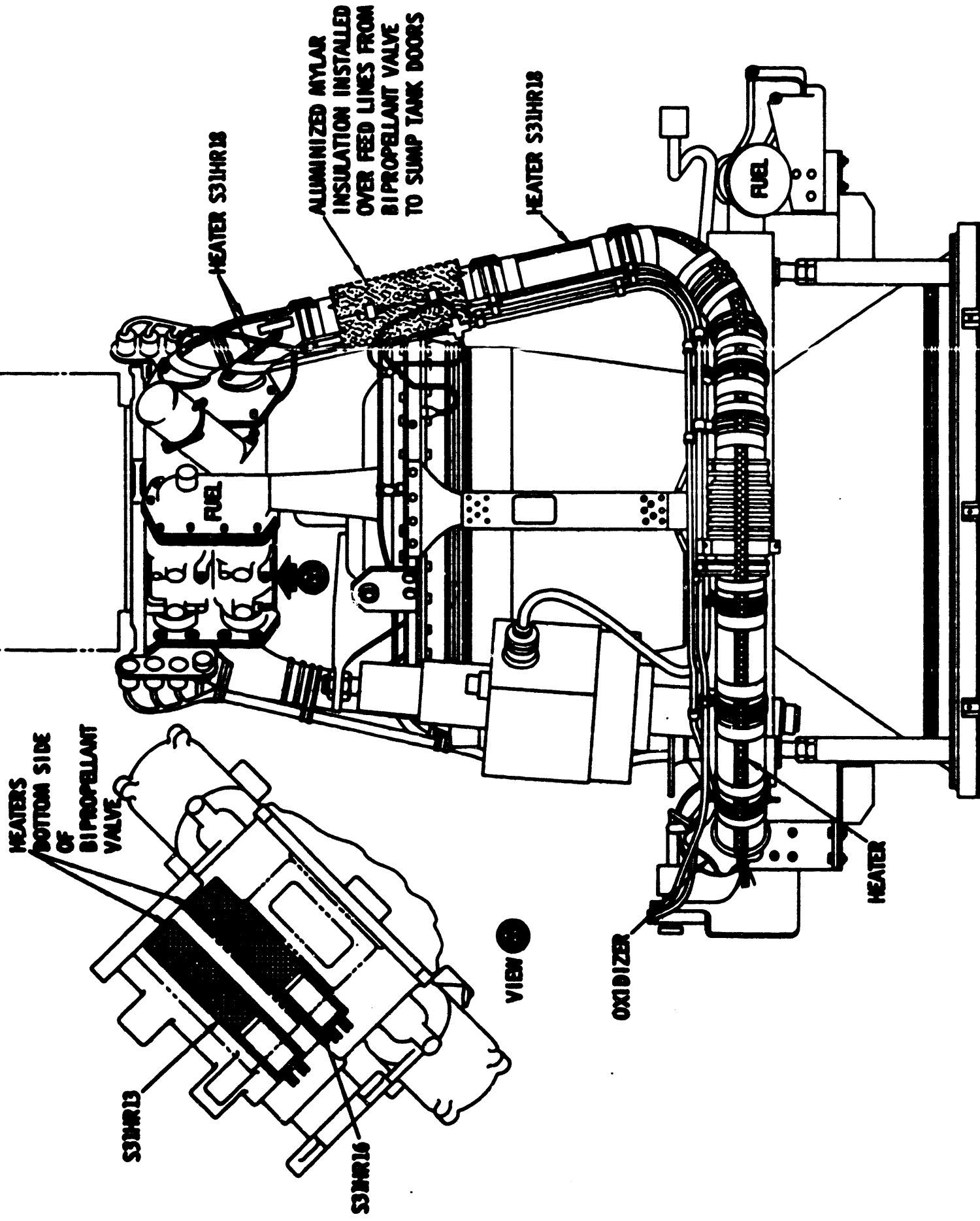
ALUMINIZED MYLAR INSULATION,
INSTALLED OVER FEED LINES
FROM BIPROPELLANT VALVE TO
SUMP TANK DOORS



VIEW A

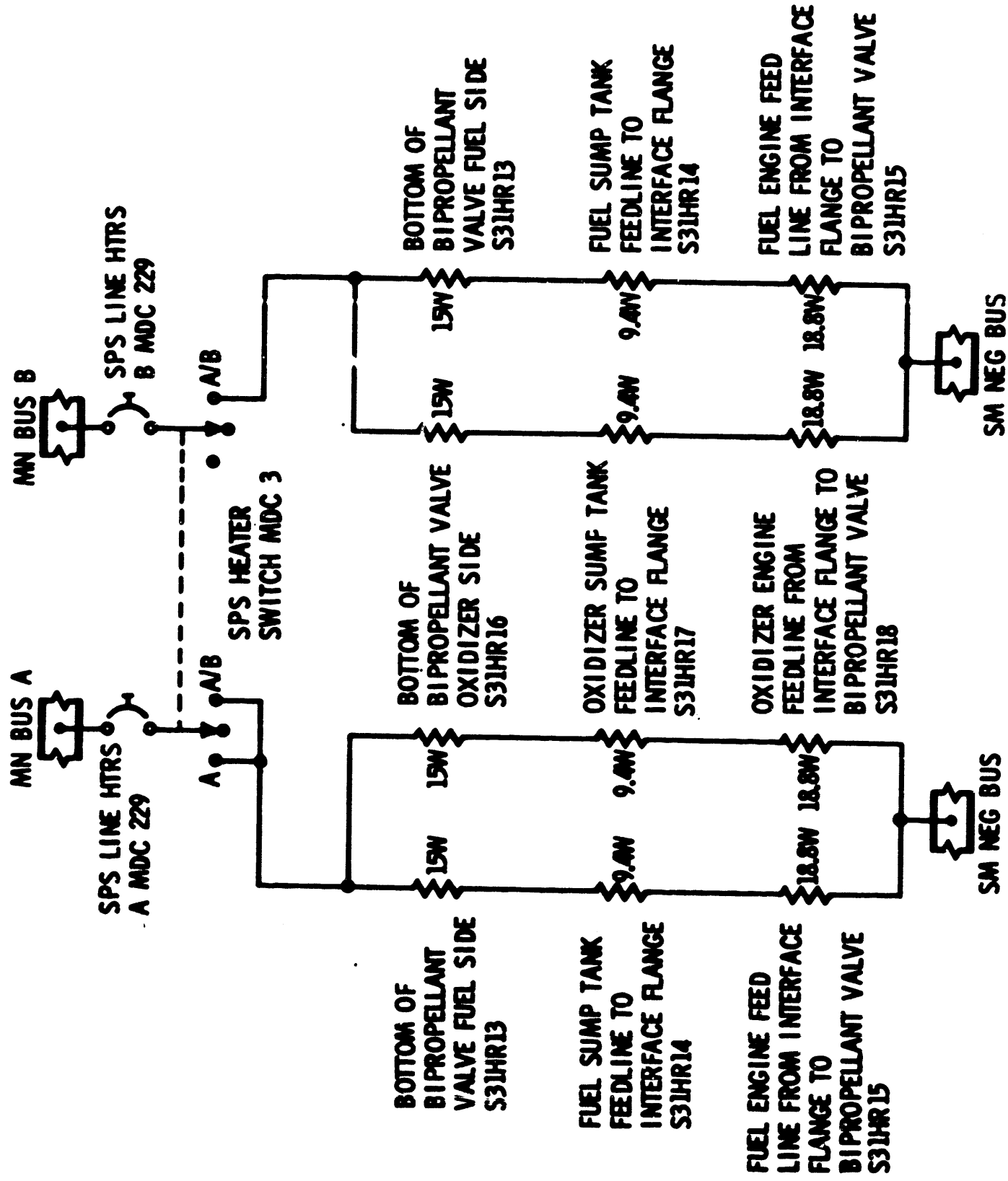


SPS HEATER INSTALLATION ENGINE FEED LINES CSM 106 & SUBS

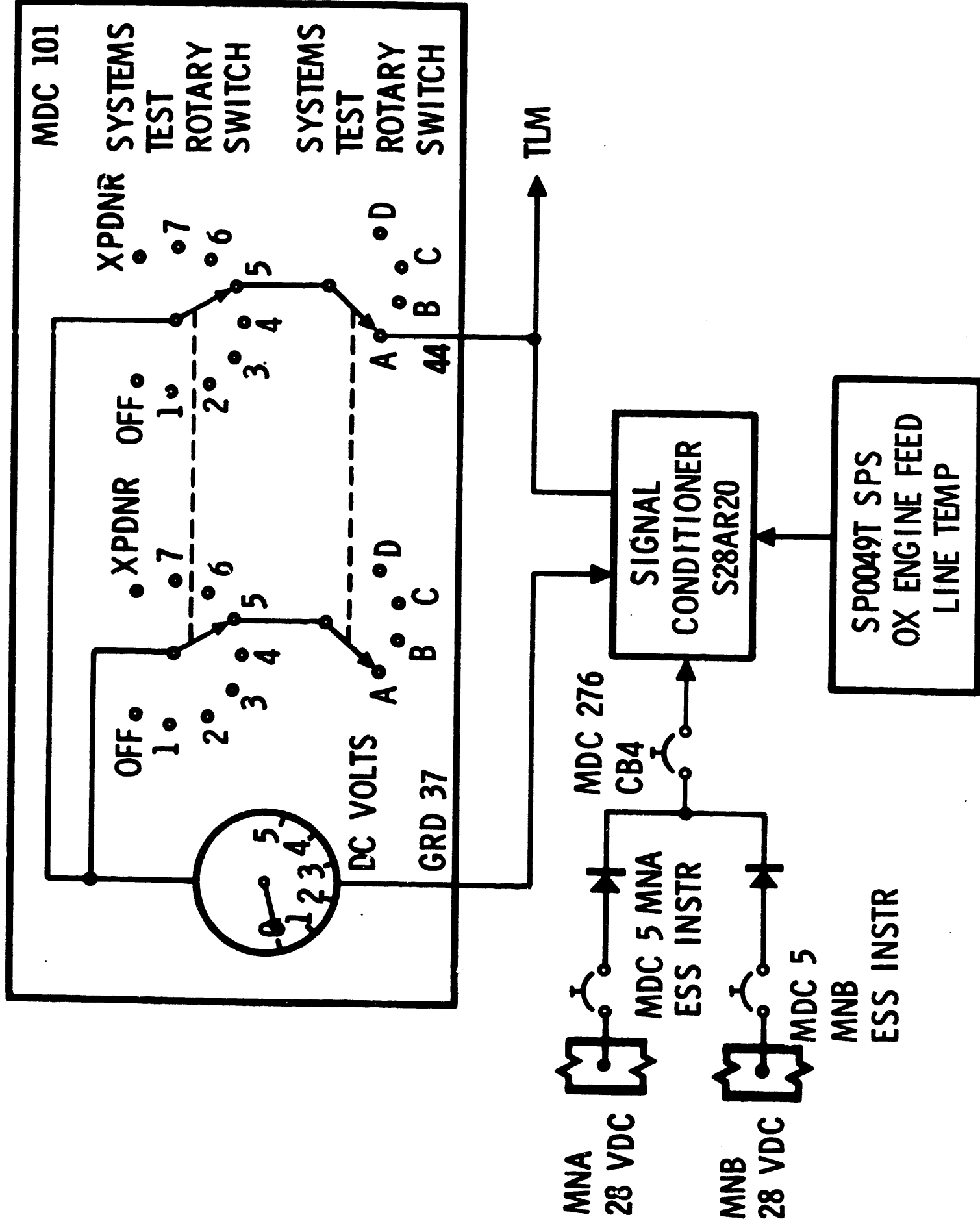


SPS ENGINE HEATER SCHEMATIC

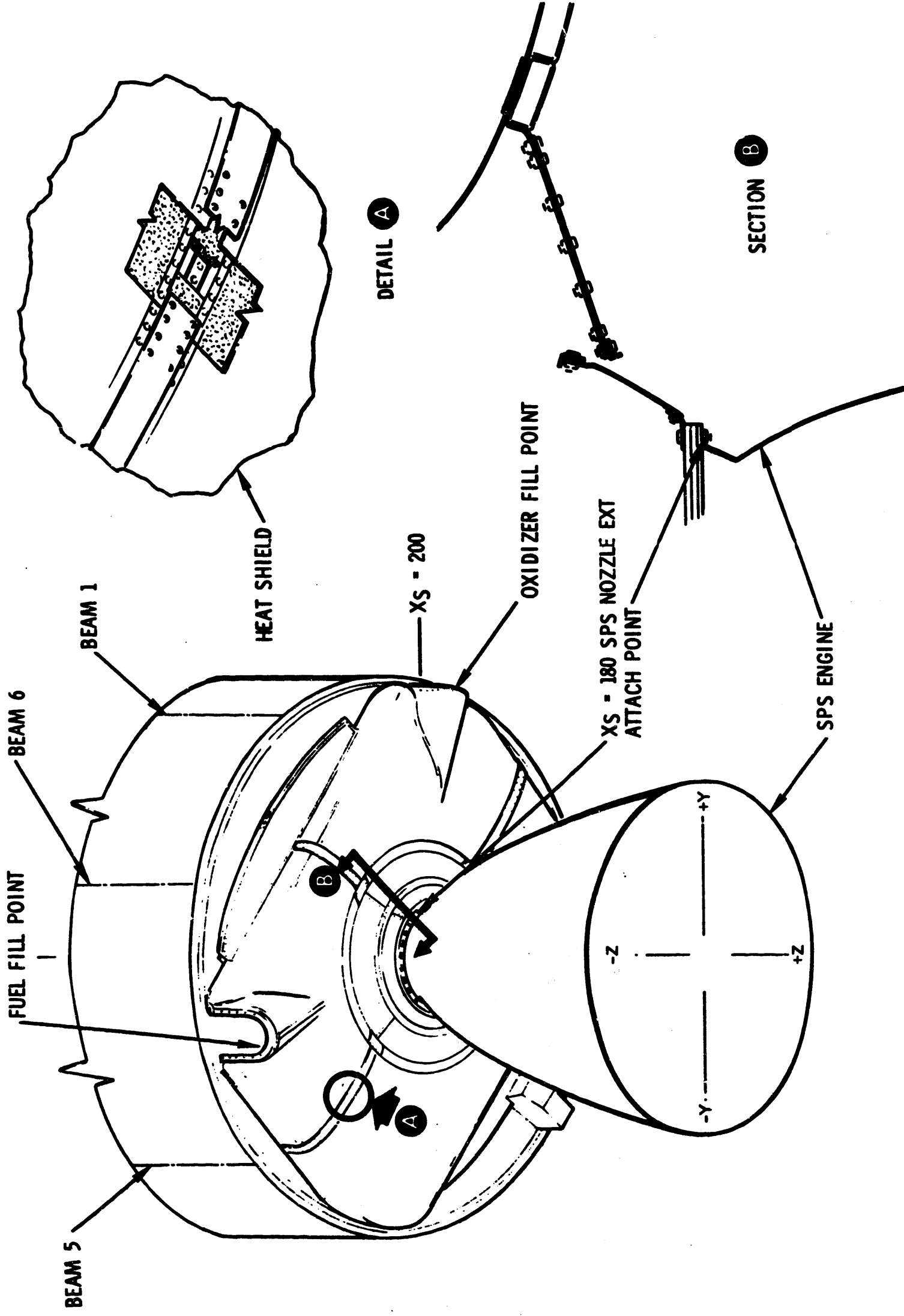
CSM 106 & SUBS



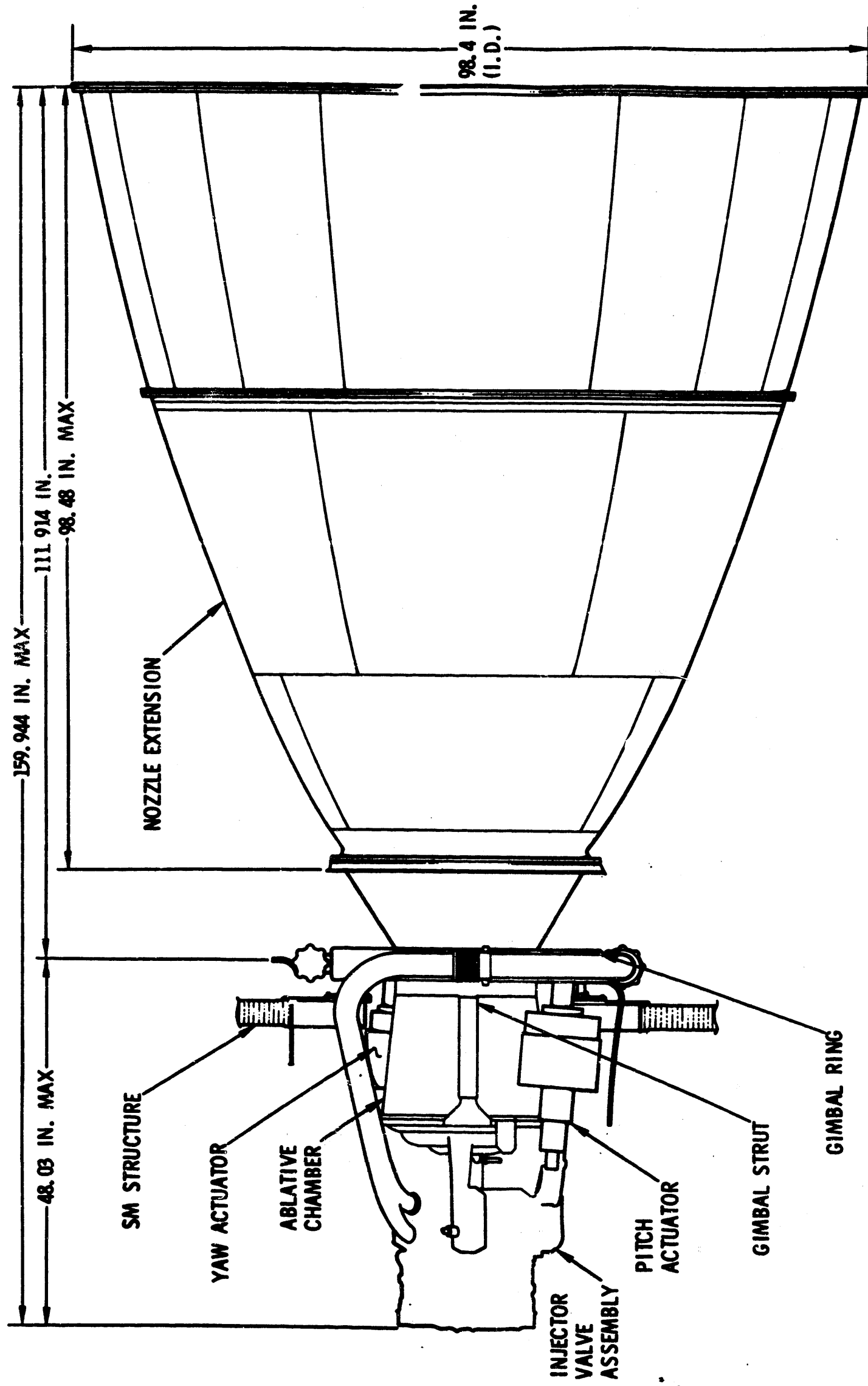
SPS OXIDIZER ENGINE FEED LINE TEMPERATURE MONITORING



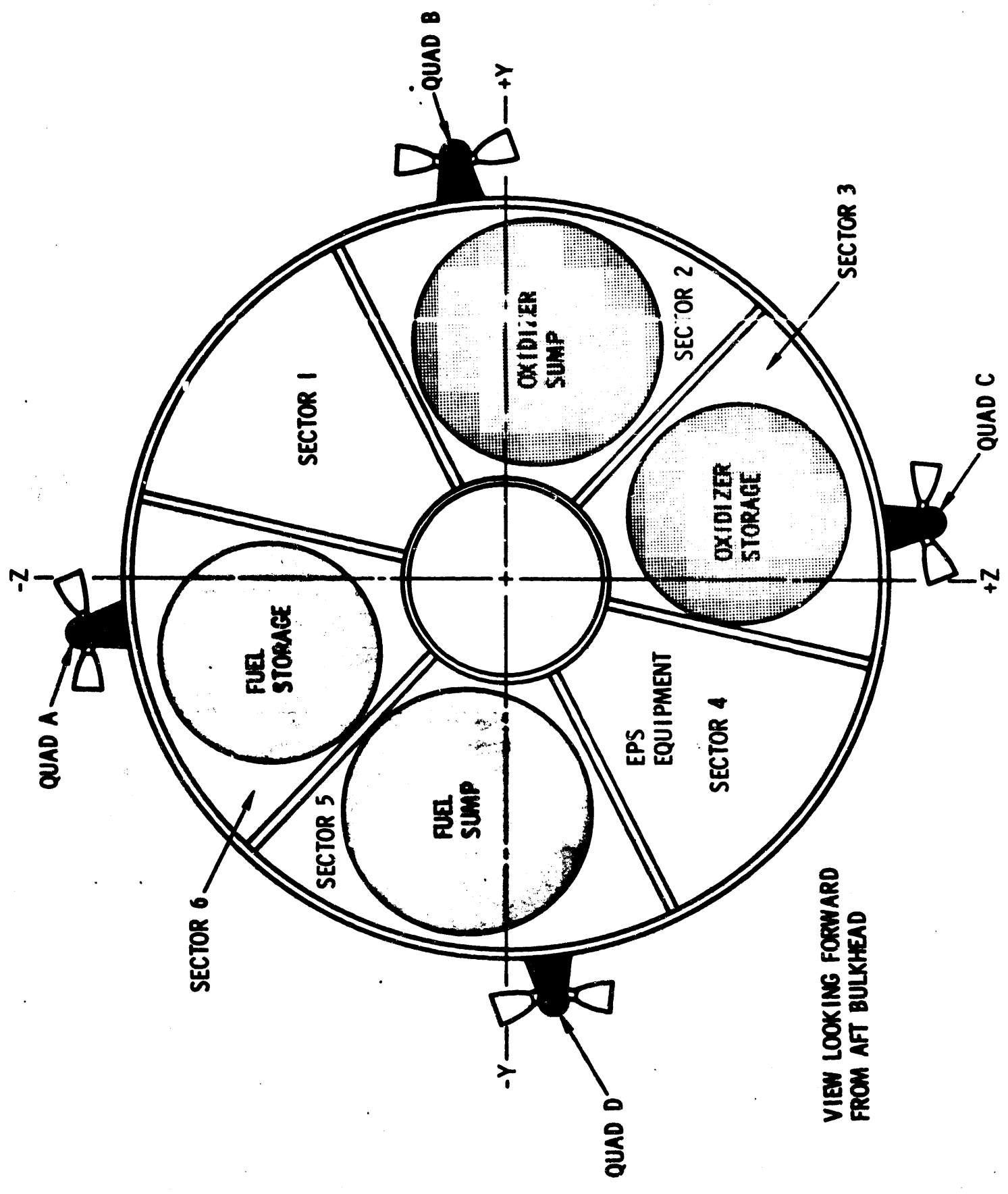
SM SPS NOZZLE AND HEAT SHIELD



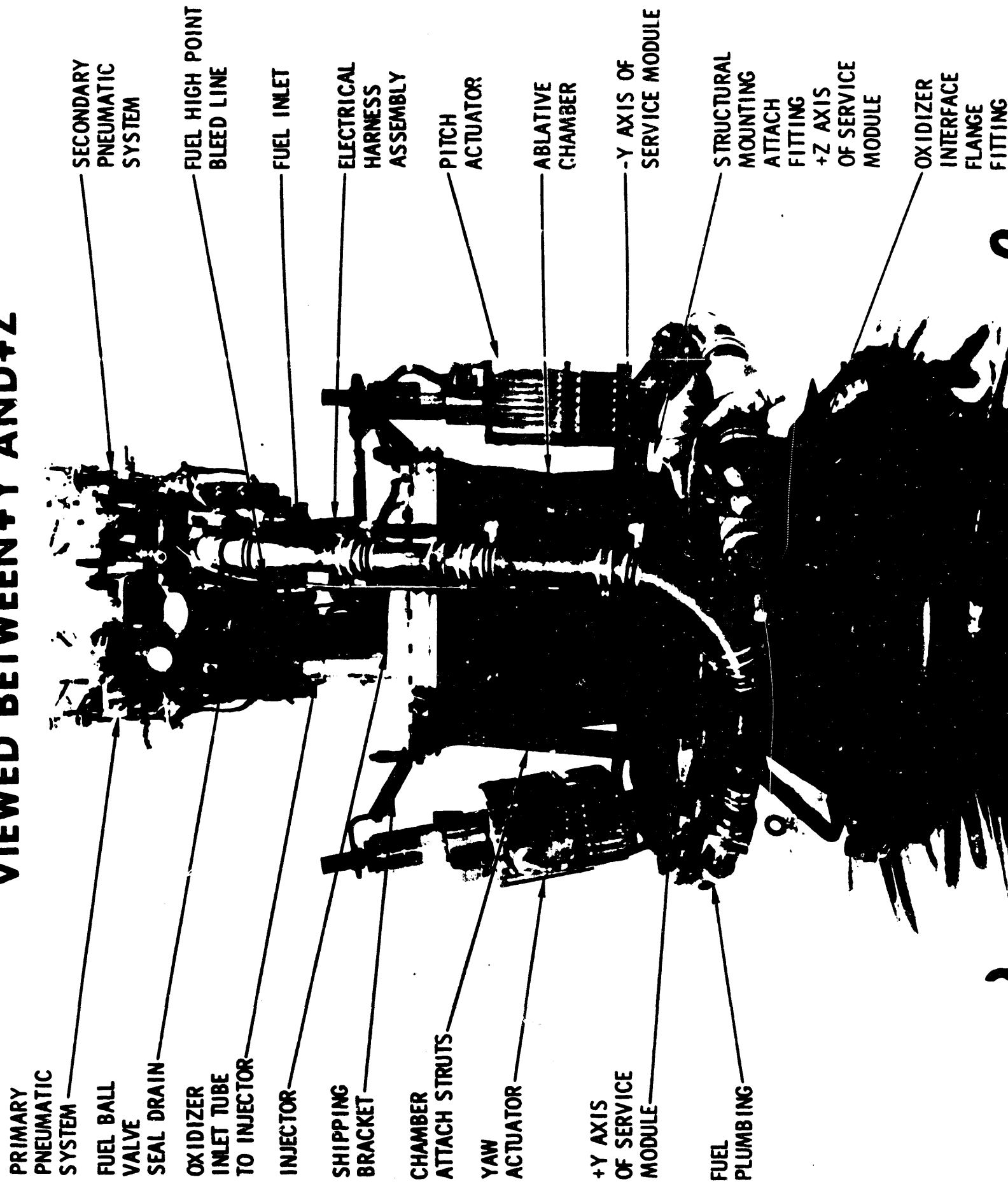
SPS ENGINE AJ10-137



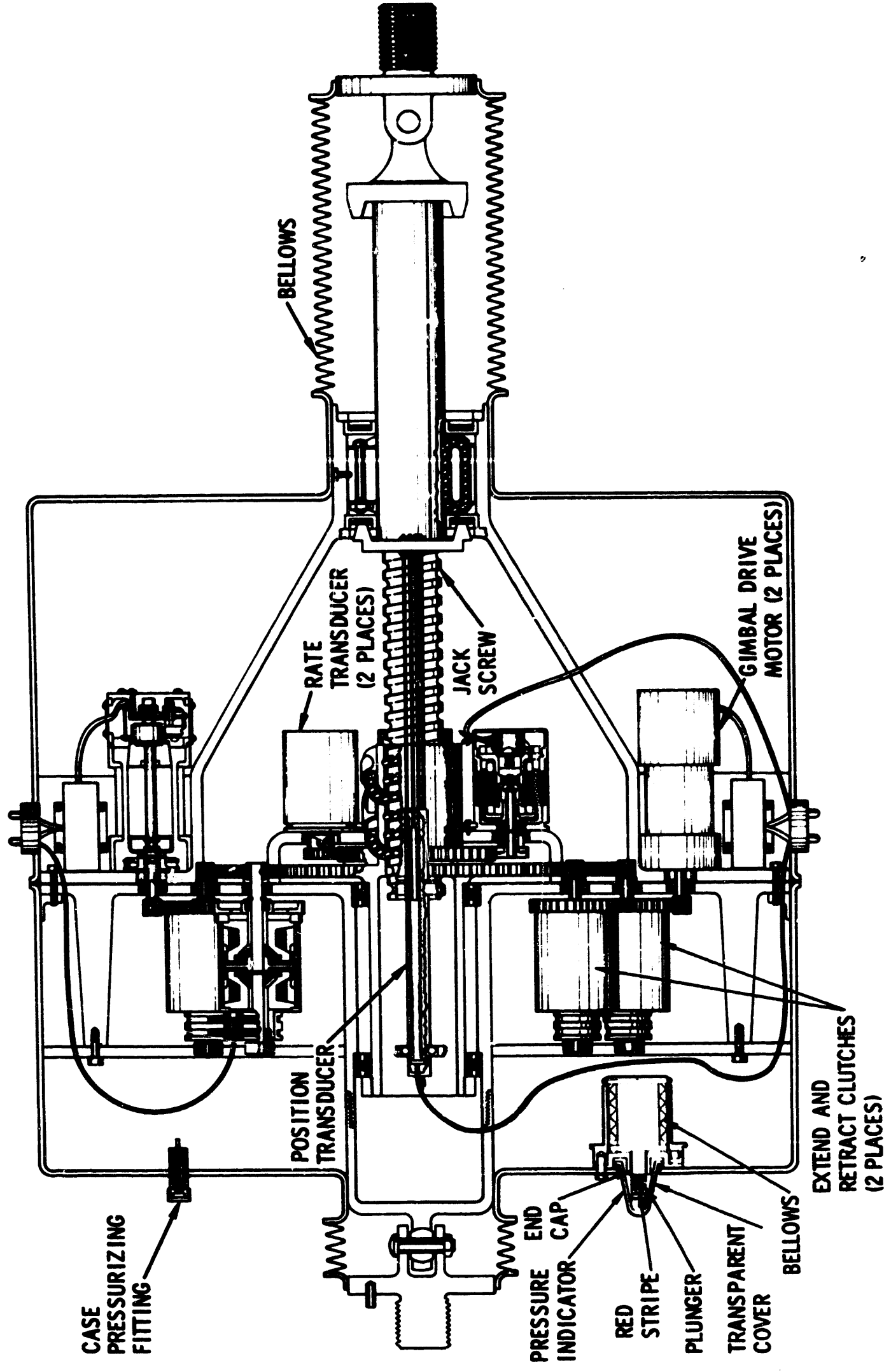
SERVICE MODULE SECTORS

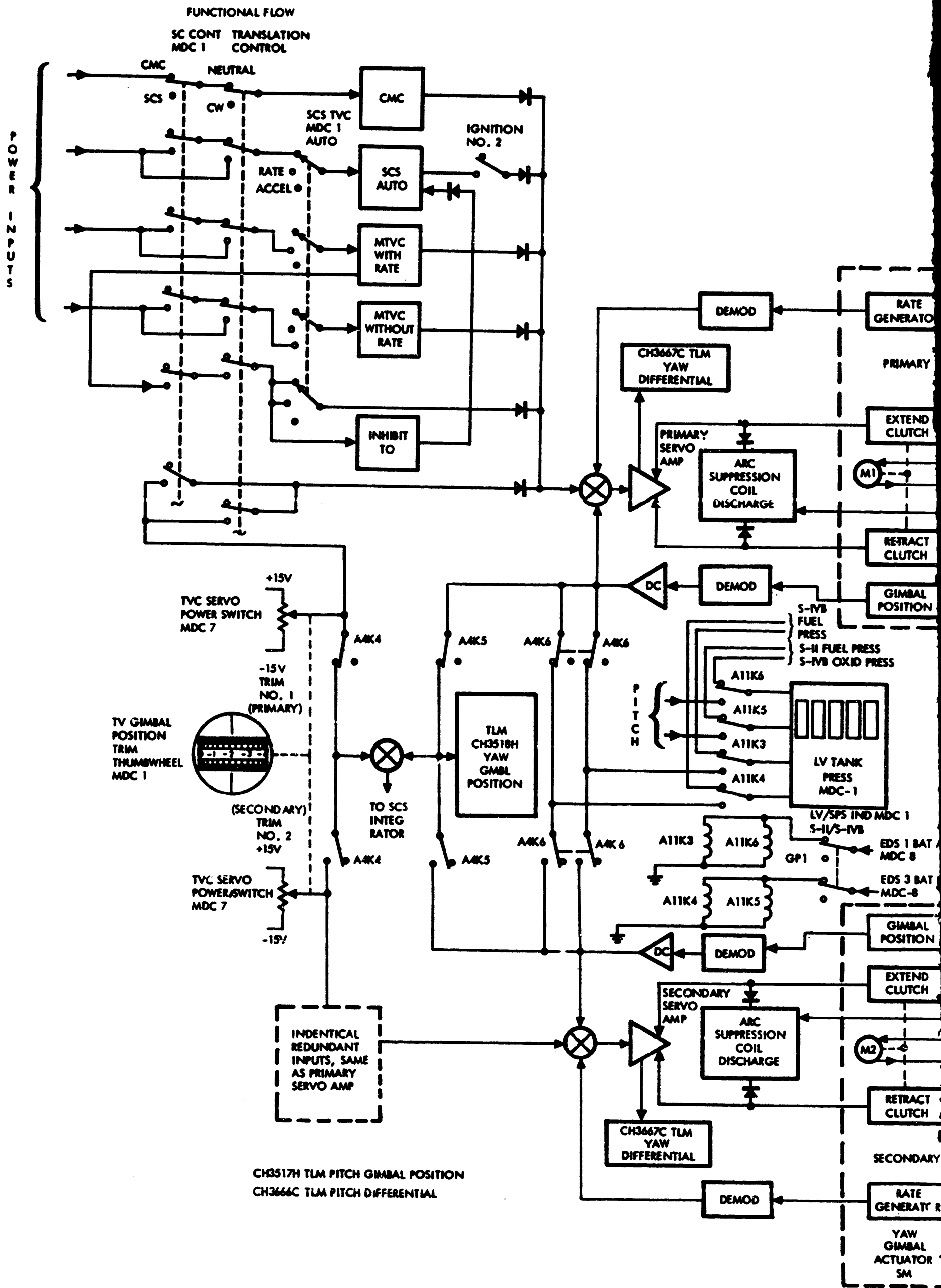


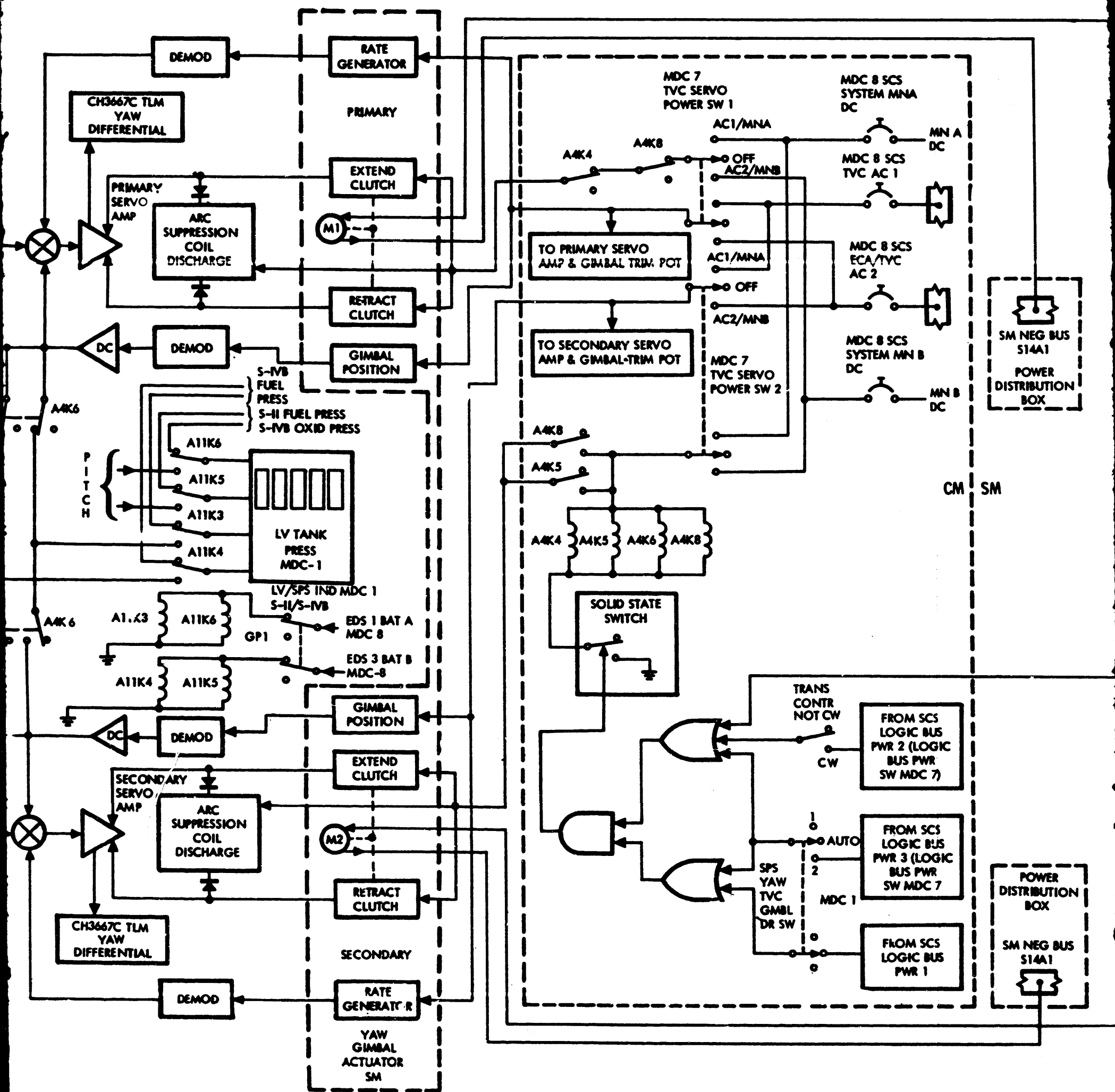
SPS ENGINE VIEWED BETWEEN +Y AND +Z

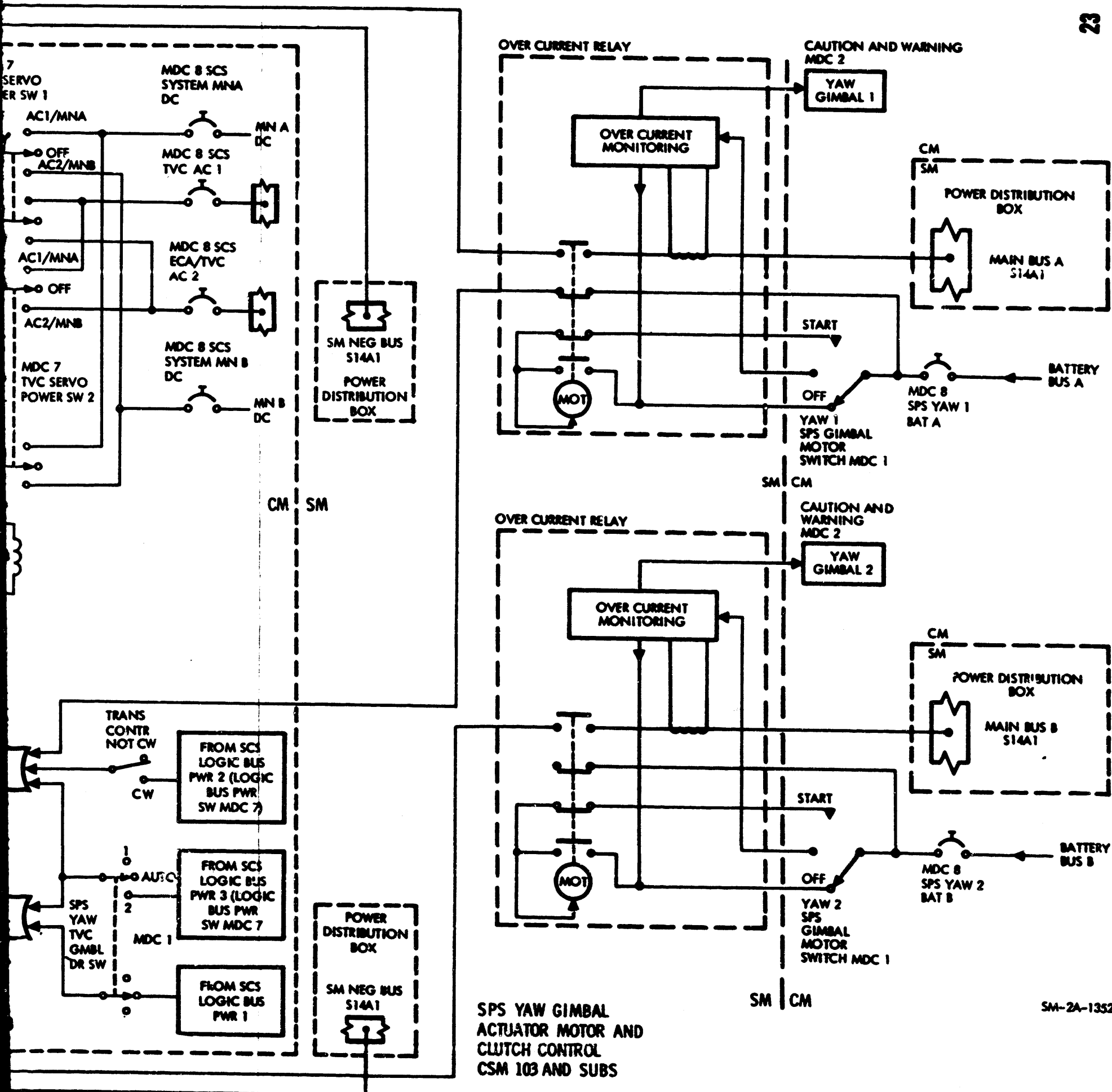


SPS ELECTRO-MECHANICAL GIMBAL ACTUATOR



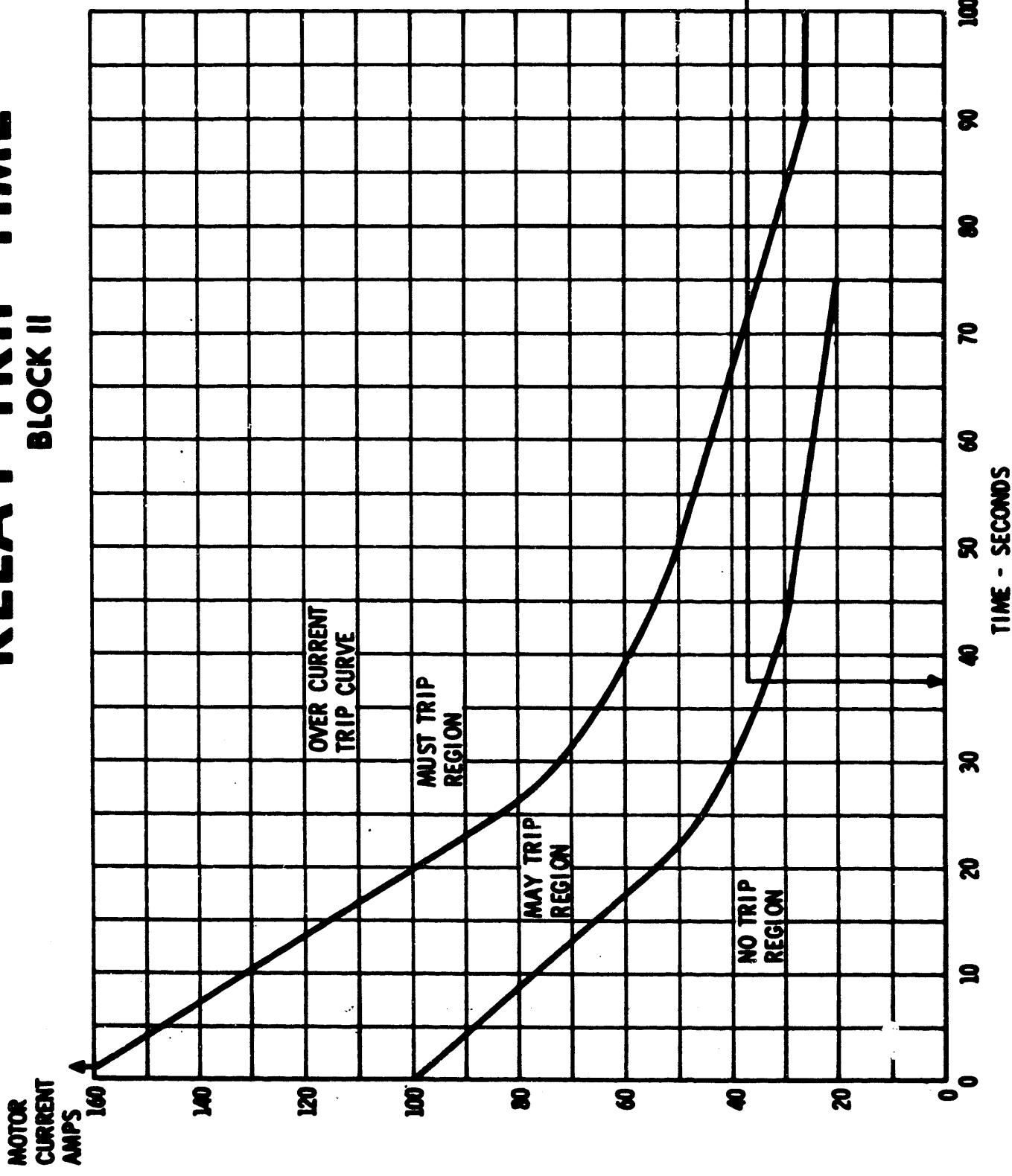






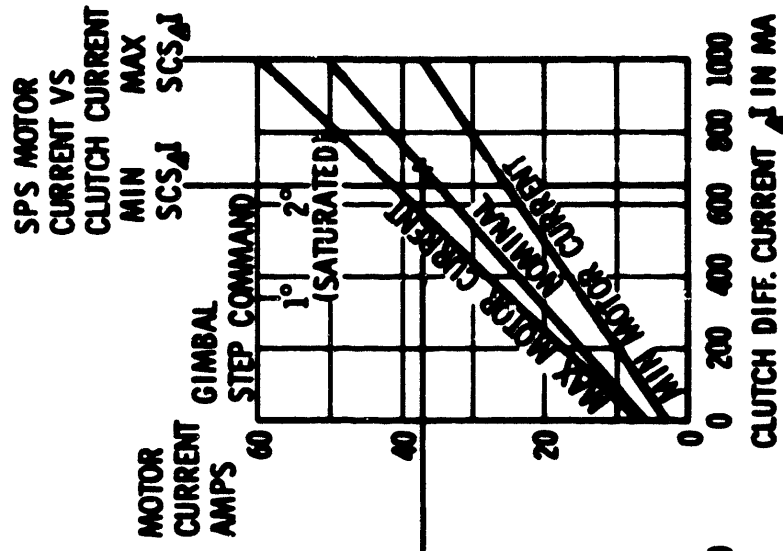
SPS GIMBAL ACTUATOR OVER CURRENT RELAY TRIP TIME

BLOCK II



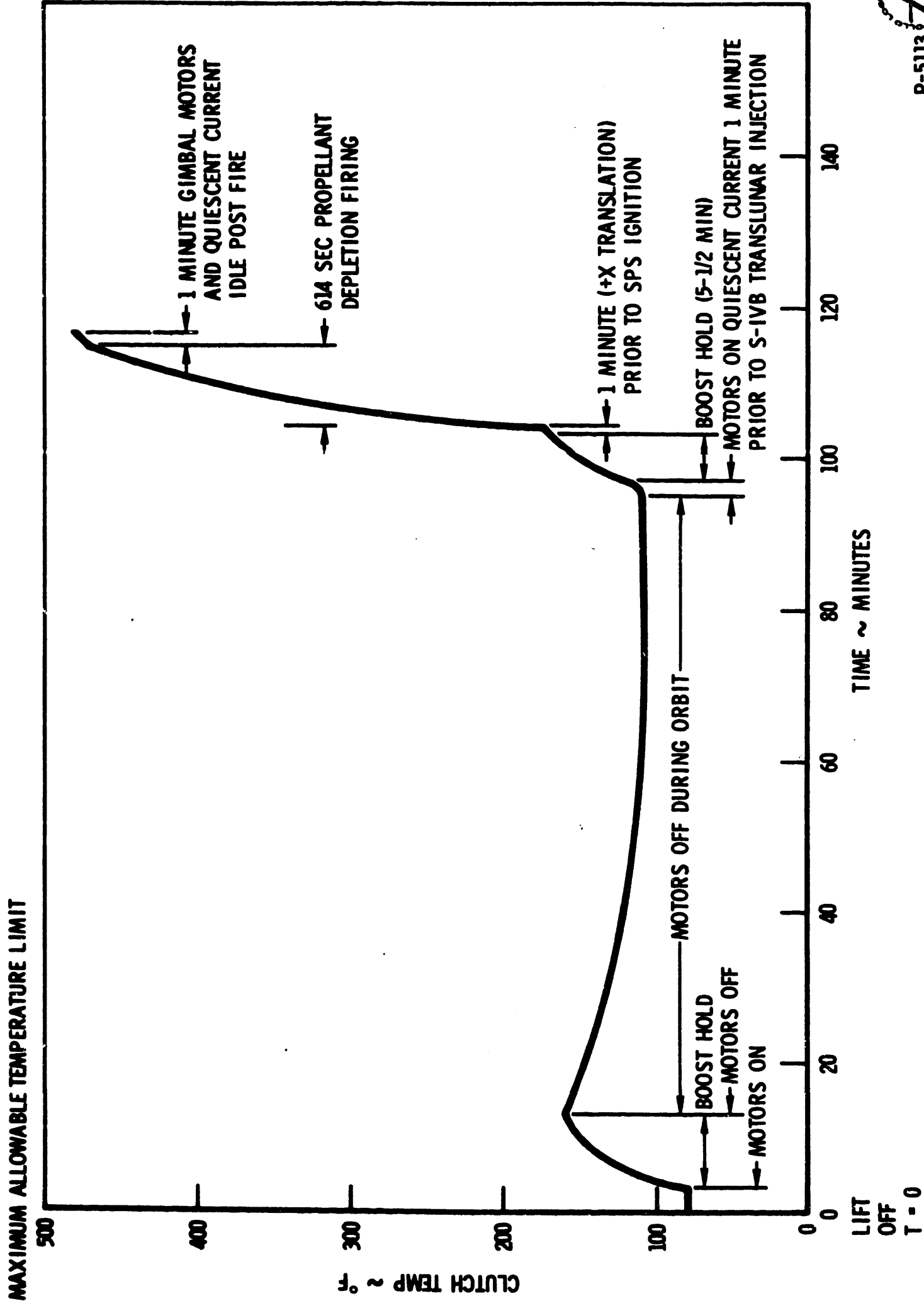
EXAMPLE

1. SATURATED SCS AMPLIFIER
OUTPUT CURRENT OF 700 MA
CAUSES SPS GIMBAL MOTOR TO
DRAW 37 AMPS, OVER CURRENT
RELAY TRIPS IN 37.5 SECONDS
NOMINAL



P-5050A

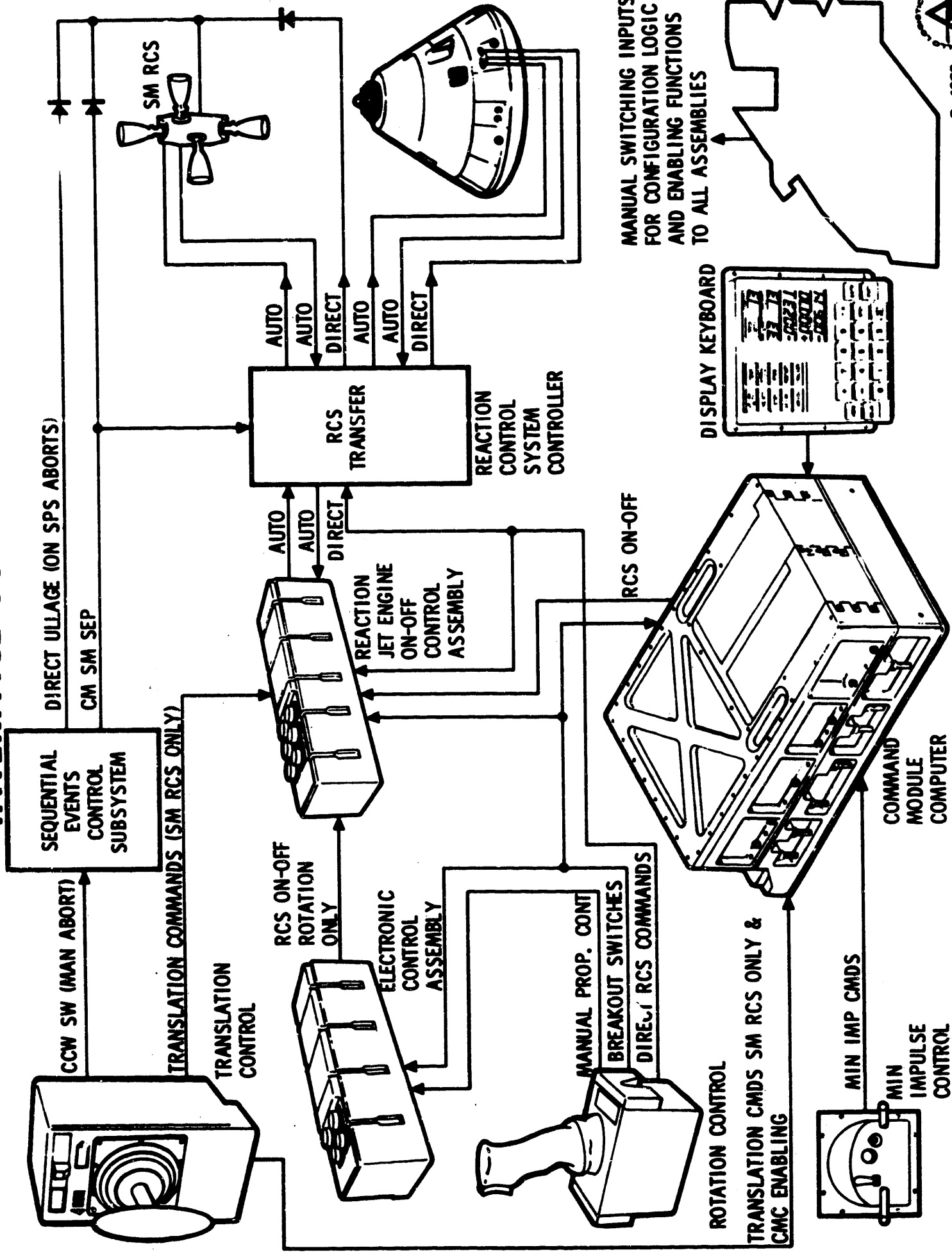
SPS ENGINE GIMBAL ACTUATOR CLUTCH TEMPERATURE



CSM 106 THROUGH CSM 111



CSM REACTION CONTROL SYSTEM INTERFACE CONTROL



P-6078

SM REACTION CONTROL SYSTEM

PURPOSE: PROVIDE THRUST FOR ATTITUDE AND/OR TRANSLATION
CONTROL AFTER SEPARATION FROM THE S-IVB BOOSTER UP UNTIL CM/SM SEPARATION

TYPE OF SYSTEM: NON THROTTLEABLE, PRESSURE FEED SYSTEM, HYPERGOLIC PROPELLANTS AND RADIANT COOLED THRUST CHAMBERS

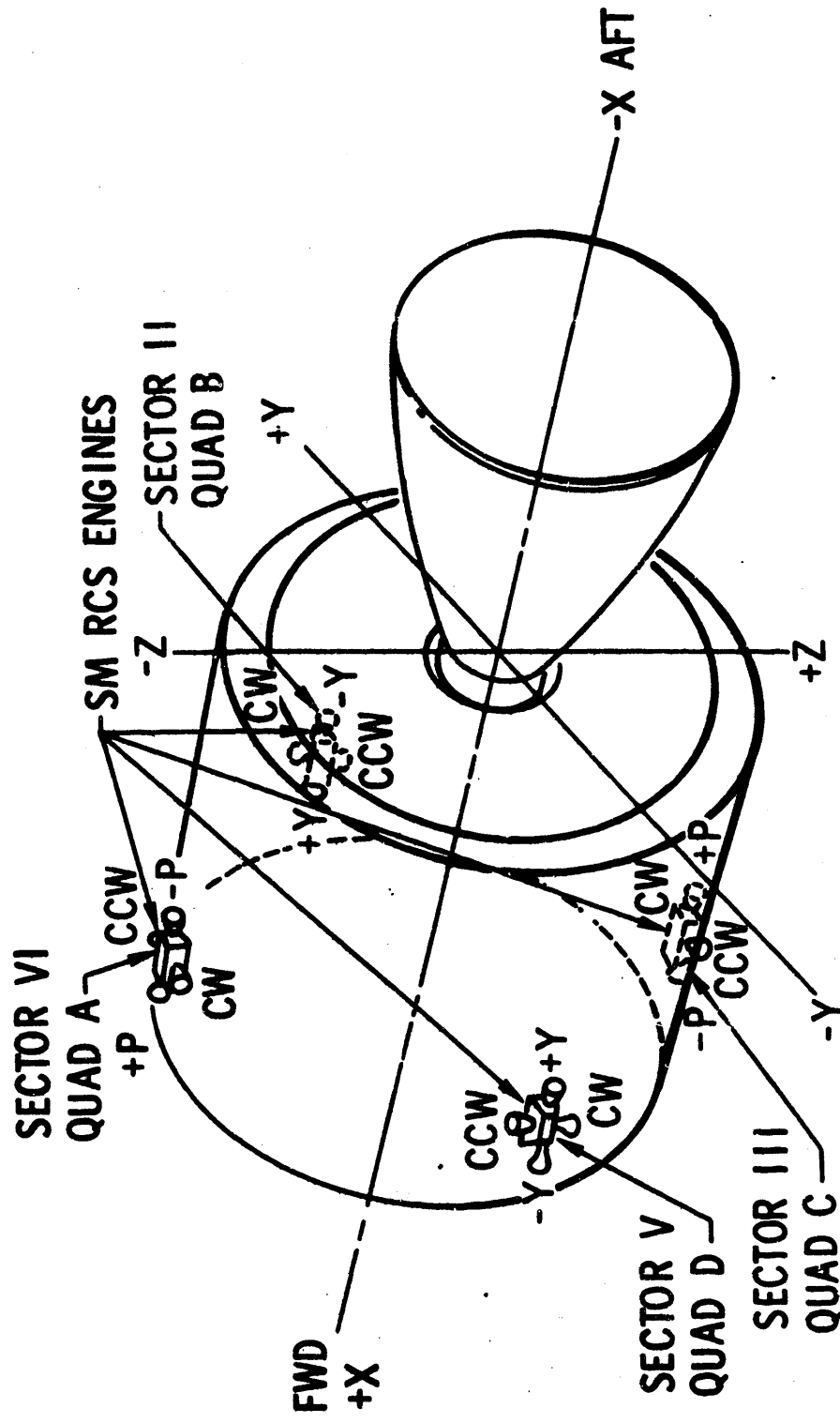
SYSTEM DESCRIPTION: FOUR INDIVIDUAL QUADS

EACH QUAD CONSISTS OF A SEPARATE PROPELLANT

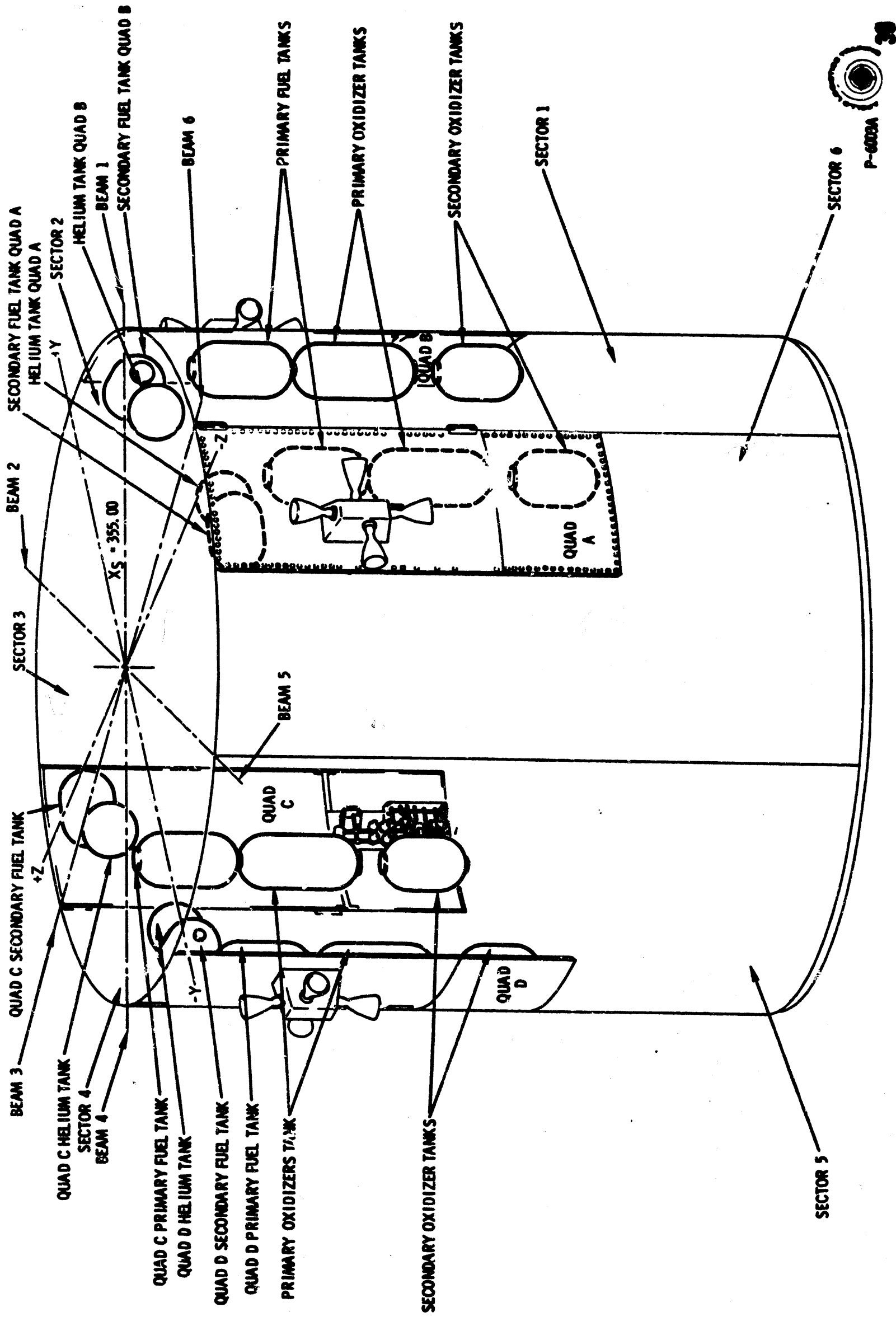
FEED SYSTEM AND FOUR THRUST CHAMBERS

PROPELLANTS - INHIBITED NITROGEN Tetroxide AND MONOMETHYL HYDRAZINE

SYSTEM CONTROLLED BY CMC OR SCS WITH A DIRECT CONTROL BACK-UP CAPABILITY

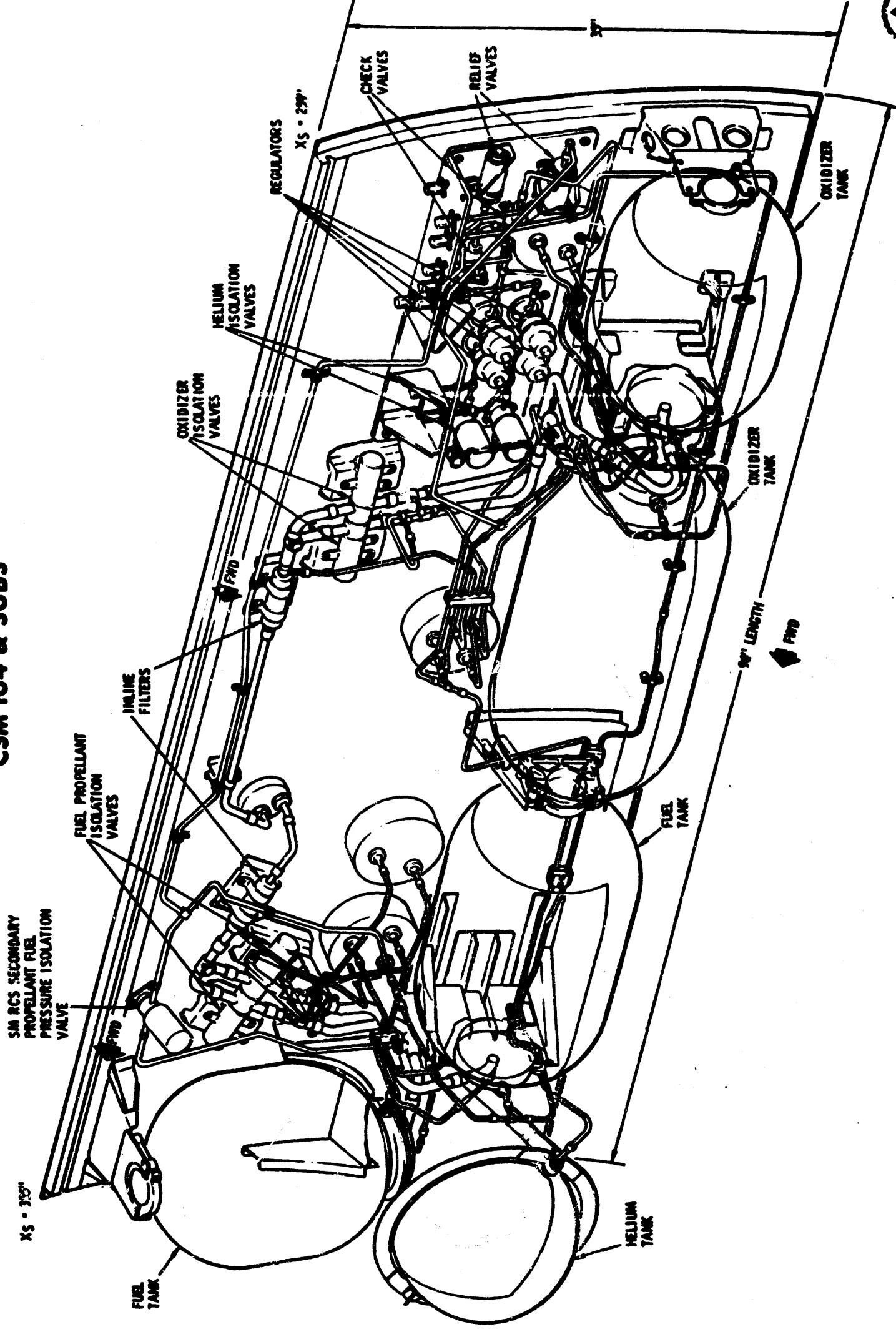


SM RCS QUADS



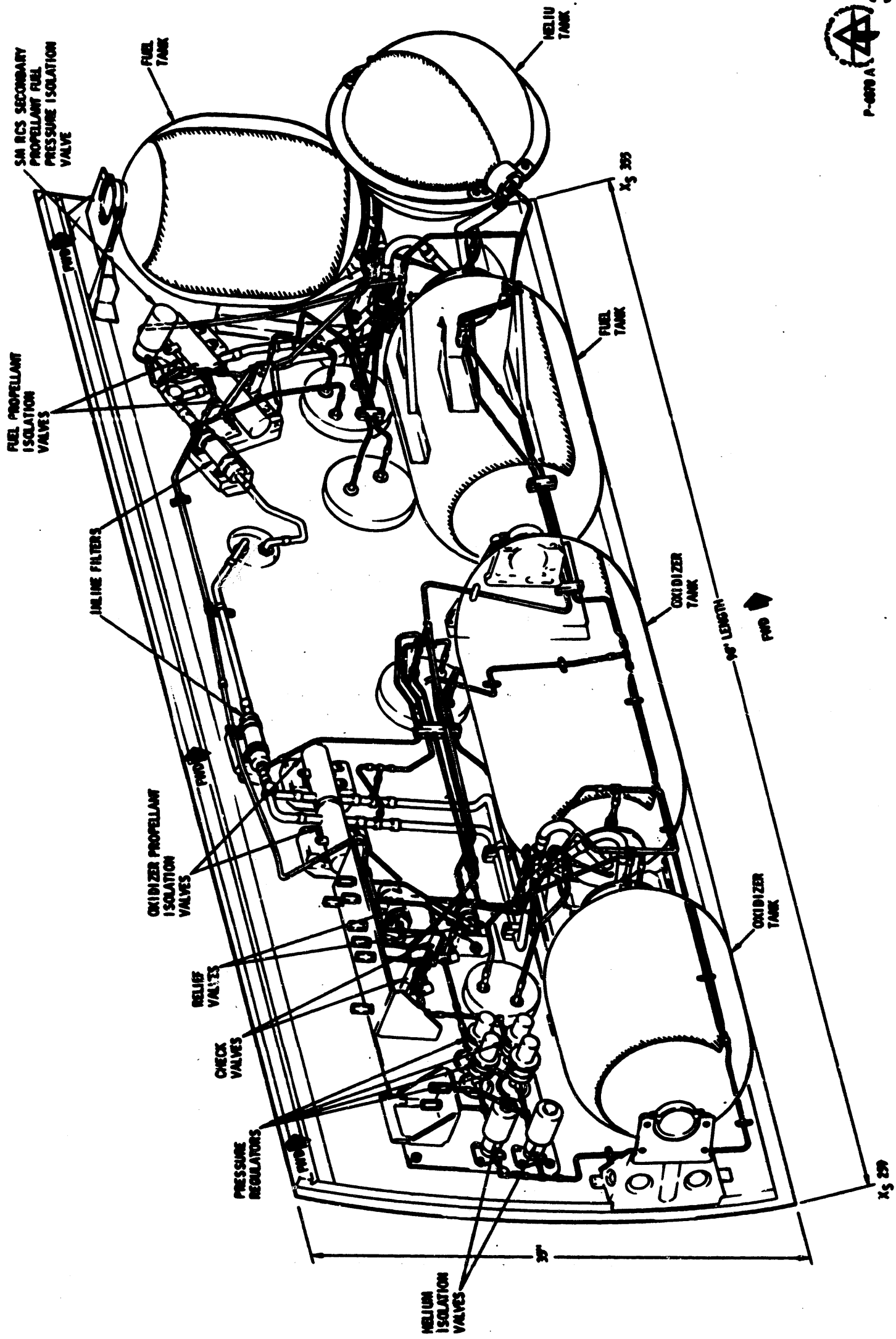
SM RCS PANEL ASSEMBLY QUAD A & C

CSM 104 & SUBS



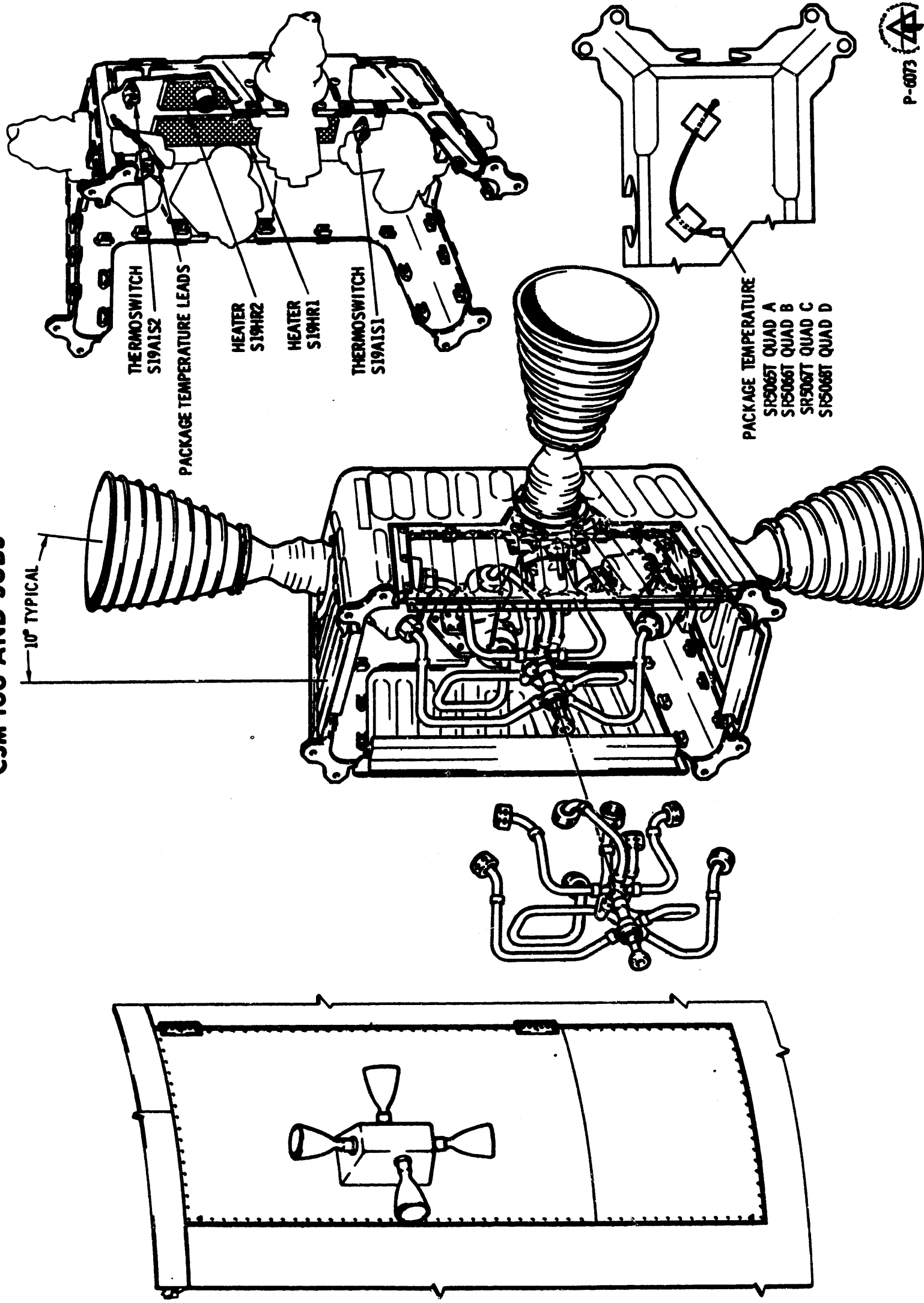
SM RCS PANEL ASSEMBLY QUAD B & D

CSM 104 & SUBS



SM RCS ENGINE HOUSING

CSM 106 AND SUBS



CSM 103 & SUBS



SM RCS SUBSYSTEM Q

CSM 104 & SUBS

NOTE: QUADS B&D CIRCUIT BREAKER MDC-8

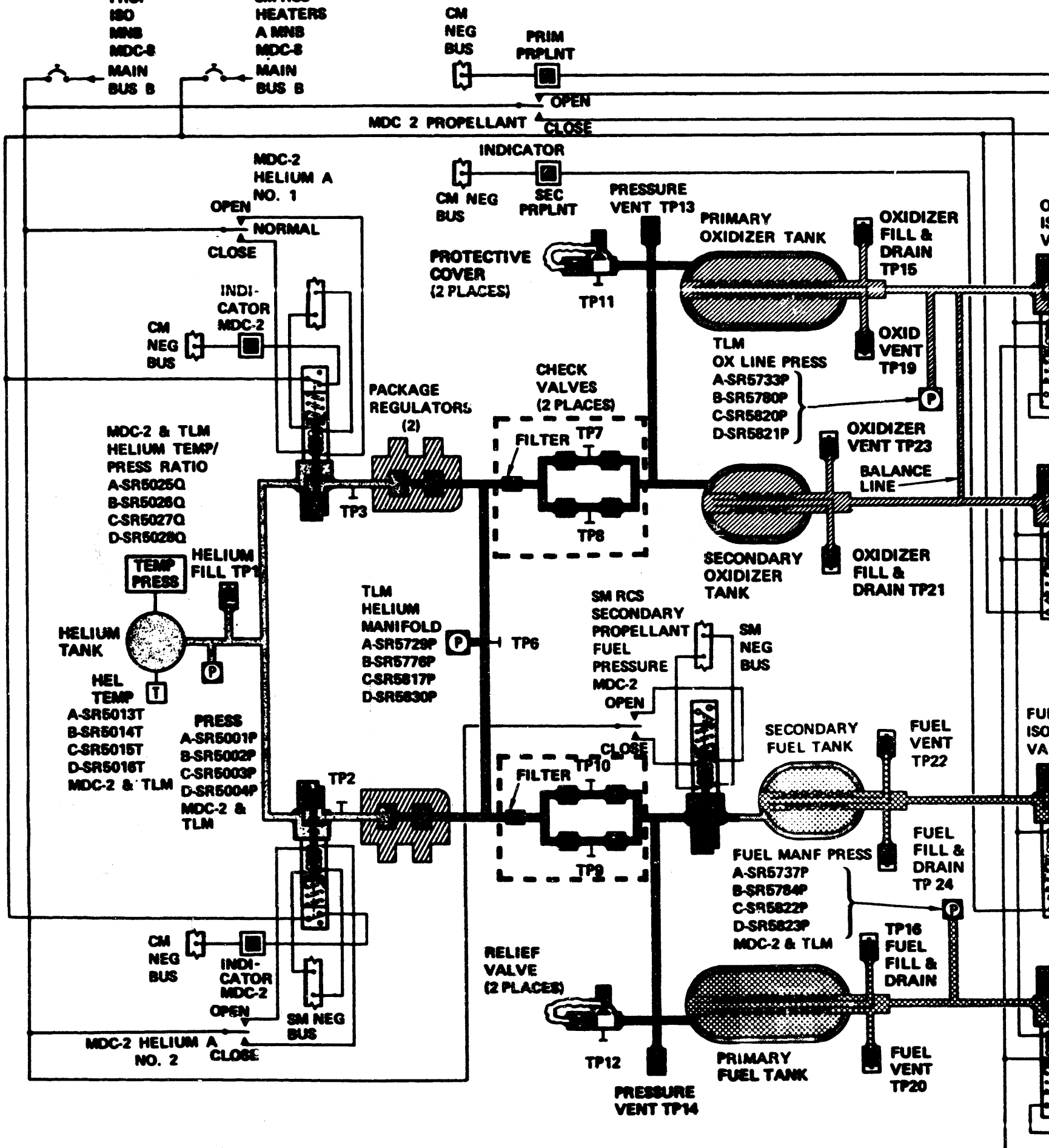
MAIN BUS A
& PROP ISOL
MAIN A

MAIN BUS A
& SM RCS
HEATERS
BMNA

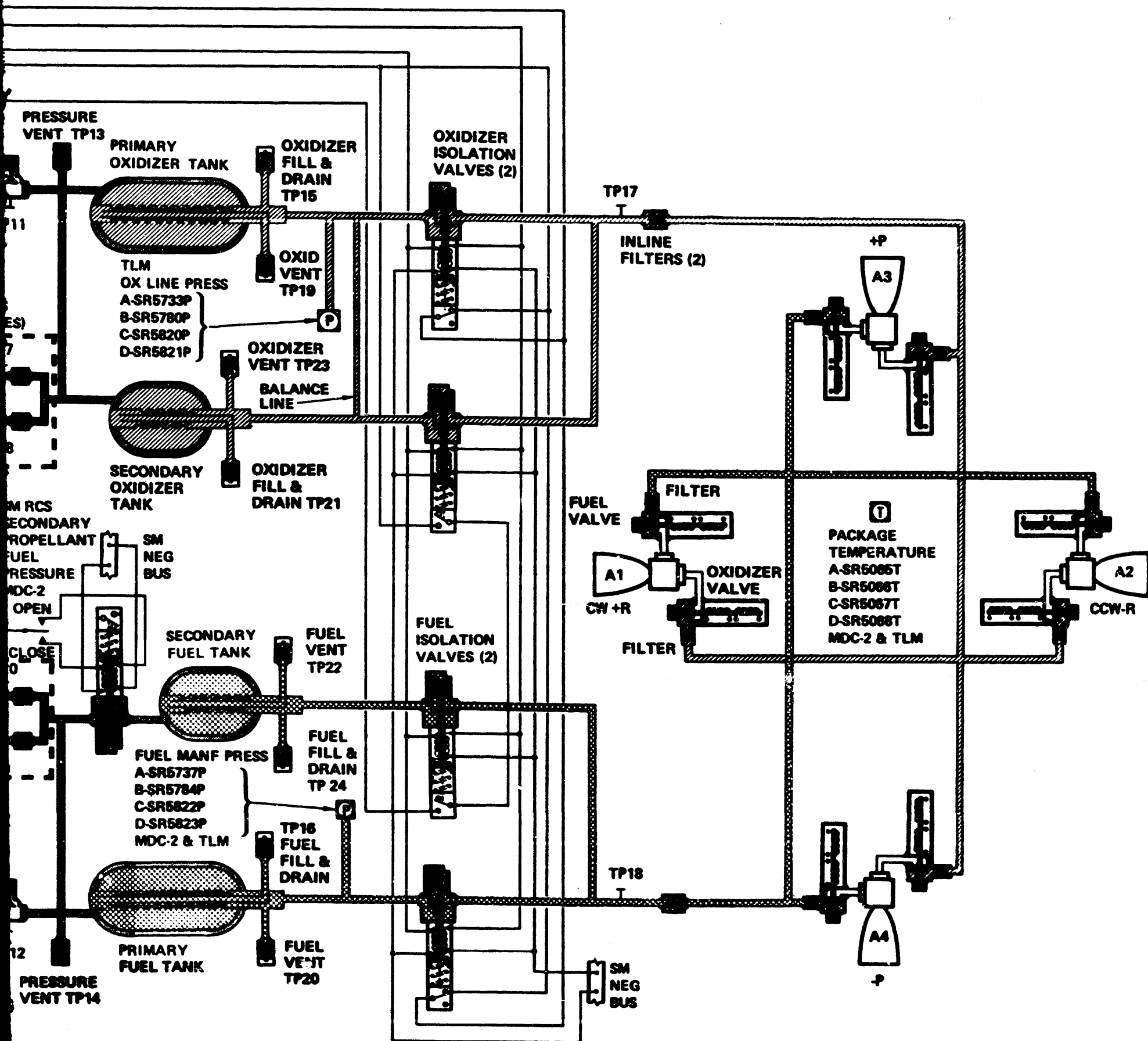
NOTE: QUADS ABC CIRCUIT BREAKER MDC-8

PROP
ISO
MNB
MDC-8
MAIN
BUS B

SM-RCS
HEATERS
A MNB
MDC-8
MAIN
BUS B

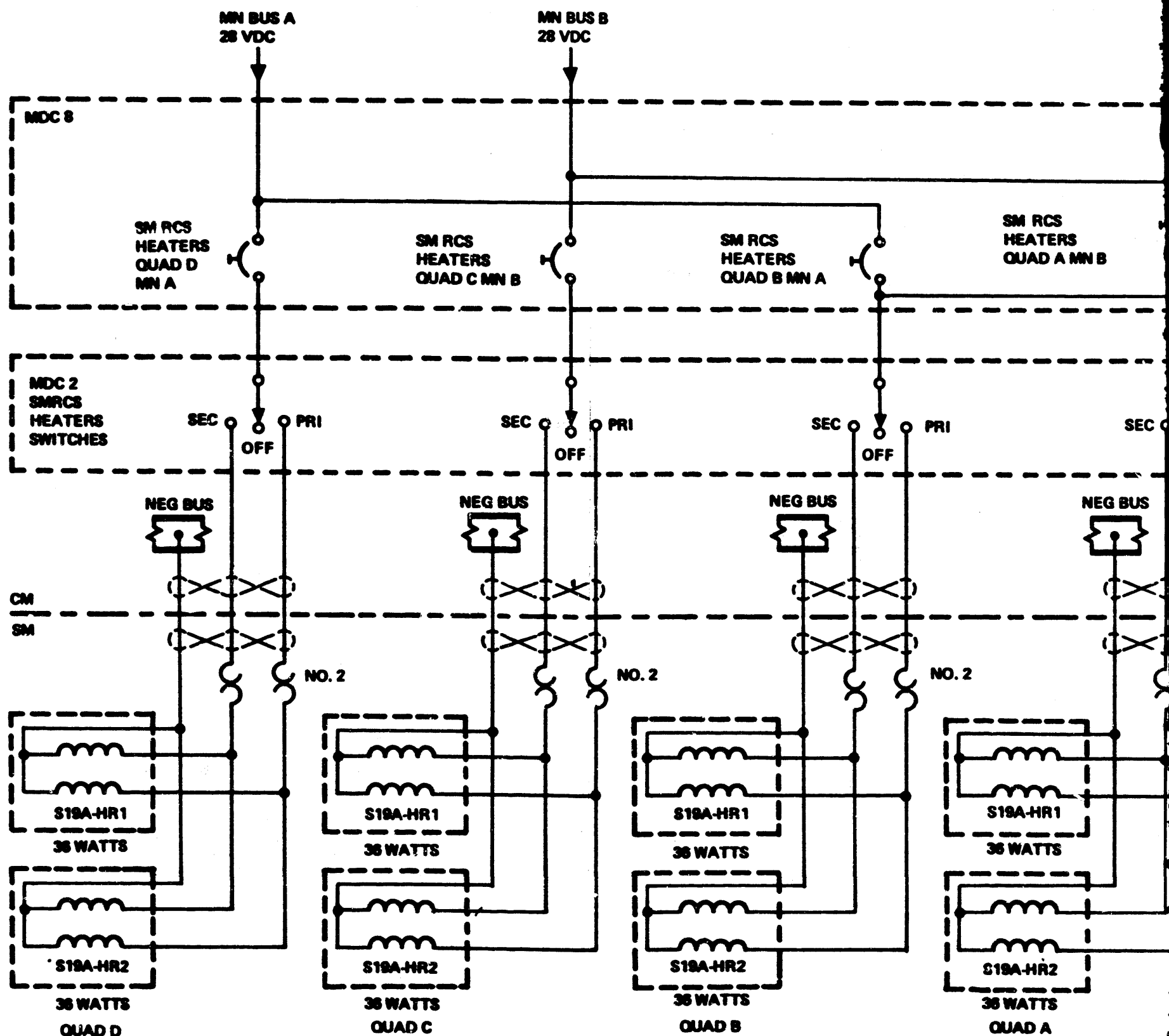


CSM 104 & SUBS



SM RCS ELECTRICAL HEATERS

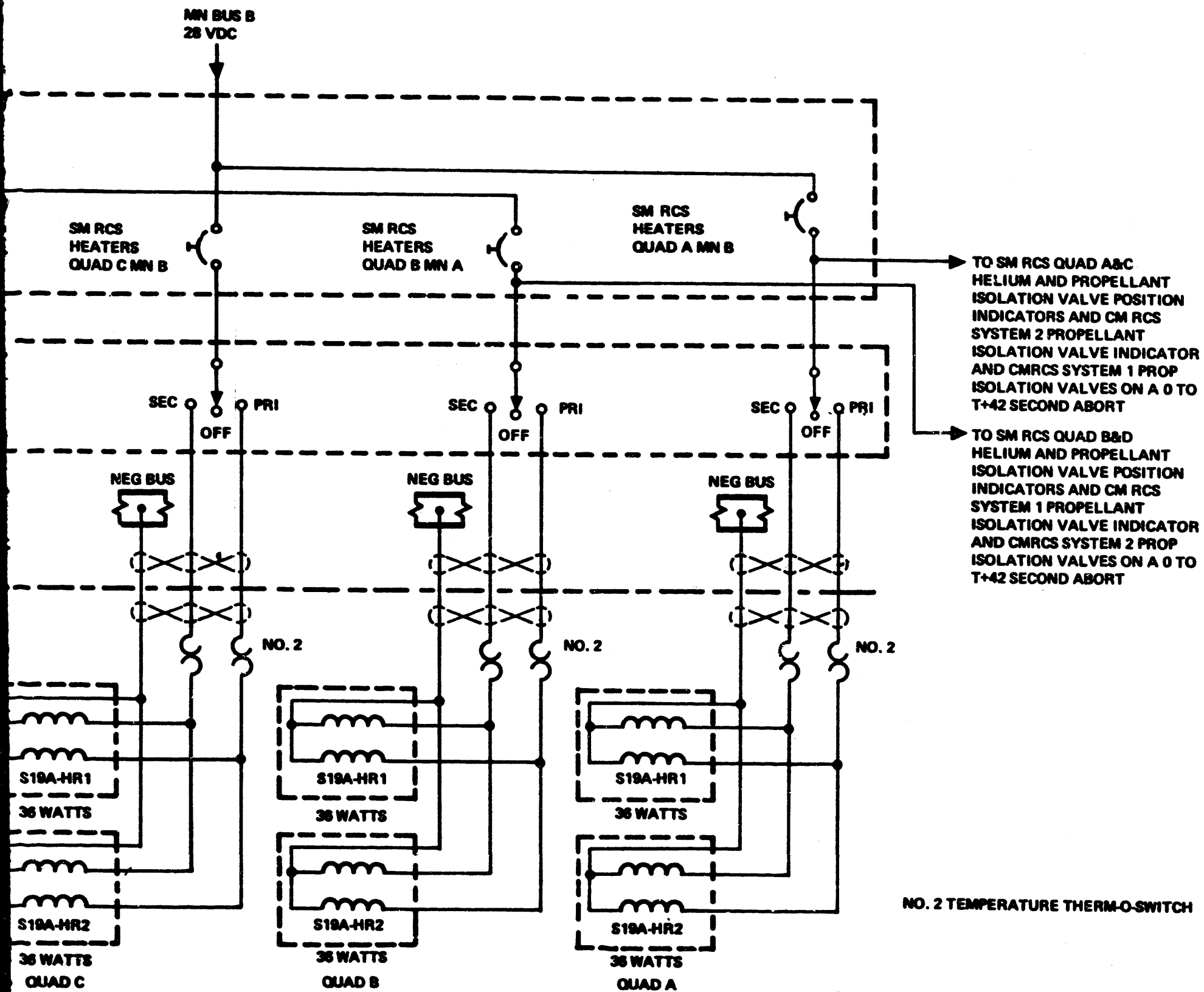
CSM 106 & SUBS



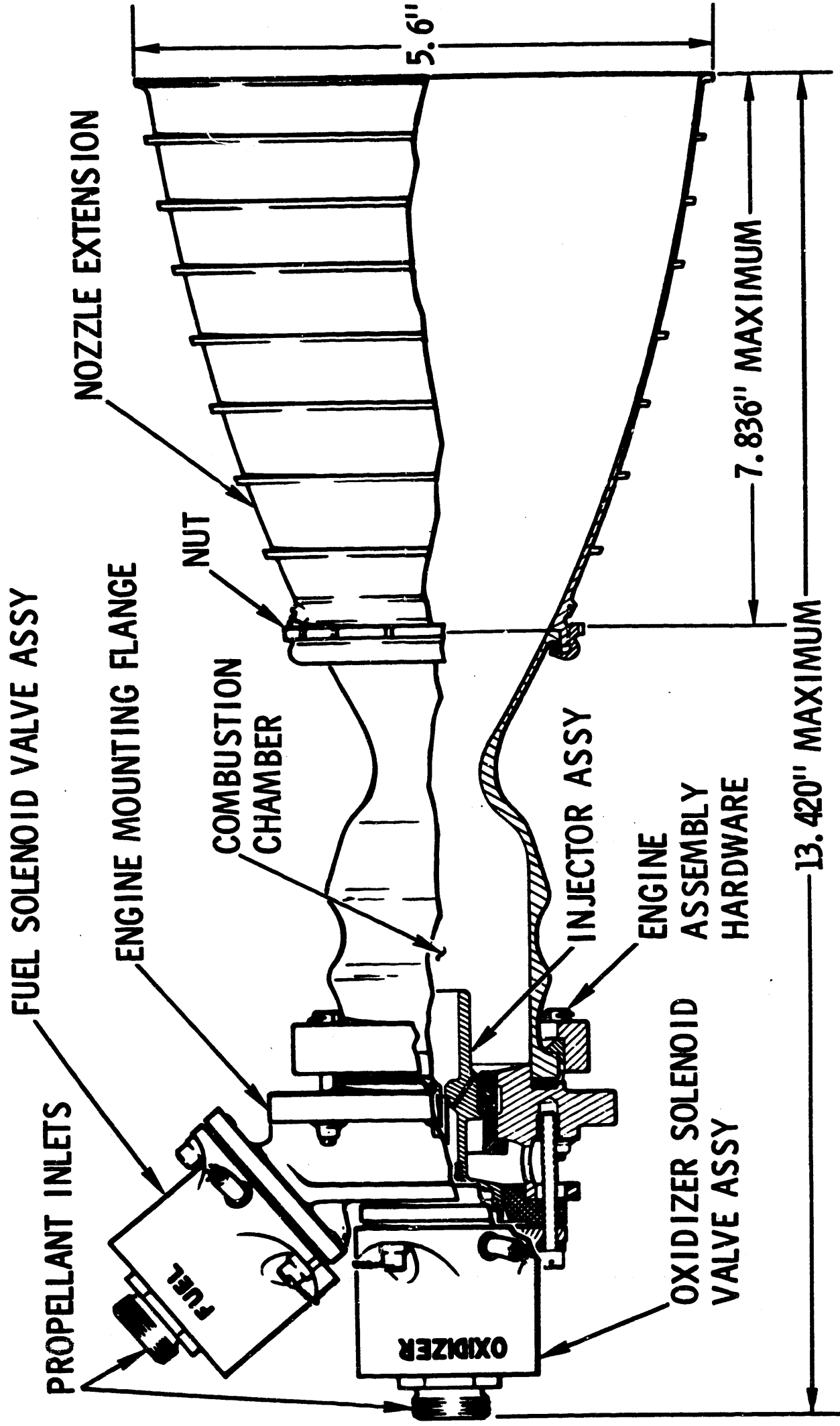
SM RCS ELECTRICAL HEATERS

CSM 106 & SUBS

3

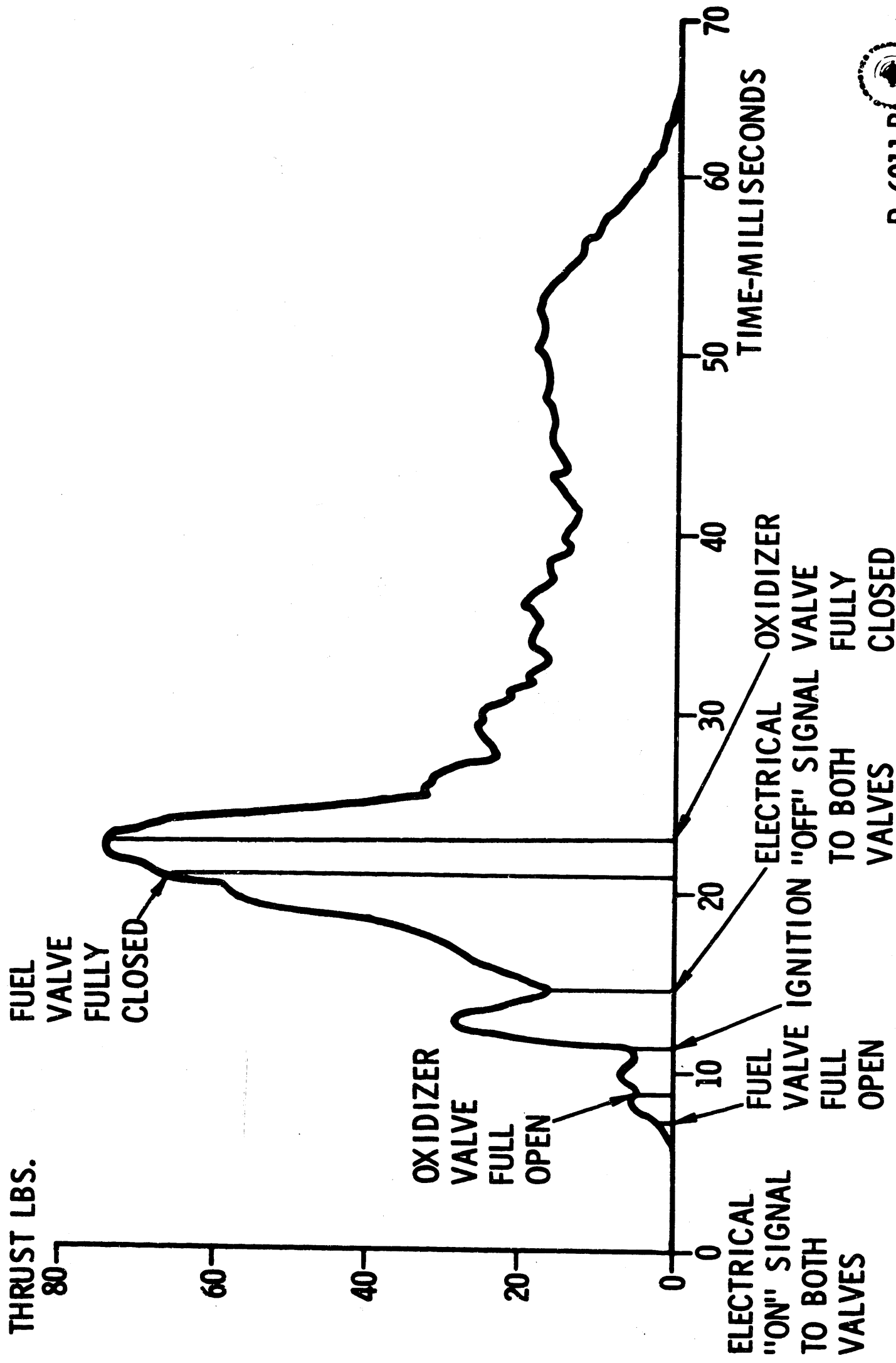


SM RCS ENGINE CUTAWAY



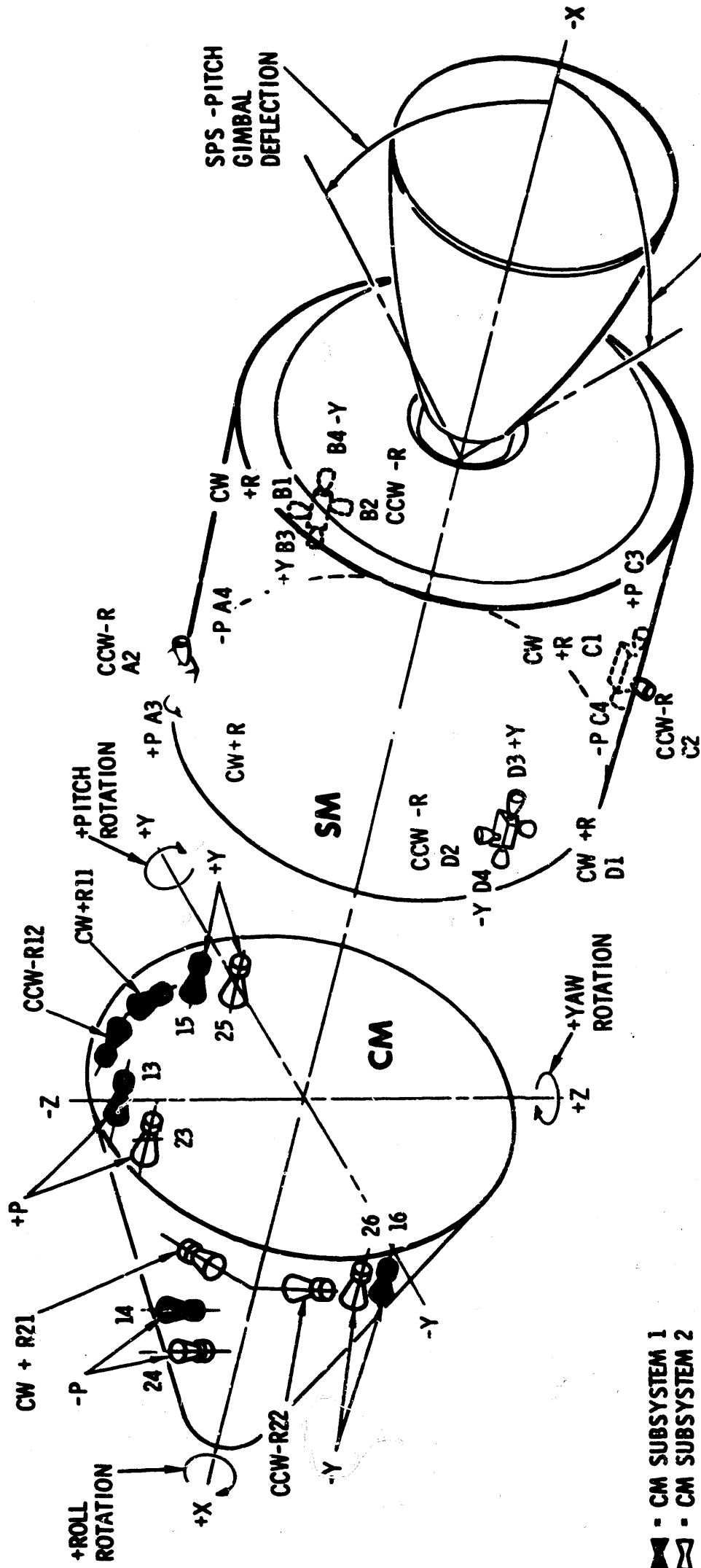
S/M RCS ENGINE-MINIMUM TOTAL IMPULSE

(CONTROLLER REACTION JET ON-OFF ASSEMBLY 15±3 MILLISECONDS)



CM SM ENGINE LOCATIONS

CSM 101 & SUBS



- CM SUBSYSTEM 1
 - CM SUBSYSTEM 2

ODD NUMBER

ENGINES - +ROTATION MANEUVER 1, 3, 5

EVEN NUMBER

ENGINES - -ROTATION MANEUVER 2, 4, 6

A- SM RCS QUAD A

B- SM RCS QUAD B

C- SM RCS QUAD C

D- SM RCS QUAD D

ODD NUMBER

ENGINES - +ROTATION MANEUVER 1, 3

EVEN NUMBER

ENGINES - -ROTATION MANEUVER 2, 4

CSM 103 AND SUBS



SM CM RCS ENGINE POWER SUPPLIES AUTOMATIC

CSM 103 & SUBS

BUS POWER	CIRCUIT BREAKER MDC 8	AUTORCS SELECT SWITCHES MDC 8	SM ENGINE				CM ENGINE				SM ENGINE UTILIZATION FOR TRANSLATION MANEUVERS						TRANSFER MOTOR
			QUAD	MANEUVER	SCS NO.	PROP. NO.	SYSTEM	MANEUVER	SCS NO.	PROP. NO.	-X	+X	-Y	+Y	-Z	+Z	
MMA	SCS MMA PITCH	PITCH C3 +X13	C	+P	C3	S1M303	1	+P	13	C1907	C3					S1	
		PITCH A4 +X14	A	-P	A4	S1M323	1	-P	14	C19011	A4					S1	
		PITCH A3 -X23	A	+P	A3	S1M301	2	+P	23	C1908	A3					S2	
		PITCH C4 -X24	C	-P	C4	S1M301	2	-P	24	C19012	C4					S2	
MNB	SCS MNB PITCH	PITCH C3 +X13	C	+P	C3	S1M303	1	+P	13	C1907	C3					S1	
		PITCH A4 +X14	A	-P	A4	S1M323	1	-P	14	C19011	A4					S1	
		PITCH A3 -X23	A	+P	A3	S1M301	2	+P	23	C1908	A3					S2	
		PITCH C4 -X24	C	-P	C4	S1M301	2	-P	24	C19012	C4					S2	
MMA	SCS MMA YAW	YAW B3 +X25	D	+Y	D3	S1M401	2	+Y	25	C19010	D3					S2	
		YAW B4 +X26	B	-Y	B4	S1M421	2	-Y	26	C1902	B4					S2	
		YAW B3 -X15	B	+Y	B3	S1M423	1	+Y	15	C1909	B3					S1	
		YAW B4 -X16	D	-Y	D4	S1M403	1	-Y	16	C1901	B4					S1	
MNB	SCS MNB YAW	YAW B3 +X25	D	+Y	D3	S1M401	2	+Y	25	C19010	D3					S2	
		YAW B4 +X26	B	-Y	B4	S1M421	2	-Y	26	C1902	B4					S2	
		YAW B3 -X15	B	+Y	B3	S1M423	1	+Y	15	C1909	B3					S1	
		YAW B4 -X16	D	-Y	D4	S1M403	1	-Y	16	C1901	B4					S1	
MMA	SCS MMA B/D ROLL	B/D ROLL B1 +X11	B	CW+R	B1	S1M202	1	CW+R	11	C1905					B1	S1	
		B/D ROLL B2 +X22	D	CCW-R	D2	S1M404	2	CCW-R	22	C1904					B2	S2	
		B/D ROLL B1 -Z21	D	CW+R	D1	S1M402	2	CW+R	21	C1906				D1		S2	
		B/D ROLL B2 -Z12	B	CCW-R	B2	S1M204	1	CCW-R	12	C1903				B2		S1	
MNB	SCS MNB B/D ROLL	B/D ROLL B1 +X11	B	CW+R	B1	S1M202	1	CW+R	11	C1905					B1	S1	
		B/D ROLL B2 +X22	D	CCW-R	D2	S1M404	2	CCW-R	22	C1904					B2	S2	
		B/D ROLL B1 -Z21	D	CW+R	D1	S1M402	2	CW+R	21	C1906				D1		S2	
		B/D ROLL B2 -Z12	B	CCW-R	B2	S1M204	1	CCW-R	12	C1903				B2		S1	
MMA	SCS MMA A/C ROLL	A/C ROLL A1 +Y	A	CW+R	A1	S1M104								A1			
		A/C ROLL C2 +Y	C	CCW-R	C2	S1M302								C2			
		A/C ROLL C1 -Y	C	CW+R	C1	S1M304								C1			
		A/C ROLL A2 -Y	A	CCW-R	A2	S1M102								A2			
MNB	SCS MNB A/C ROLL	A/C ROLL A1 +Y	A	CW+R	A1	S1M104								A1			
		A/C ROLL C2 +Y	C	CCW-R	C2	S1M302								C2			
		A/C ROLL C1 -Y	C	CW+R	C1	S1M304								C1			
		A/C ROLL A2 -Y	A	CCW-R	A2	S1M102								A2			

HEAD FACED AT
CM SM SEPARATION
AND LES ABORTS

CSM 103 AND SUBS



SM CM RCS ENGINE POWER SUPPLIES DIRECT ROTATION CONTROL NO. 1

CSM 103 AND SUBS

BUS POWER	CIRCUIT BREAKER MDC-8	DIRECT I SWITCH MDC 1	ROTATION CONTROL	SM ENGINE			CM ENGINE			TRANSFER MOTOR		
				QUAD	MANEUVER	SCS NO.	PROP NO.	SYSTEM	MANEUVER		SCS NO.	PROP NO.
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A/MMB R.C. 1 MM A/MMB R.C. 1	I-P I-P	C	+P +P	C3 A3	S19A3B3 S19A1B1	1 2	+P +P	13 23	C1987 C1988	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A R.C. 1 MM A R.C. 1	I-P I-P	C	+P +P	C3 A3	S19A3B3 S19A1B1	1 2	+P +P	13 23	C1987 C1988	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A/MMB R.C. 1 MM A/MMB R.C. 1	I-P I-P	A	-P -P	A4 C4	S19A1B3 S19A3B1	1 2	-P -P	14 24	C19811 C19812	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A R.C. 1 MM A R.C. 1	I-P I-P	A	-P -P	A4 C4	S19A1B3 S19A3B1	1 2	-P -P	14 24	C19811 C19812	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A/MMB R.C. 1 MM A/MMB R.C. 1	I-P I-P	B	+Y +Y	B3 D3	S19A2B3 S19A4B1	1 2	+Y +Y	15 25	C1989 C19810	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A R.C. 1 MM A R.C. 1	I-P I-P	B	+Y +Y	B3 D3	S19A2B3 S19A4B1	1 2	+Y +Y	15 25	C1989 C19810	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A/MMB R.C. 1 MM A/MMB R.C. 1	I-P I-P	D	-Y -Y	D4 B4	S19A4B3 S19A2B1	1 2	-Y -Y	16 26	C1981 C1982	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A R.C. 1 MM A R.C. 1	I-P I-P	D	-Y -Y	D4 B4	S19A4B3 S19A2B1	1 2	-Y -Y	16 26	C1981 C1982	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A/MMB R.C. 1 MM A/MMB R.C. 1	I-P I-P	C	CW+R CW+R	C1 A1	S19A3B4 S19A1B4	1 2	CW+R DEADFACED AT CM SM SEPARATION AND LES ABORTS	11 21	C1985 C1986	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A R.C. 1 MM A R.C. 1	I-P I-P	C	CW+R CW+R	C1 A1	S19A3B4 S19A1B4	1 2	CW+R DEADFACED AT CM SM SEPARATION AND LES ABORTS	11 21	C1985 C1986	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A/MMB R.C. 1 MM A/MMB R.C. 1	I-P I-P	D	CW+R CW+R	D1 B1	S19A4B2 S19A2B2	1 2	CW+R DEADFACED AT CM SM SEPARATION AND LES ABORTS	21 22	C1986 C1987	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A R.C. 1 MM A R.C. 1	I-P I-P	C	CCW-R CCW-R	C2 A2	S19A3B2 S19A1B2	1 2	CCW-R DEADFACED AT CM SM SEPARATION AND LES ABORTS	12 22	C1983 C1984	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A/MMB R.C. 1 MM A/MMB R.C. 1	I-P I-P	D	CCW-R CCW-R	D2 B2	S19A4B4 S19A2B4	1 2	CCW-R DEADFACED AT CM SM SEPARATION AND LES ABORTS	22 23	C1984 C1985	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A R.C. 1 MM A R.C. 1	I-P I-P	C	CCW-R CCW-R	C2 A2	S19A3B2 S19A1B2	1 2	CCW-R DEADFACED AT CM SM SEPARATION AND LES ABORTS	12 22	C1983 C1984	S2 & S1 S1 & S2
MM A MM B	CONTR/DIRECT RCS 1A CONTR/DIRECT RCS 1B	MM A/MMB R.C. 1 MM A/MMB R.C. 1	I-P I-P	D	CCW-R CCW-R	D2 B2	S19A4B4 S19A2B4	1 2	CCW-R DEADFACED AT CM SM SEPARATION AND LES ABORTS	22 23	C1984 C1985	S2 & S1 S1 & S2

SM CM RCS ENGINE POWER SUPPLIES DIRECT ROTATION CONTROL NO. 2

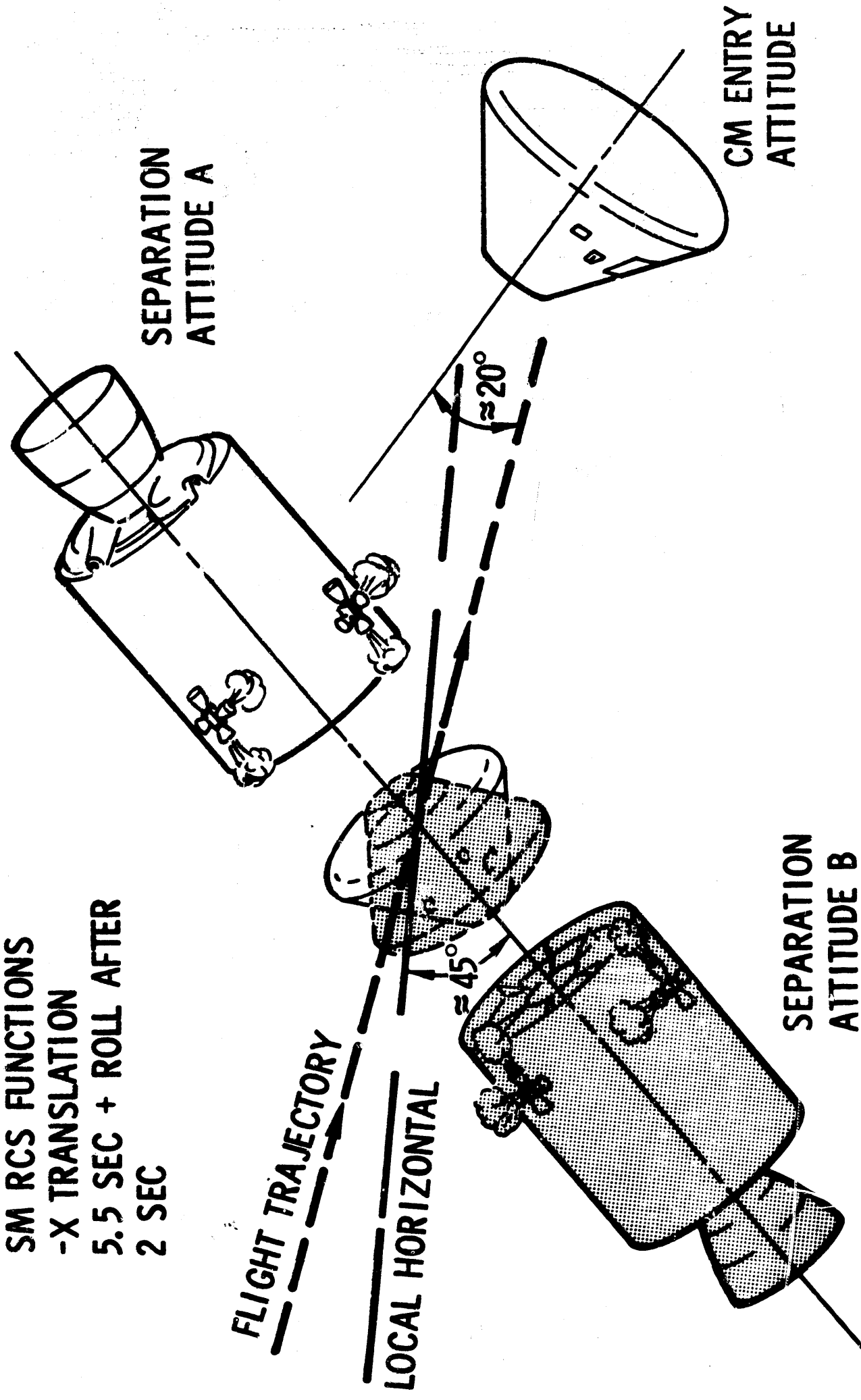
CSM 103 AND SUBS

BUS POWER	CIRCUIT BREAKER MDC-8	DIRECT 2 SWITCH MDC 1	ROTATION CONTROL	SM ENGINE				CM ENGINE				TRANSFER MOTOR								
				QUAD		MANEUVER		SCS NO.		PROP NO.			SYSTEM		MANEUVER		SCS NO.		PROP NO.	
MIN B	CONTR/DIRECT RCS 2B	MINA/MNB R.C. 2	2+P	C	+P	C3	S19A3B3	1	+P	13	C19B7	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MINA/MNB R.C. 2	2+P	A	+P	A3	S19A1B1	2	+P	23	C19B8	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MNB R.C. 2	2+P	C	+P	C3	S19A3B3	1	+P	13	C19B7	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MINA/MNB R.C. 2	2+P	A	+P	A3	S19A1B1	2	+P	23	C19B8	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MINA/MNB R.C. 2	2-P	A	-P	A4	S19A1B3	1	-P	14	C19B11	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MINA/MNB R.C. 2	2-P	C	-P	C4	S19A3B1	2	-P	24	C19B12	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MNB R.C. 2	2-P	A	-P	A4	S19A1B3	1	-P	14	C19B11	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MNB R.C. 2	2-P	C	-P	C4	S19A3B1	2	-P	24	C19B12	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MINA/MNB R.C. 2	2+Y	B	+Y	B3	S19A2B3	1	+Y	15	C19B9	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MINA/MNB R.C. 2	2+Y	D	+Y	D3	S19A4B1	2	+Y	25	C19B10	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MNB R.C. 2	2+Y	B	+Y	B3	S19A2B3	1	+Y	15	C19B9	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MNB R.C. 2	2+Y	D	+Y	D3	S19A4B1	2	+Y	25	C19B10	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MINA/MNB R.C. 2	2-Y	D	-Y	D4	S19A4B3	1	-Y	16	C19B1	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MINA/MNB R.C. 2	2-Y	B	-Y	B4	S19A2B1	2	-Y	26	C19B2	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MNB R.C. 2	2-Y	D	-Y	D4	S19A4B3	1	-Y	16	C19B1	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MNB R.C. 2	2-Y	B	-Y	B4	S19A2B1	2	-Y	26	C19B2	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MINA/MNB R.C. 2	2+R	C	CW+R CW+R	C1 A1	S19A3B4 S19A1B4	1 2	CW+R DEADFACED AT CM SM SEPARATION AND LES ABORTS	11 11	C19B5	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MINA/MNB R.C. 2	2+R	D	CW+R CW+R	D1 B1	S19A4B2 S19A2B2	1 2	CW+R DEADFACED AT CM SM SEPARATION AND LES ABORTS	21 21	C19B6	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MNB R.C. 2	2+R	C	CW+R CW+R	C1 A1	S19A3B4 S19A1B4	1 2	CW+R DEADFACED AT CM SM SEPARATION AND LES ABORTS	11 11	C19B5	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MNB R.C. 2	2+R	D	CW+R CW+R	D1 B1	S19A4B2 S19A2B2	1 2	CW+R DEADFACED AT CM SM SEPARATION AND LES ABORTS	21 21	C19B6	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MINA/MNB R.C. 2	2-R	C	CCW-R CCW-R	C2 A2	S19A3B2 S19A1B2	1 2	CCW-R DEADFACED AT CM SM SEPARATION AND LES ABORTS	12 12	C19B3	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MINA/MNB R.C. 2	2-R	D	CCW-R CCW-R	D2 B2	S19A4B4 S19A2B4	1 2	CCW-R DEADFACED AT CM SM SEPARATION AND LES ABORTS	22 22	C19B4	S1 & S2								
MIN B	CONTR/DIRECT RCS 2B	MNB R.C. 2	2-R	C	CCW-R CCW-R	C2 A2	S19A3B2 S19A1B2	1 2	CCW-R DEADFACED AT CM SM SEPARATION AND LES ABORTS	12 12	C19B3	S2 & S1								
MIN A	CONTR/DIRECT RCS 2A	MNB R.C. 2	2-R	D	CCW-R CCW-R	D2 B2	S19A4B4 S19A2B4	1 2	CCW-R DEADFACED AT CM SM SEPARATION AND LES ABORTS	22 22	C19B4	S1 & S2								

NORMAL CM-SM SEPARATION AND SM JETTISON

BLOCK II

SM RCS FUNCTIONS
-X TRANSLATION
5.5 SEC + ROLL AFTER
2 SEC



CM REACTION CONTROL SYSTEM

PURPOSE: PROVIDE THRUST FOR ATTITUDE
CONTROL AFTER CM/SM
SEPARATION FOR ENTRY

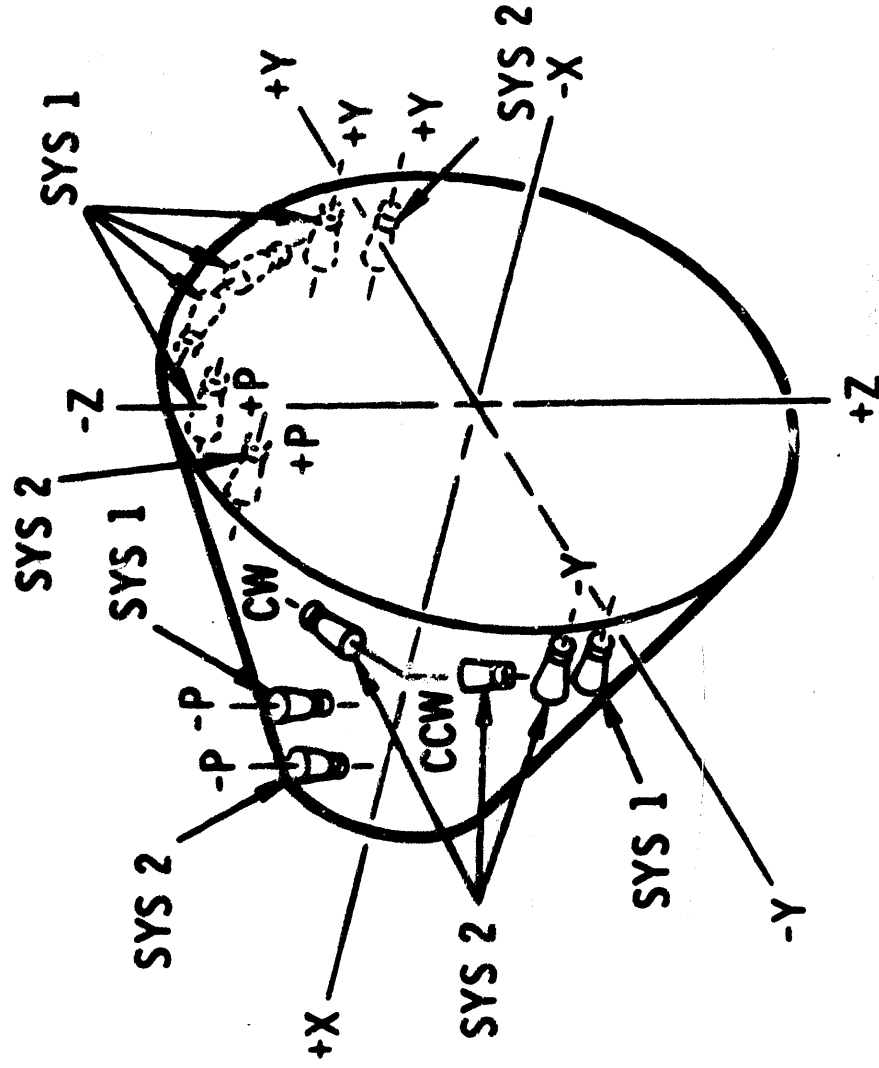
TYPE OF SYSTEM: NON THROTTLEABLE
PRESSURE-FEED SYSTEM,
HYPERGOLIC PROPELLANTS
AND ABLATIVE THRUST CHAMBERS

SYSTEM DESCRIPTION:

TWO INDIVIDUAL SYSTEMS
EACH SYSTEM CONSISTS OF A
SEPARATE PROPELLANT FEED SYSTEM
AND SIX THRUST CHAMBERS

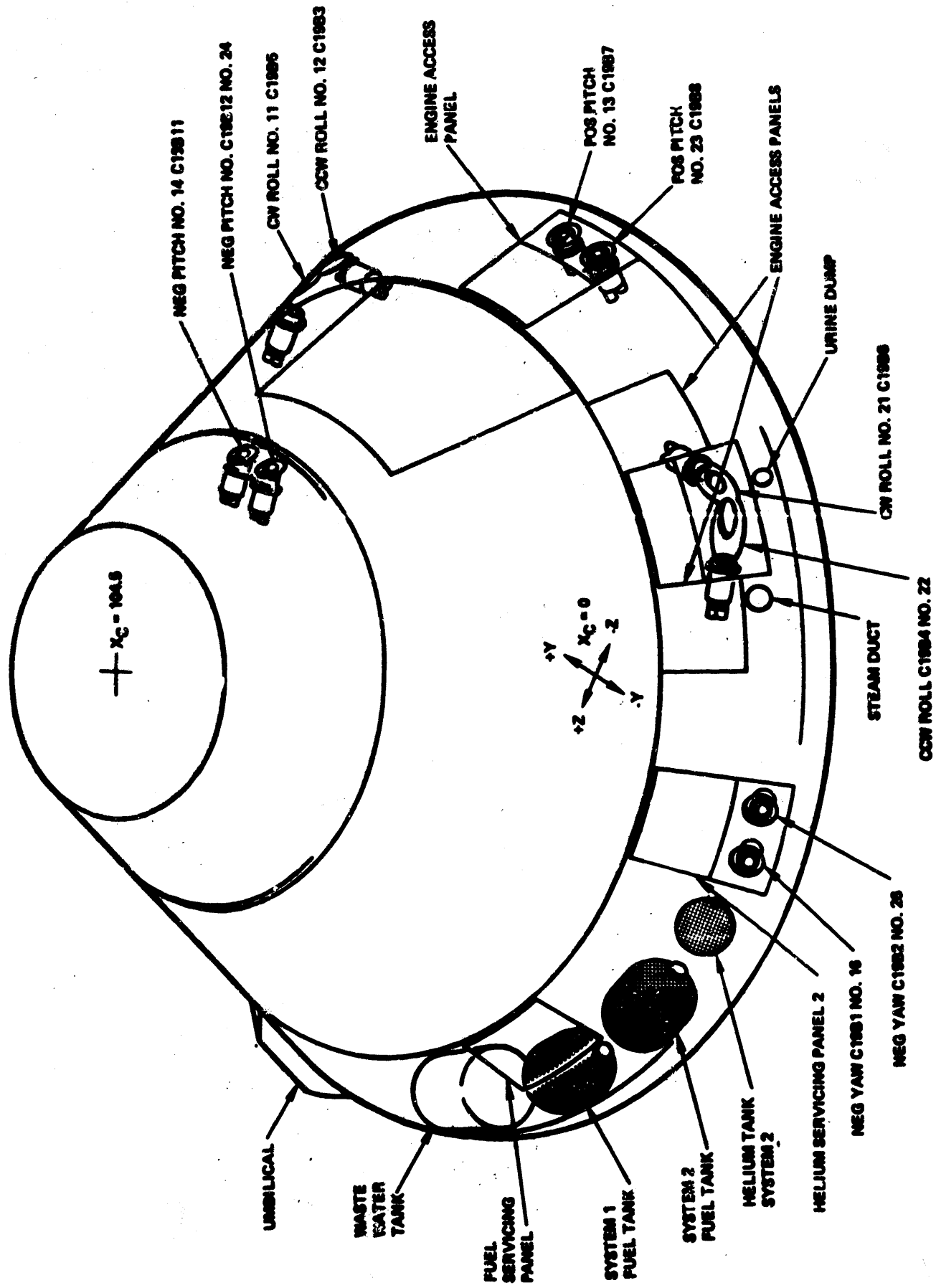
PROPELLANTS-INHIBITED NITROGEN
TETROXIDE AND MONOMETHYL
HYDRAZINE

SYSTEM CONTROLLED BY CMC OR
SCS WITH A DIRECT CONTROL
BACKUP CAPABILITY



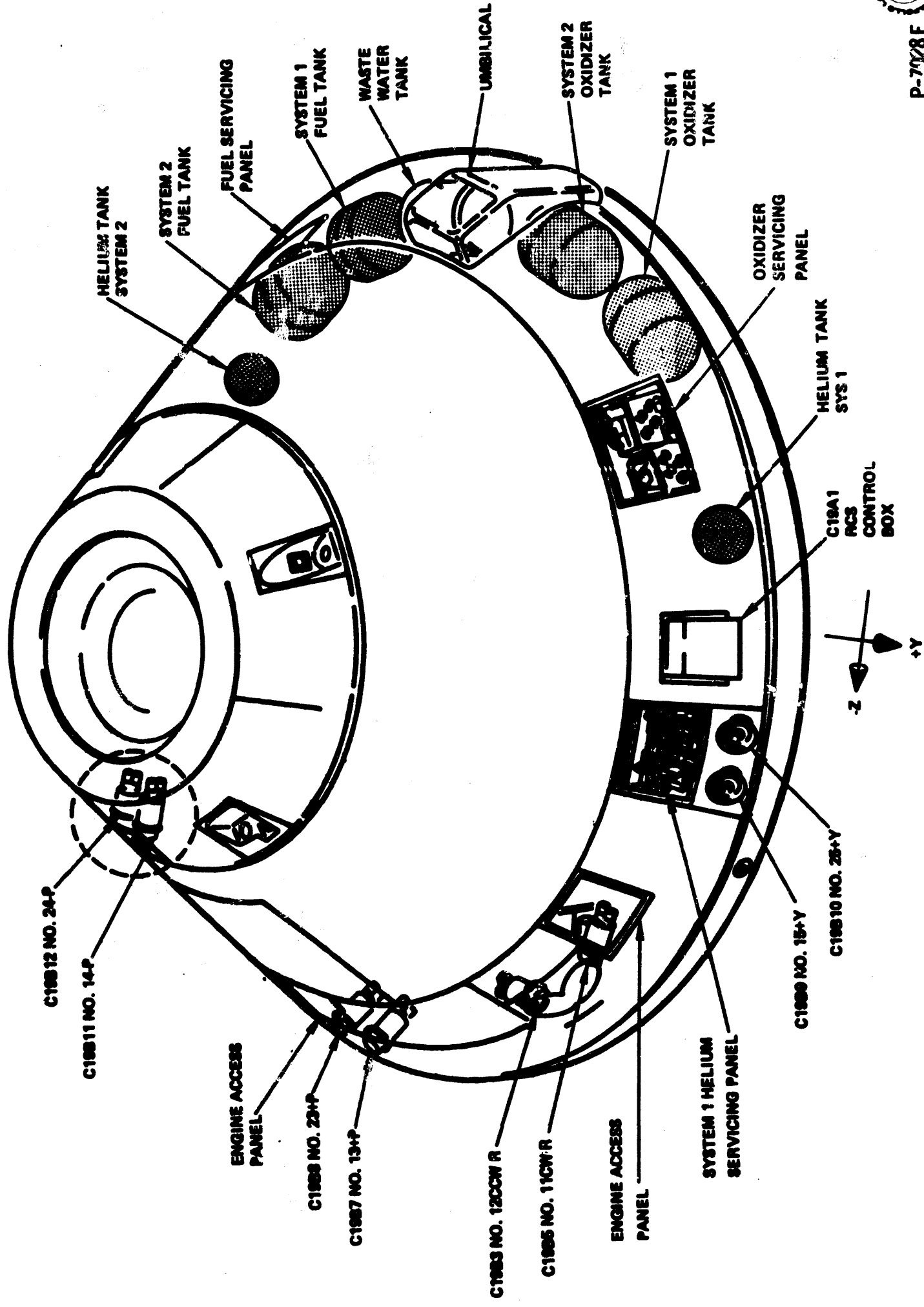
CMRCS COMPONENT LOCATION

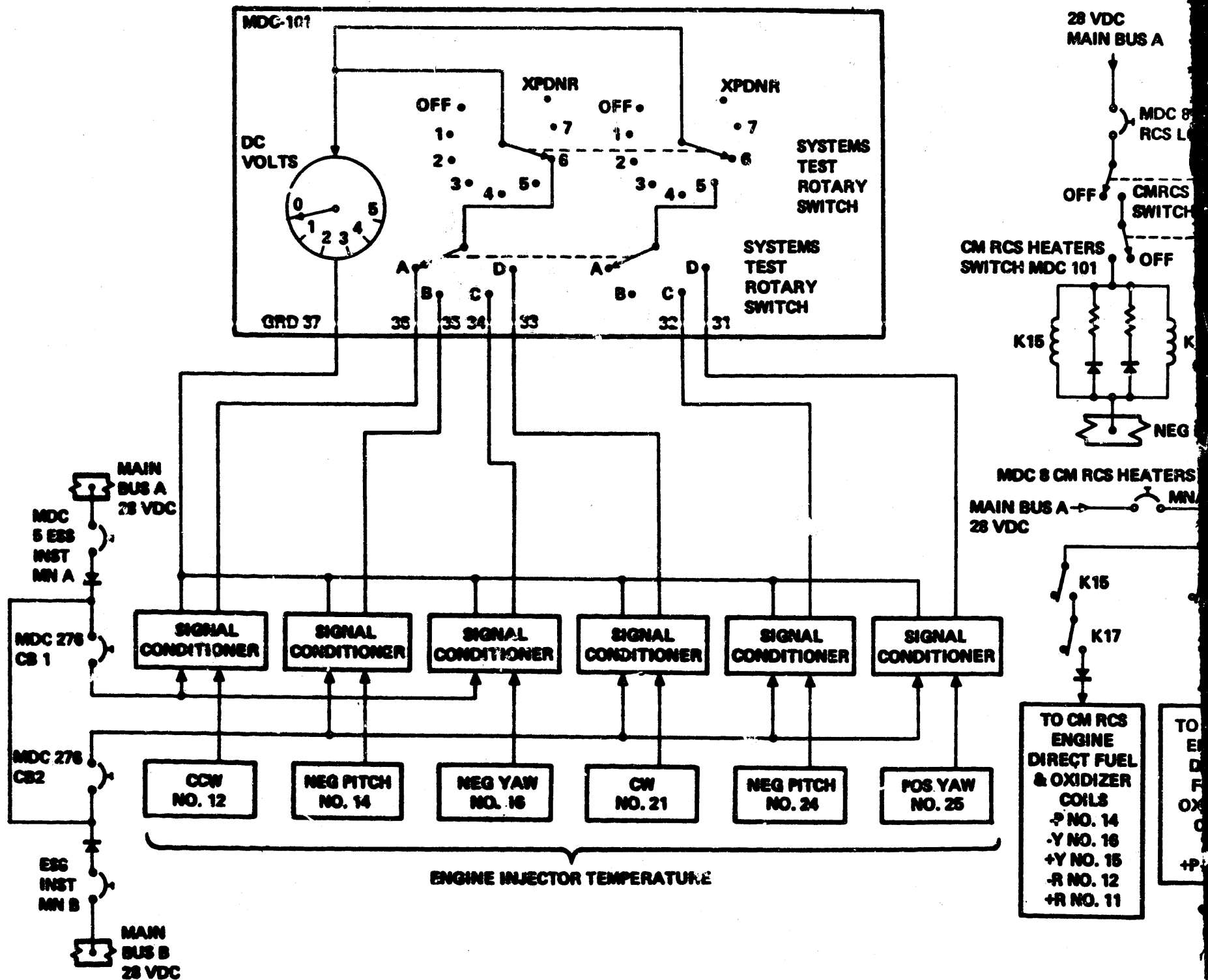
VIEW A

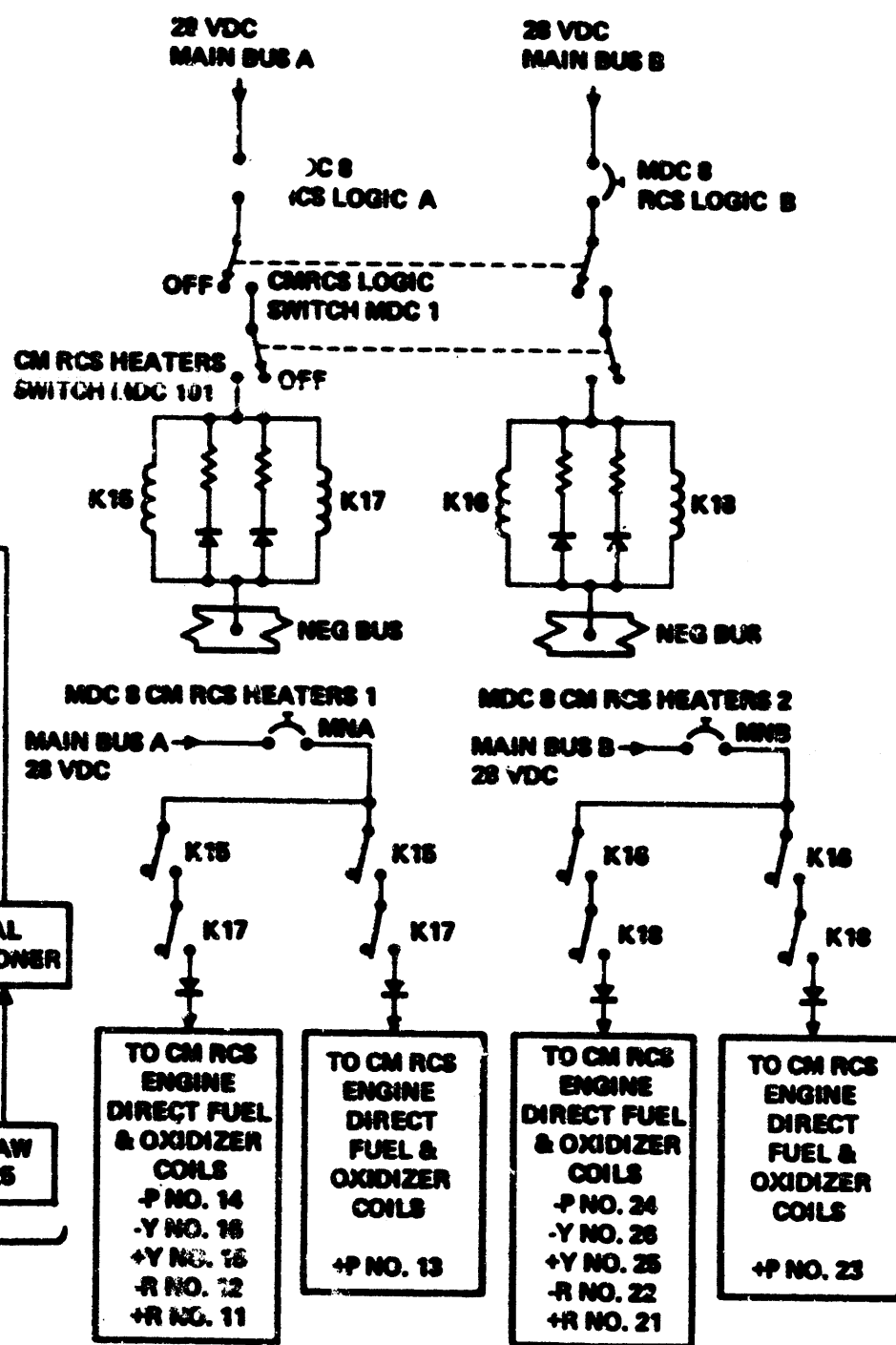
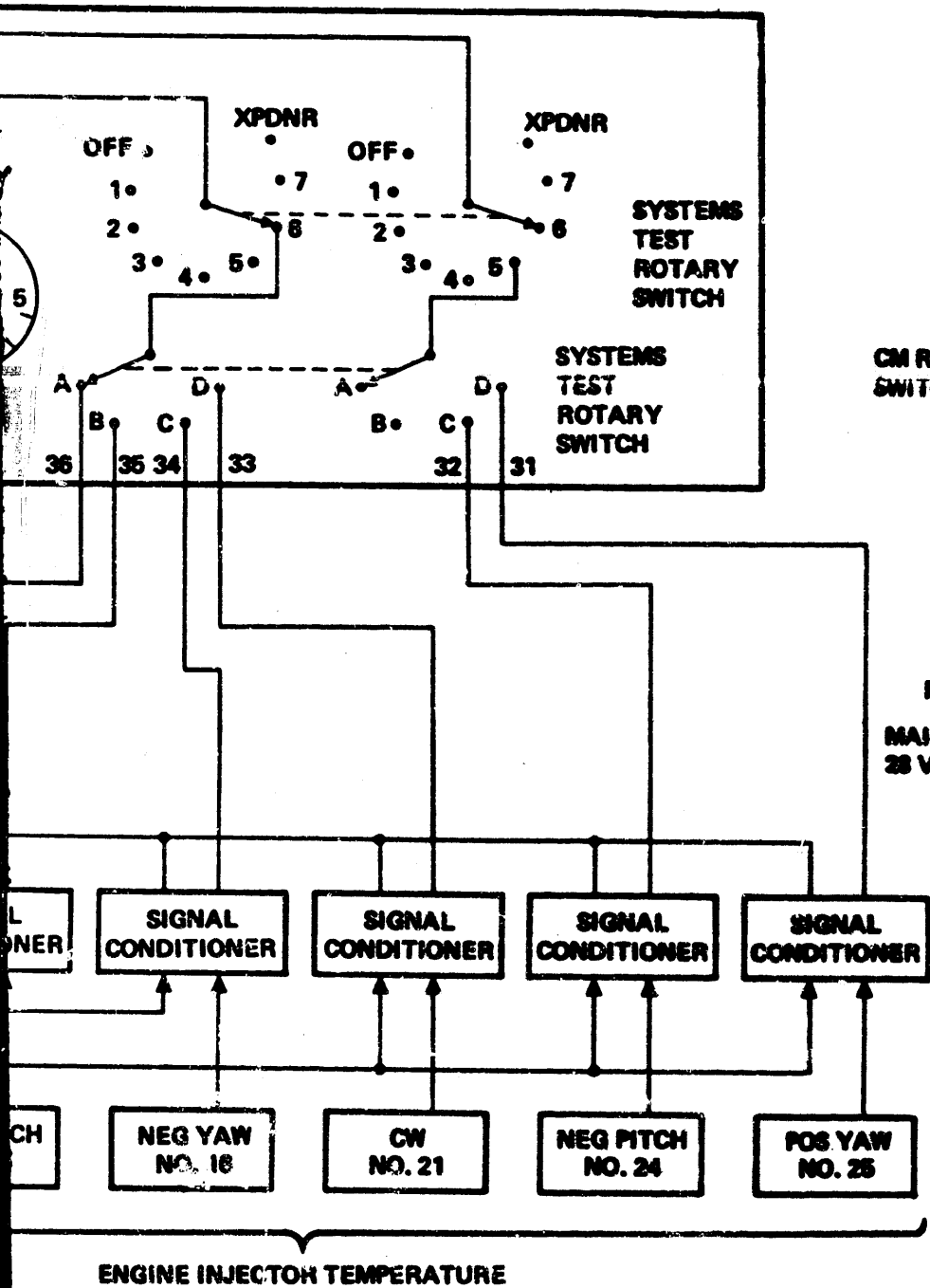


CMRCS COMPONENT LOCATION

VIEW B

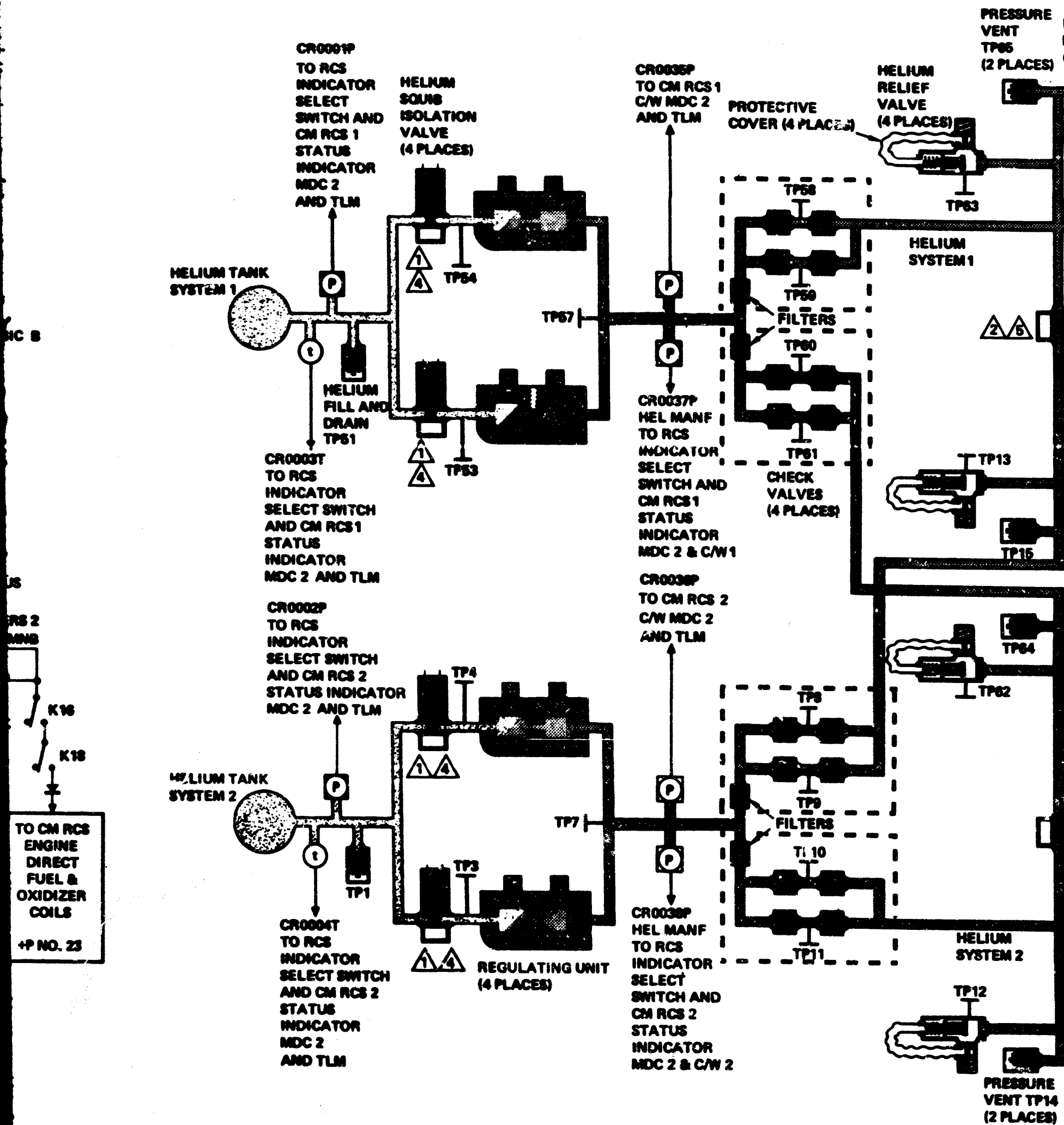


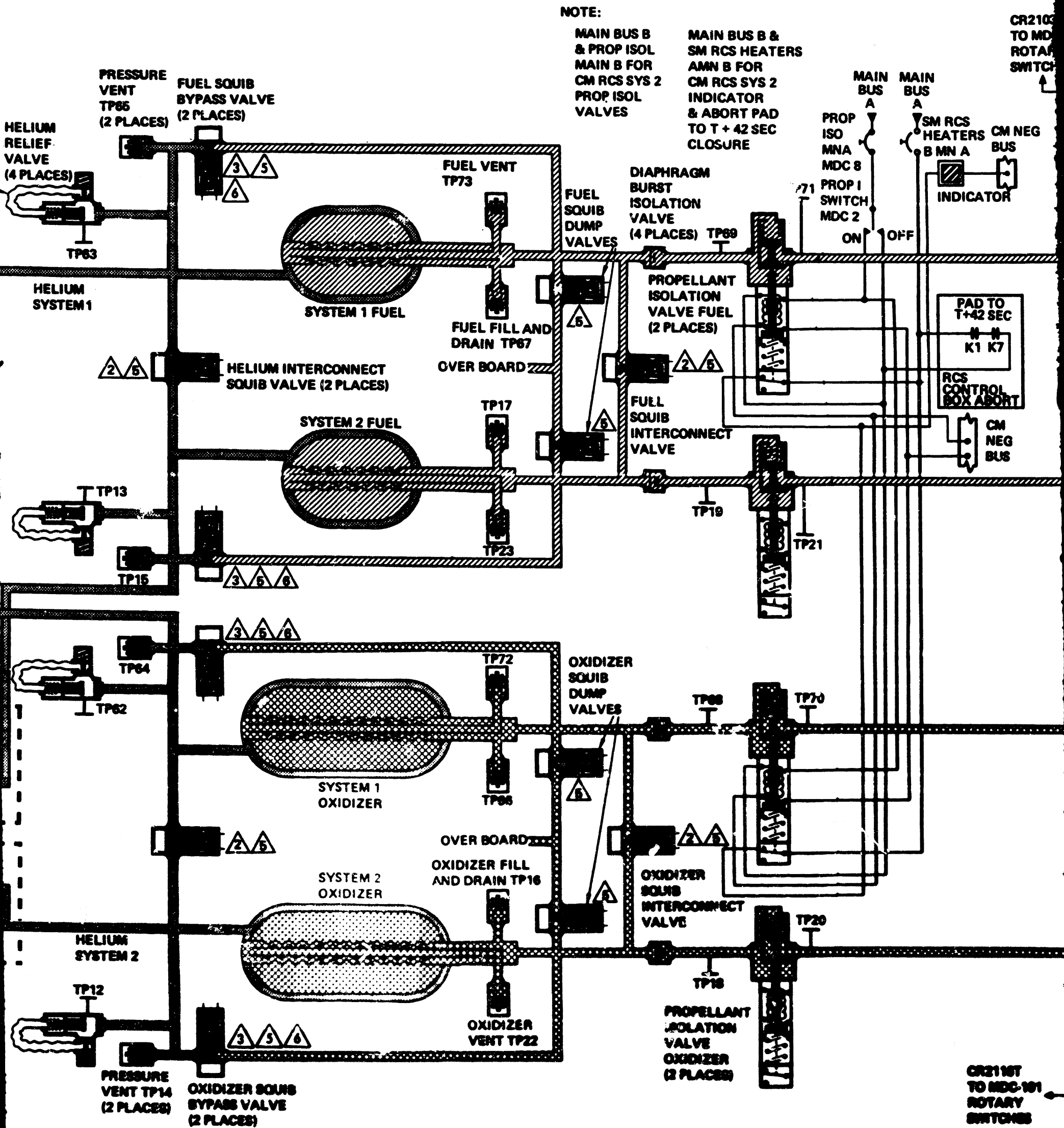


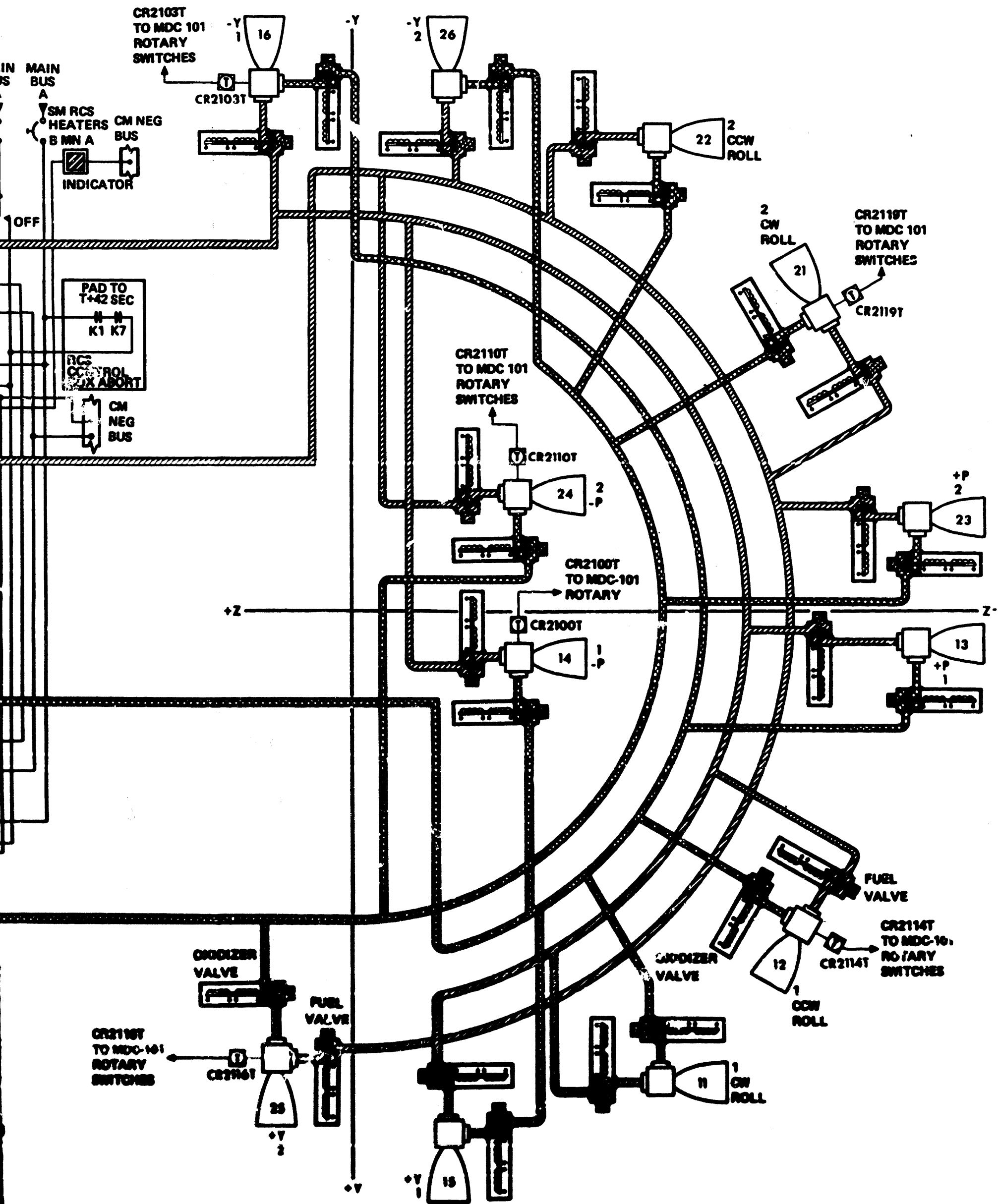


HELIUM
SYSTEM 1

HELIUM
SYSTEM 2

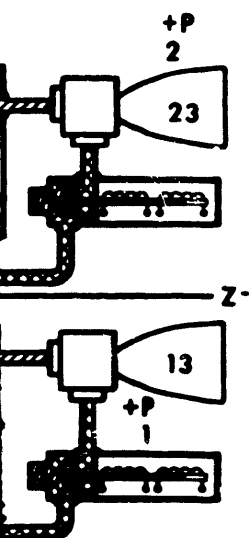






2119T
MDC 101
TARY
ITCHES

2119T



FUEL
VALVE

CR2114T
TO MDC-101
ROTARY
SWITCHES

PHASE	SEQUENCE	SQUIB VALVE INDICATION (TOTAL:				
		HELIUM ISOLATION (4)	FUEL SIDE HELIUM INTERCONNECT (1)	OXIDIZER SIDE HELIUM INTERCONNECT (1)	FUEL INTERCONNECT (1)	OXIDIZER INTERCONNECT (1)
NORMAL ENTRY	1. PRIOR TO CM SM SEPARATION	1 (NORMAL OR BACKUP) (BACKUP OR NORMAL) 4				
	2. MAIN PARACHUTE LINE STRETCH		2	2	2	2
	3. AFTER PROPELLANT DEPLETION					
ABORT PAD TO T + 42 SECONDS	1. SIMULTANEOUS WITH EDS OR MANUAL ABORT INITIATION	4 (NORMAL) (BACKUP) 7	5	5	5	5
	2. ABORT INITIATION PLUS 5 SECONDS.					
	3. 13 SECONDS AFTER FUEL DUMP INITIATION					
ABORT AFTER T + 42 SECONDS	1. SIMULTANEOUS WITH EDS OR MANUAL ABORT INITIATION.	4 (NORMAL) (BACKUP) 7				
	2. MAIN PARACHUTE LINE STRETCH		2	2	2	2
	3. AFTER PROPELLANT DEPLETION					
ALL SPS ABORTS	SAME AS NORMAL ENTRY					

- HELIUM
- OXIDIZER
- FUEL
- [P] PRESSURE TRANSDUCER
- [T] TEMPERATURE TRANSDUCER
- C/W CAUTION AND WARNING

CM RCS FUNCTIONAL FLOW
CSM 104 AND SUBS

SQUIB VALVE INDICATION (TOTAL: 16 VALVES)

HELIUM INDICATION (4)	FUEL SIDE HELIUM INTERCONNECT (1)	OXIDIZER SIDE HELIUM INTERCONNECT (1)	FUEL INTERCONNECT (1)	OXIDIZER INTERCONNECT (1)	FUEL TANK BYPASS (2)	OXIDIZER TANK BYPASS (2)	OXIDIZER DUMP (2)	FUEL DUMP (2)
HELIUM INDICATION (4) NORMAL BACKUP OR NORMAL 4								
	2	2	2	2				
					3 NORMAL BACKUP 6	3 NORMAL BACKUP 6		
HELIUM INDICATION (4) NORMAL BACKUP OR NORMAL 7	5	5	5	5			5	
								5
					5	5		
HELIUM INDICATION (4) NORMAL BACKUP OR NORMAL 7								
	2	2	2	2				
					3 NORMAL BACKUP 6	3 NORMAL BACKUP 6		

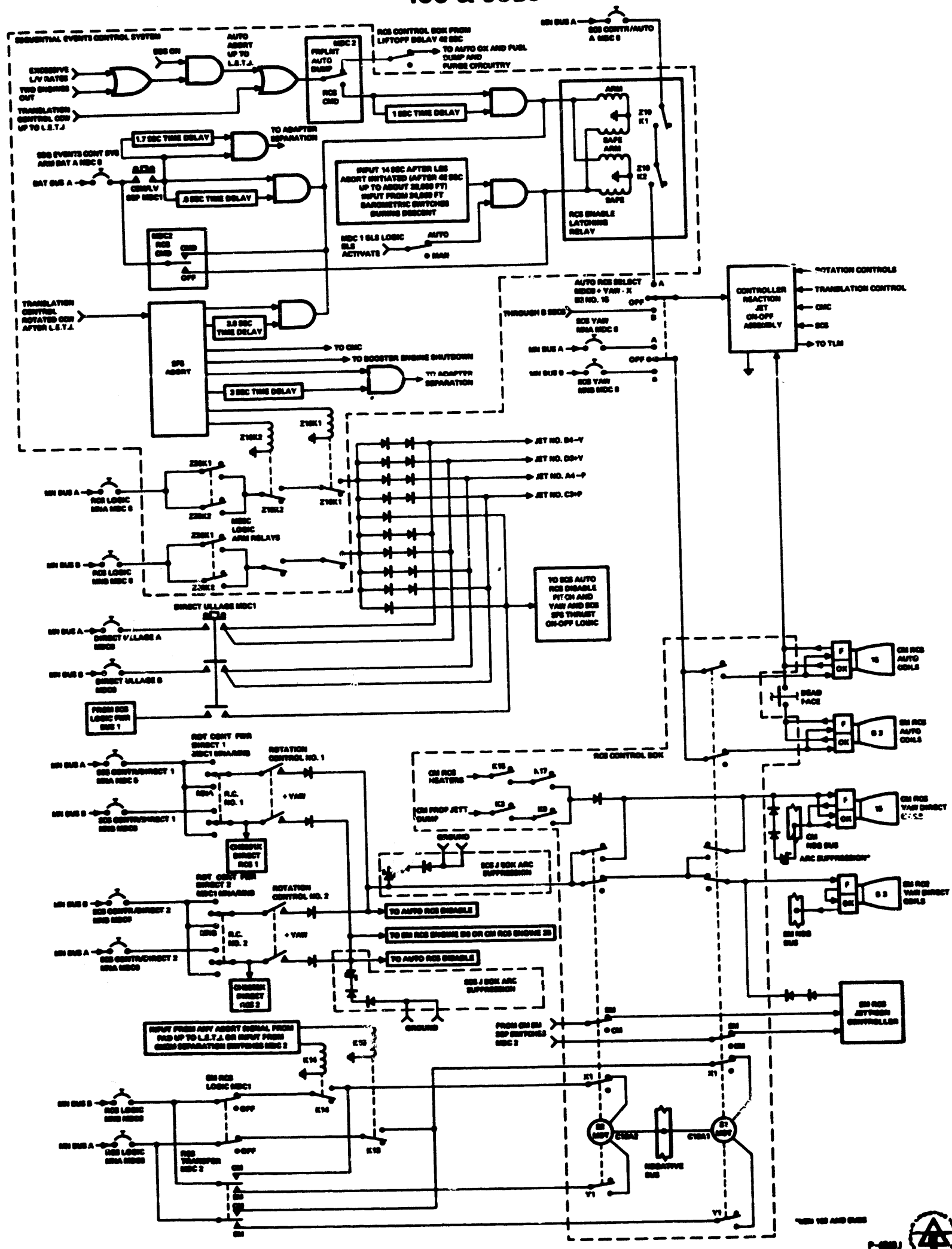
LEGEND FOR SQUIB VALVE ACTUATION

- 1 CM RCS PRESSURIZE, SWITCH MDC 2
- 2 CM PROPELLANT DUMP SWITCH MDC 1
- 3 CM PROPELLANT PURGE SWITCH MDC 1
- 4 AUTOMATIC (CM SM SEPARATION)
- 5 AUTOMATIC (RCS CONTROL BOX)
- 6 CM RCS HELIUM DUMP PUSHBUTTON, MDC 1

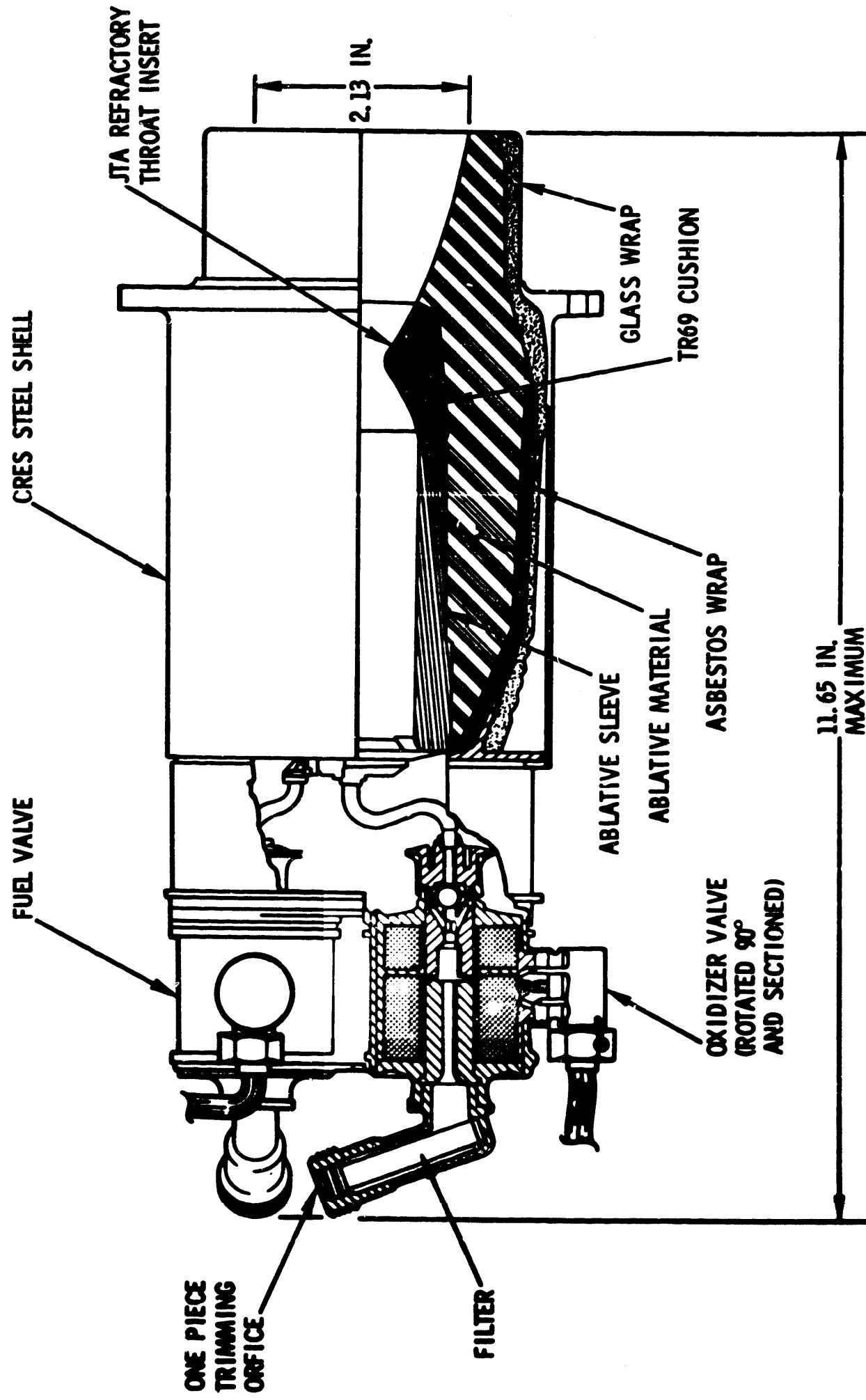
CM RCS FUNCTIONAL FLOW
CM 104 AND SUBS

RCS ELECTRICAL CSM CONTROL 103 & SUBS

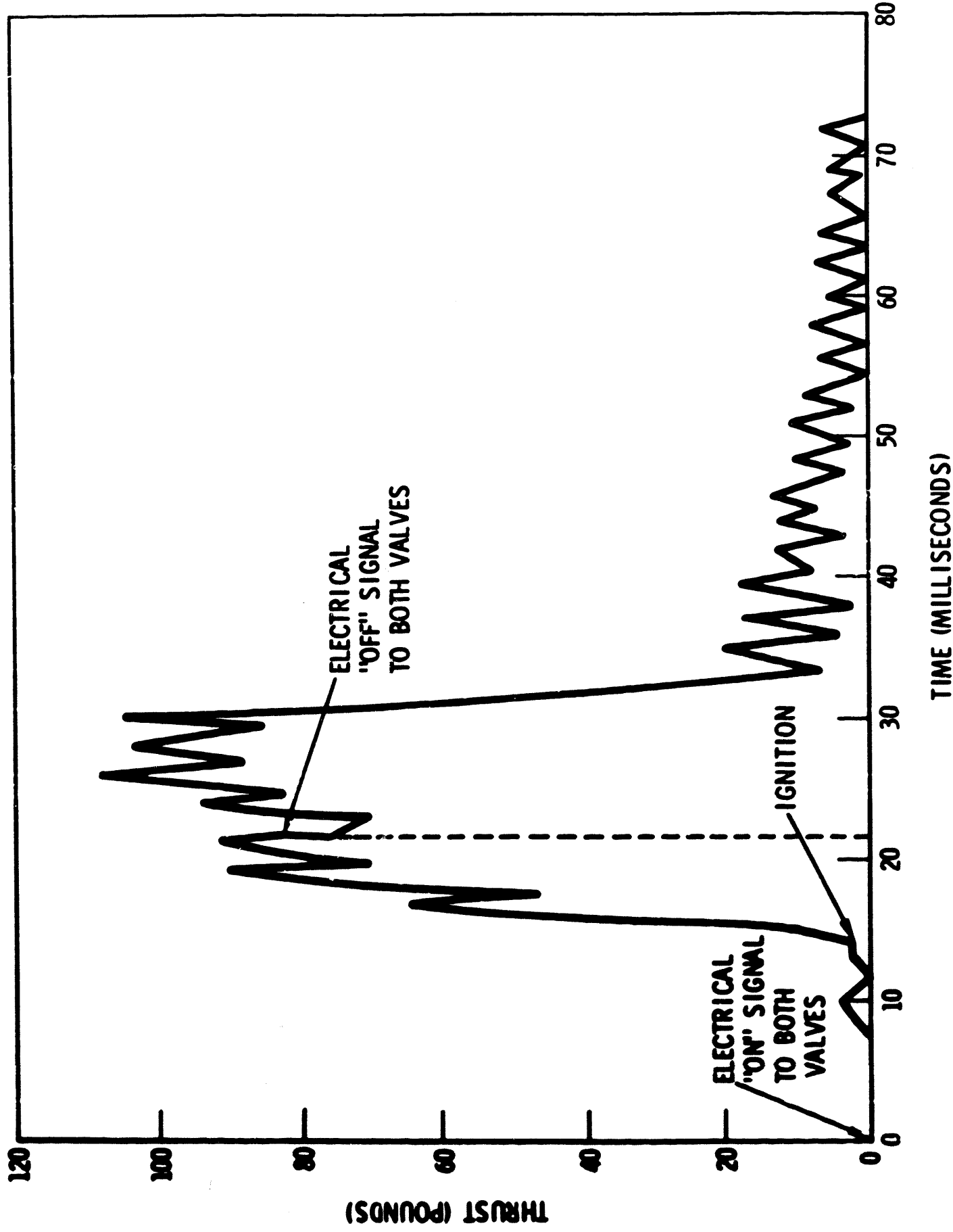
48



CM RCS ENGINE

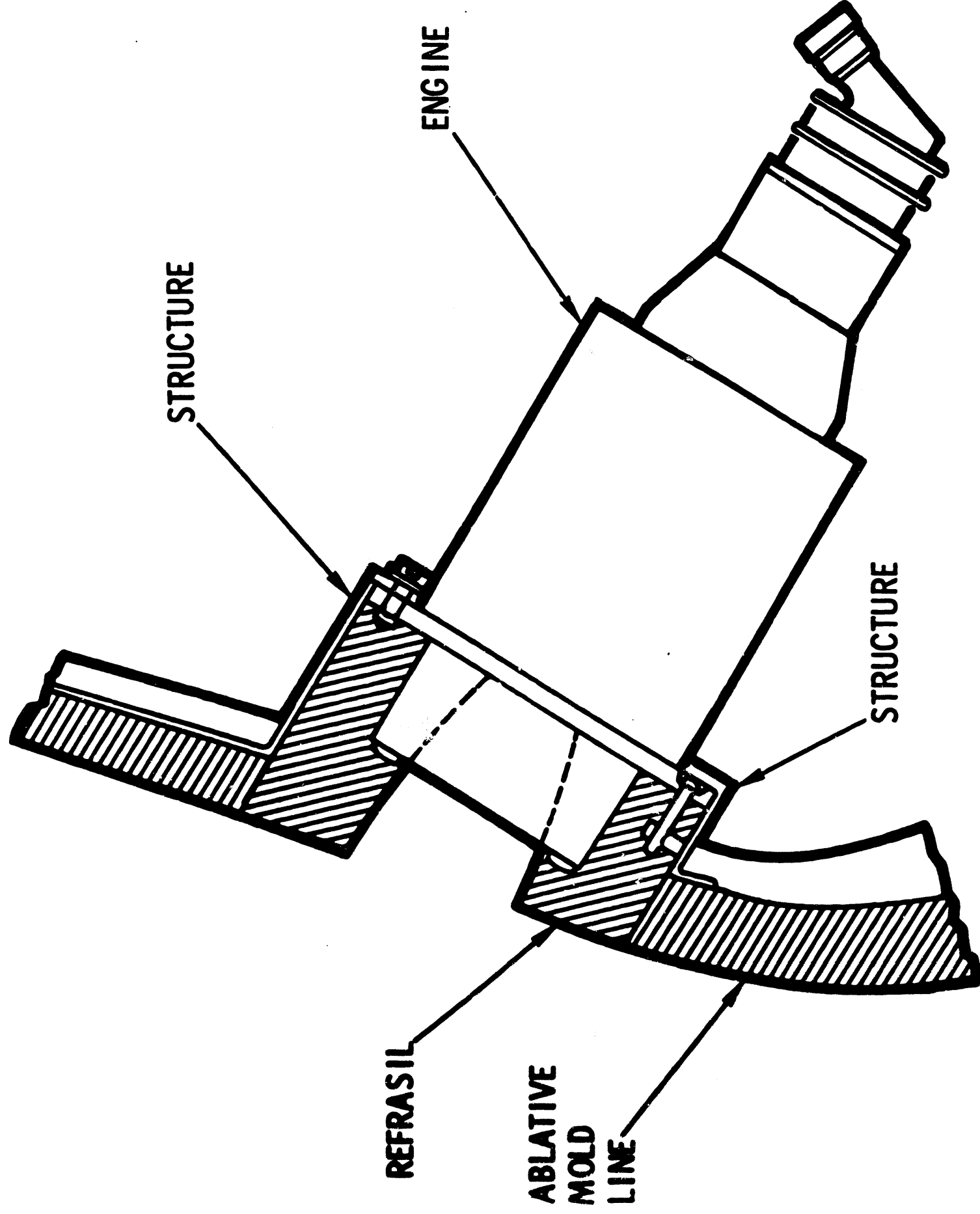


TYPICAL COMMAND MODULE RCS ENGINE THRUST

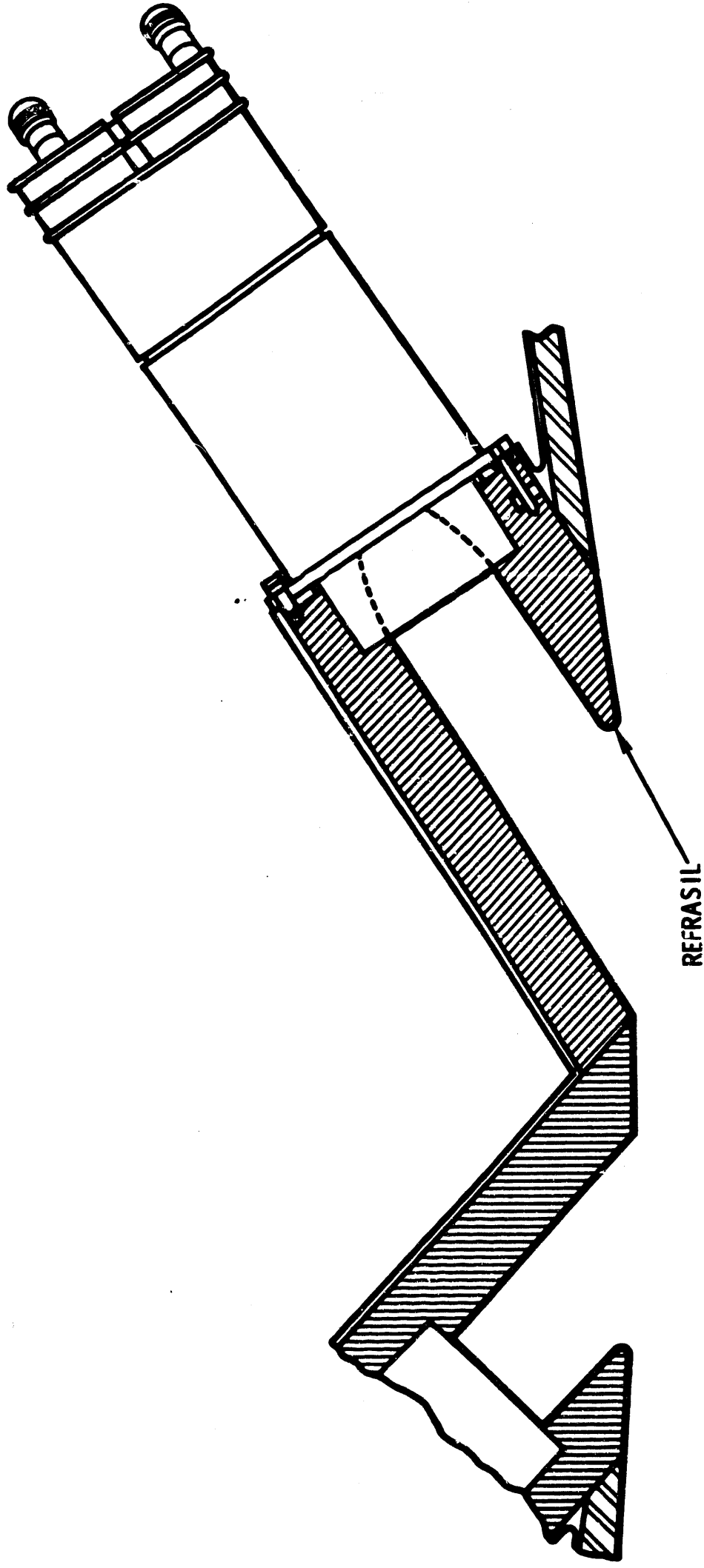


P-7031

C/M RCS TYPICAL NOZZLE EXTENSION PITCH AND YAW

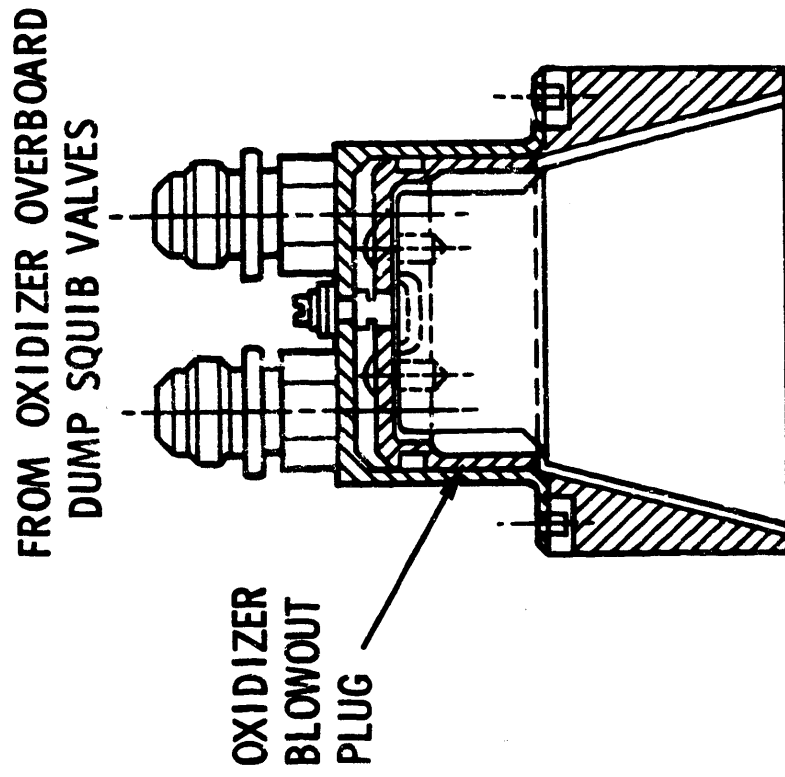
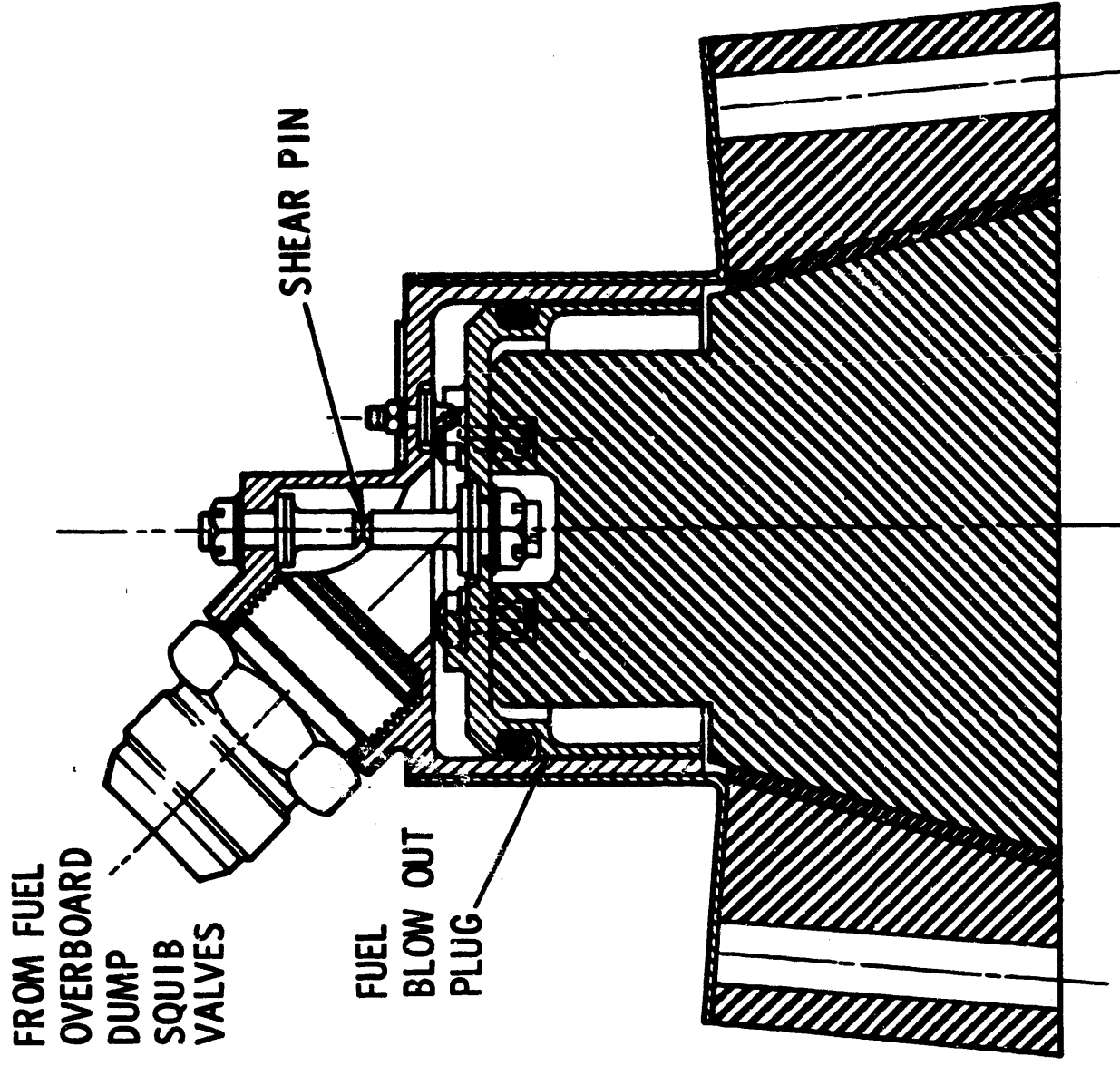
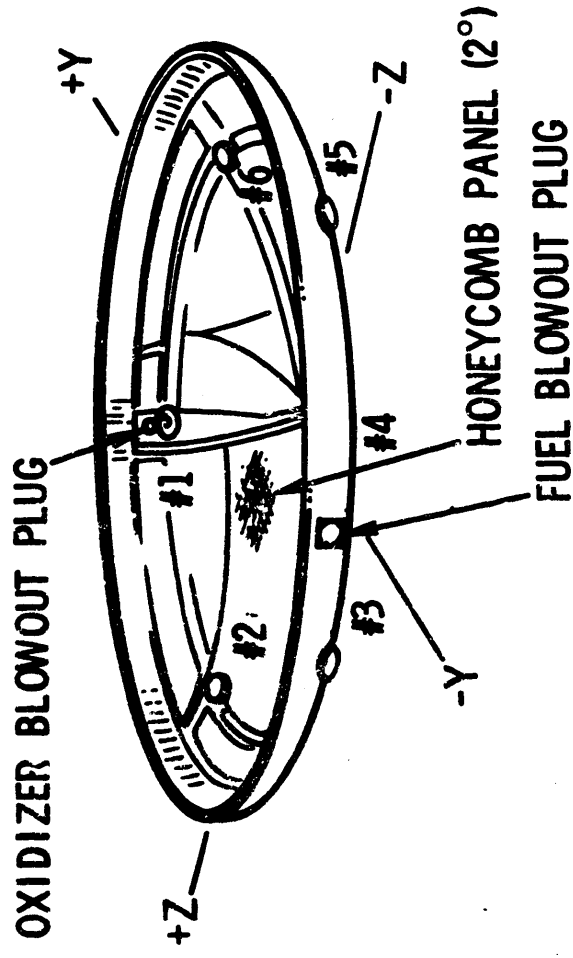


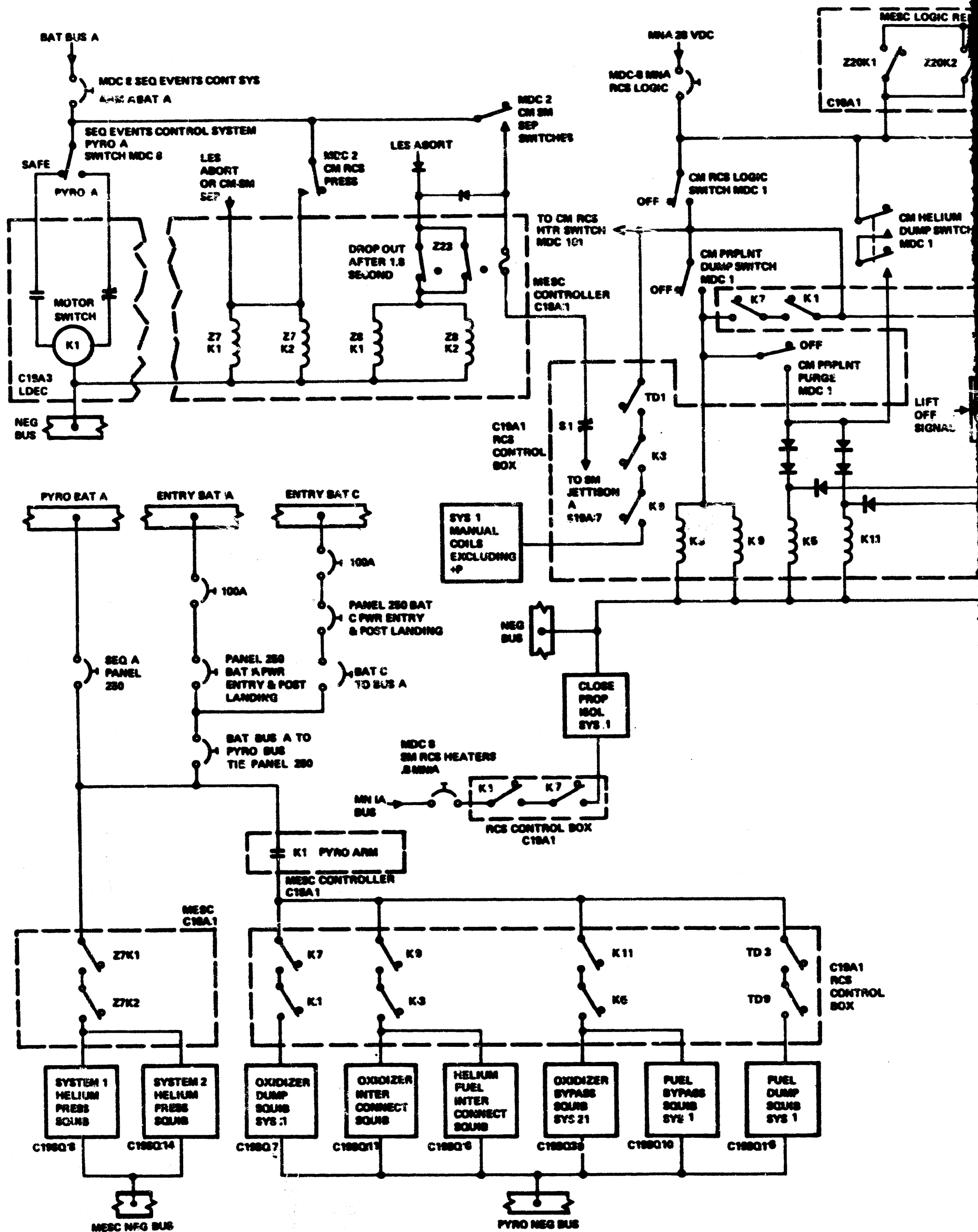
C/M RCS TYPICAL NOZZLE EXTENSION ROLL

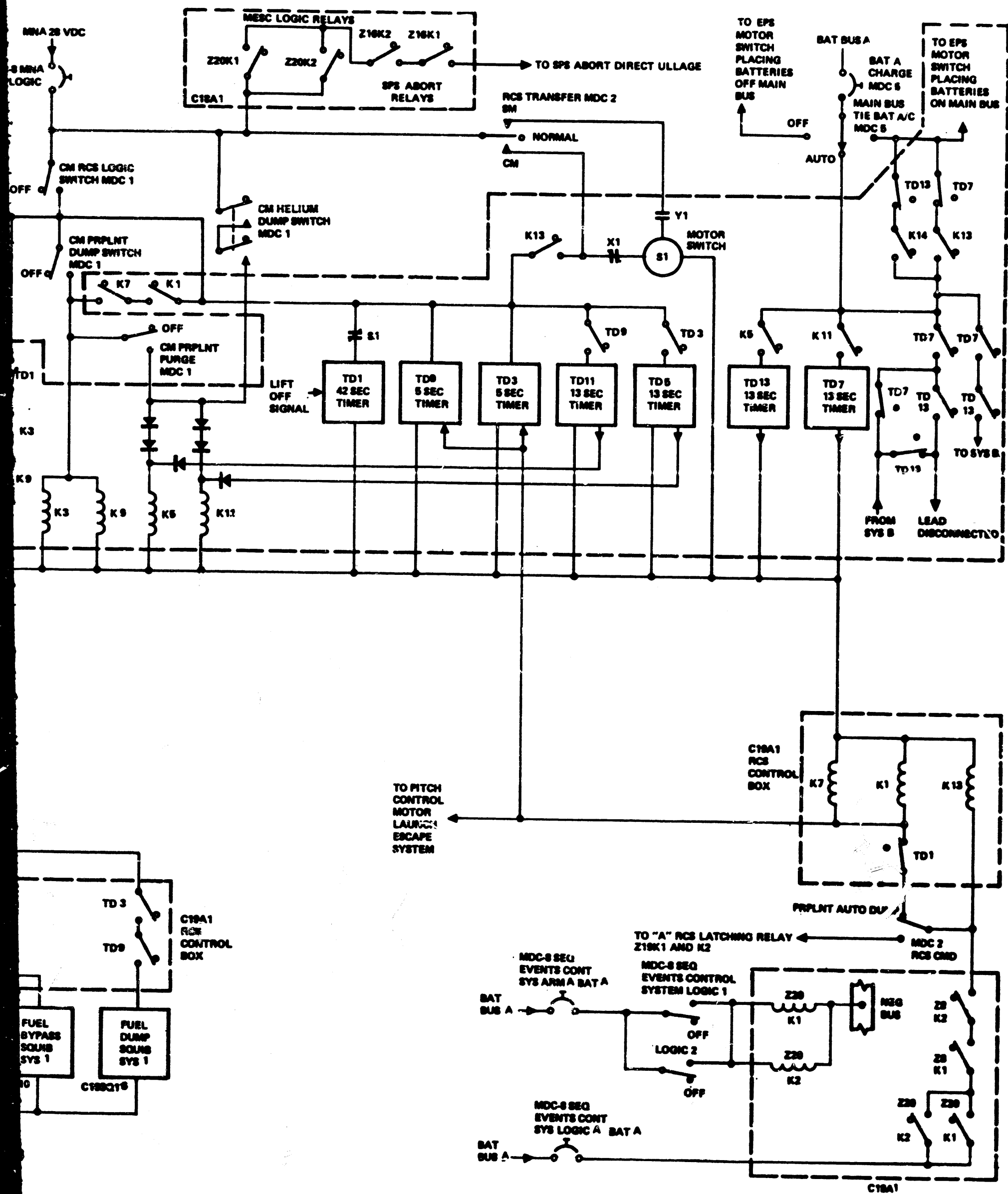


CMRCS RAPID FUEL AND OXIDIZER OVERBOARD DUMP PLUGS

BLOCK II









SM-2A-L240 T

SPS DATA

HELIUM TANKS (2)

3600±50 PSIA nominal fill pressure, 4400 maximum operating pressure, capacity 19.4 cubic ft. each. Inside diameter 40" and a wall thickness of .46". Weight 393 lbs. each.

REGULATOR UNITS (2)

Working regulator, (Class I) primary 186±4 PSIG, secondary 191±4 PSIG. Primary lockup 195 PSIG. Secondary lockup 200 PSIG. Inlet filter 10 micron nominal, 25 micron absolute.

(Class II) Normally locked-up (closed) regulator, primary 181±4 PSIG, secondary 191±4 PSIG. Primary lockup 195 PSIG. Secondary lockup 205 PSIG. Inlet filter 10 micron nominal, 25 micron absolute.

CHECK VALVES - FILTERS

40 micron nominal and 74 micron absolute inlet port, test ports 50 micron nominal and 74 micron absolute, one at inlet to check valve assembly, one at each test port.

PRESSURE TRANSDUCERS (2)

Fuel and oxidizer under pressure setting (SPS PRESS LIGHT MDC-2) 157 PSIA. Fuel and oxidizer over pressure setting (SPS PRESS LIGHT MDC-2) 200 PSIA.

PROPELLANT UTILIZATION

Increase position - Approximately 3% more than nominal flow

Normal position - Nominal flow

Decrease position - Approximately 3 - 1/2 % less than nominal flow

Response time, normal to increase or vice versa or normal to decrease or vice versa is 3.5 seconds.

QUANTITY SENSING SYSTEM

Indicators - Difference between actual quantity and total quantity indicated for each propellant not to exceed ± .35% of full tank plus +.35% of propellant remaining separately to total fuel and oxidizer separately.

TLM - Difference between actual quantity on each tank and that represented to TLM be within ±.35% of full tank plus +.35% of propellant remaining.

SPS DATA

HELIUM RELIEF VALVE (2)

Diaphragm rupture 219±6 PSIG
 Filter, 10 micron, 25 micron absolute
 Relief valve relieves at 212 psig minimum to 225 psig maximum
 Relief valve reseats at 208 PSIG minimum
 Flow capacity 3 lbs/minute maximum at 60°F and 225 PSIG
 Bleed device closes when increasing pressure reaches no greater than
 150 PSIG in cavity and reopens when decreasing pressure has reached
 no less than 20 PSIG.

OXIDIZER STORAGE TANK #1

Total tank capacity 11,284.69 lbs.
 Fill pressure 110 PSIA
 Height 154.47"
 Inside diameter 45"
 Wall thickness .054"
 128.52 cubic feet

OXIDIZER SUMP TANK #2

Total tank capacity 13,923.72 lbs. = 57%
 Fill pressure 110 PSIA
 Height 153.8"
 Inside diameter 51"
 Wall thickness .054"
 161.48 cubic feet

FUEL STORAGE TANK #1

Total tank capacity 7058.36 lbs.
 Fill pressure 110 PSIA
 Height 154.47"
 Inside diameter 45"
 Wall thickness .054"
 128.52 cubic feet

FUEL SUMP TANK #2

Total tank capacity 8708.10 lbs. = 57%
 Fill pressure 110 PSIA
 Height 153.8"
 Inside diameter 51"
 Wall thickness .054"
 161.48 cubic feet

SPS DATA

TOTAL PROPELLANT (IN TANKS)

OXIDIZER 25,208.41 lbs = 103.4%
FUEL 15,766.46 lbs = 103.4%

99.9% oxidizer gaugeable = 24,389.10 lbs.
99.9% fuel gaugeable = 15,252.70 lbs.

ALL PROPELLANT TANKS

Pressurized to 10 ± 5 PSIG of helium when empty to prevent collapsing of tanks (negative pressure of .5 PSIG will collapse tanks).

INTERFACE FLANGE FILTER (2)

500 micron absolute

GN₂ BI-PROPELLANT VALVE CONTROL SYSTEMS (2)

GN₂ vessel pressure 2500 ± 50 PSI at 68°F, 2900 PSI at 130°F, I.D. 4.65", Length 9.6", 5.8 cubic inch capacity each.
Support 43 valve actuations
Regulator - Single stage, dynamic 187 PSIG minimum, lockup 195 to 225 PSIG
Relief Valve - Relieves at 350 ± 15 PSI
Reseats at not less than 250 PSI

HEATERS CSM 106 AND SUBS

*GN₂ filters, one between each GN₂ supply tank and injector pre-valve, 5 micron nominal and 18 micron absolute. One at each GN₂ regulator outlet test port, 5 micron nominal and 18 micron absolute.

6 heaters, 2 elements in each heater, 3 elements in series on the fuel side rated at 15 watts, 9.4 watts and 18.8 watts, 3 elements in series on the oxidizer side rated at 15 watts, 9.4 watts and 18.8 watts.
SPS heater switch position A/B on MDC-3 supplies 28 VDC to 12 elements.
SPS heater switch position A on MDC-3 supplies 28 VDC to 6 elements.

*CSM 108 and subs

ENGINE (I)

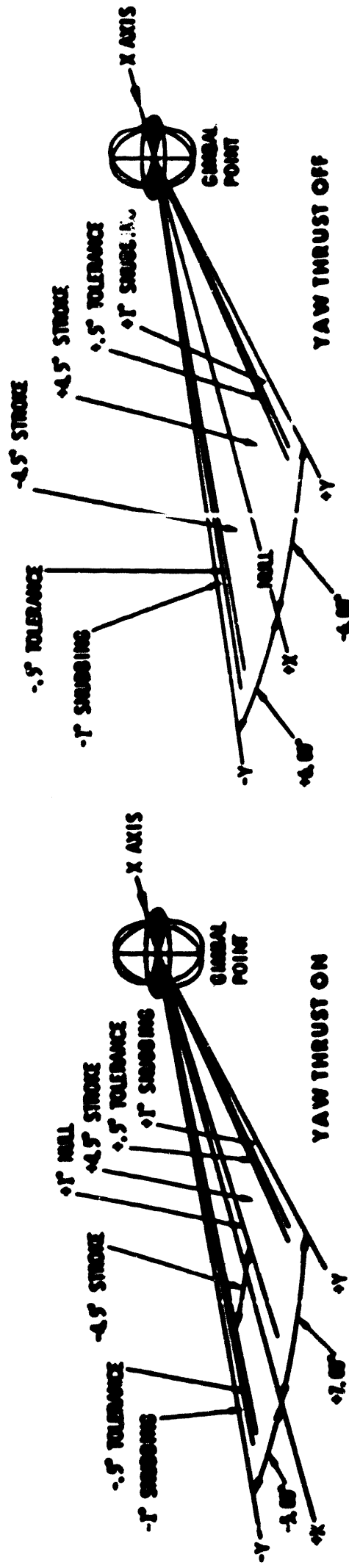
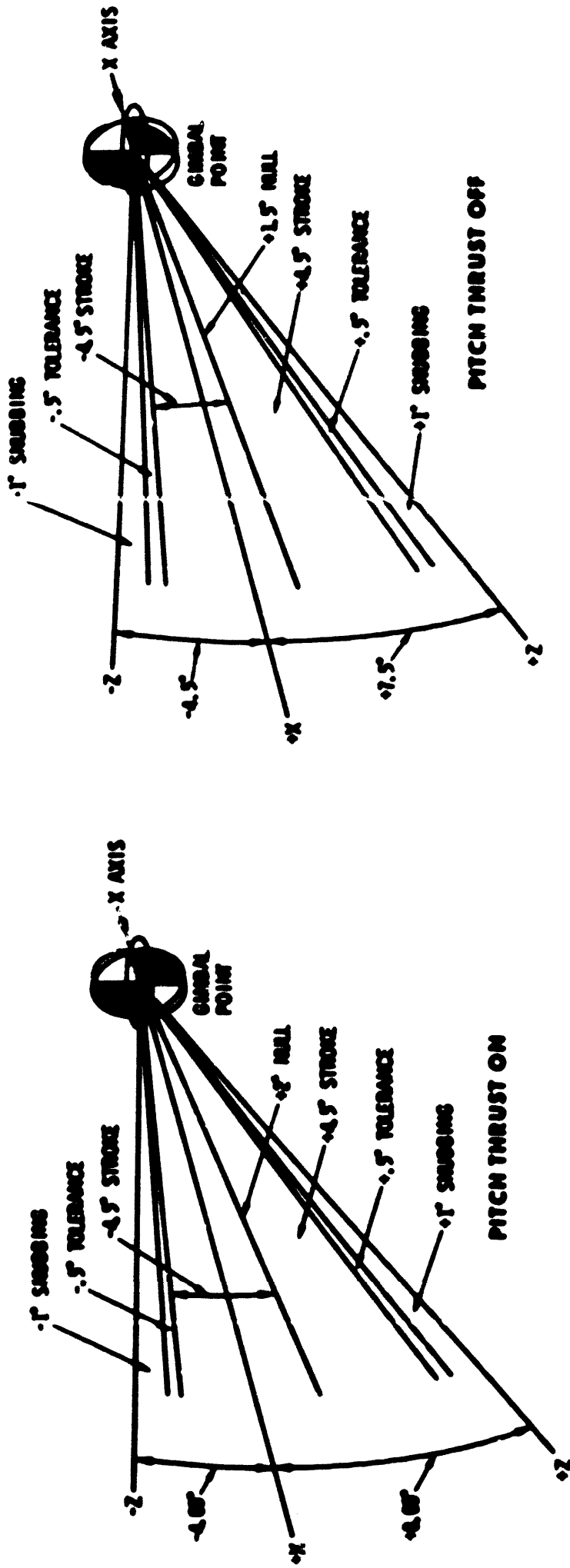
SPS DATA (Cont)

750 second service life
Support 36 Restarts Minimum
Expansion ratio = 6 to 1 at ablativ chamber exit area
= 62.5 to 1 at nozzle extension exit area
Chamber cooling = Ablation and film cooled
Nozzle extension = Radiation cooled
Injector type = Baffled, unlike impingement
Oxidizer lead 8°
Length 159.944" maximum
Nozzle extension exit diameter 98.4" Inside Diameter
Weight - Approximately 650 lbs.
Injector Flange Temperature - Illuminate SPS FLANGE TEMP HI
(Caution and Warning on MDC-2) at 480°F
SPS P_c transducer, displayed on MDC-1 through SPS P_c or switch to
SPS P_c or on MDC-1. Green range on indicator 65 to 125%
(corresponds to PSIA). Normal 95 to 105 PSIA.

GIMBAL ACTUATORS

Structural mounting pad offset 4° to $+Y$
About Z-Z axis $\pm 4\ 1/2^{\circ}$ with additional 1° for snubbing, (yaw) null
 1° to $+Y$ thrust vector during engine thrusting.
About Y-Y axis $\pm 4\ 1/2^{\circ}$ with additional 1° for snubbing, (pitch) null
 2° to $+Z$ thrust vector during engine thrusting.
Overcurrent relays (4)
Overcurrent dependent upon temperature during start transient,
steady state and gimbal angle change.
Quiescent current of 60+10-5 milliamps
Pressurized to 3.0 to 5.0 PSI of dry air
Deflection rate .12 radians per second (low side) 6.87° per second to
.132 radians per second (high side) 7.56° per second.

SPS GIMBAL ANGLES PITCH AND YAW



THE ENGINE MOUNTING PADS IN THE SERVICE MODULE ARE CANTED 6° THRU VECTOR TO THE +Y AXIS OF THE SPACECRAFT.

HOWEVER THE ACTUATOR NULL ADJUSTMENTS ARE AS FOLLOWS:

PITCH

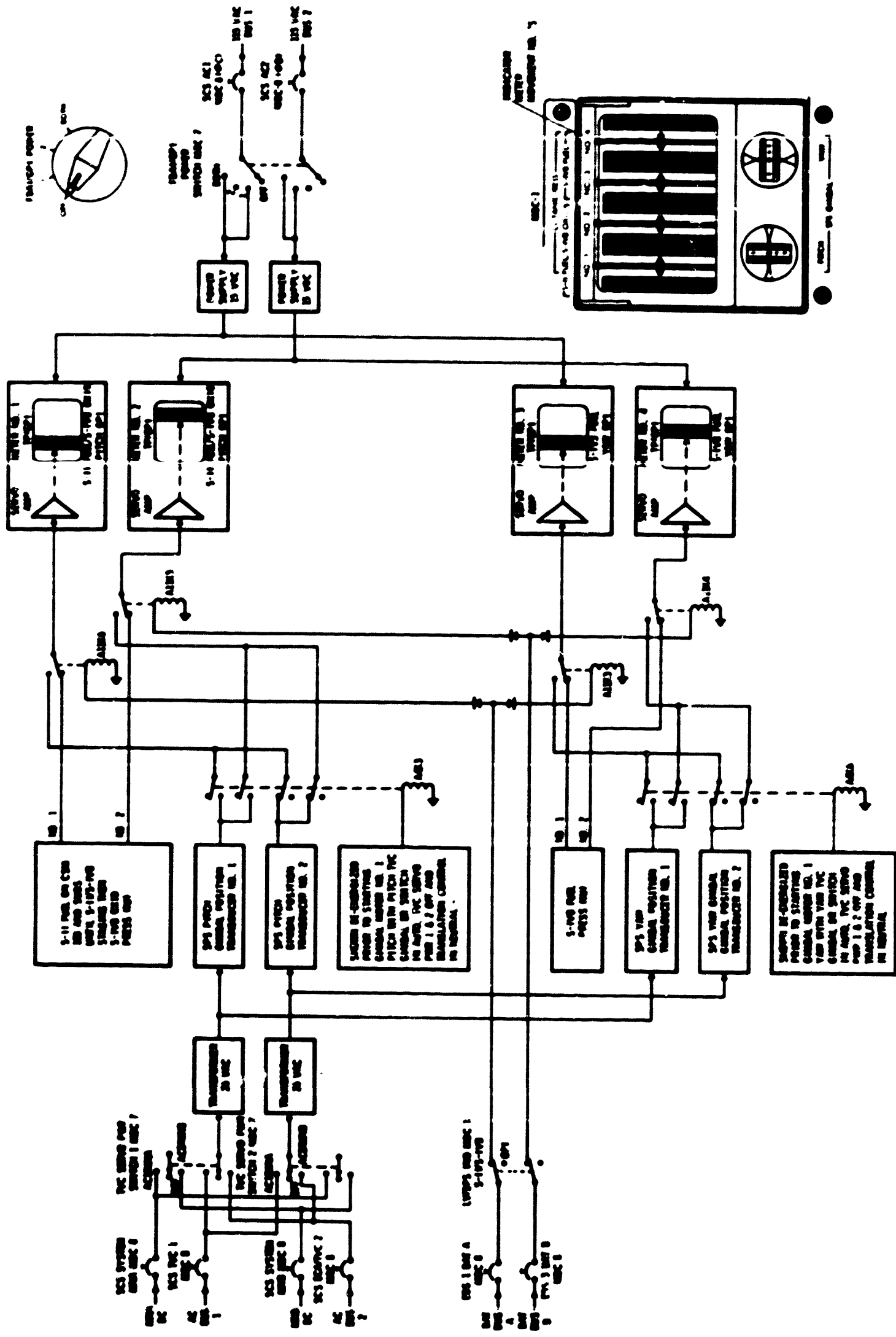
THE PITCH ACTUATOR NULL POSITION IS +2° THRU VECTOR TO THE +Z, ENGINE NOZZLE -2° TO THE -Z DURING AN SPS ENGINE FIRING. THE NULL POSITION IS +1.5° THRU VECTOR TO THE +Z, ENGINE NOZZLE -1.5° TO THE -Z DURING SPS NON-THRUSTING PERIODS. THE THUMB WHEEL ON MAC 1 & GIMBAL POSITION INDICATOR/FUEL PRESSURE INDICATOR ON MAC 1 WILL BE 0.0°.

YAW

THE YAW ACTUATOR NULL POSITION IS +1° THRU VECTOR TO THE +Y, ENGINE NOZZLE IS -1° TO THE -Y DURING AN SPS ENGINE FIRING. THE NULL POSITION IS 0° THRU VECTOR TO THE Y, ENGINE NOZZLE IS 0° TO THE Y AXIS DURING SPS NON-THRUSTING PERIODS. THE THUMB WHEEL ON MAC 1 & GIMBAL POSITION INDICATOR/FUEL PRESSURE INDICATOR ON MAC 1 WILL BE 0.0°.

THE REASON FOR THE DIFFERENCE BETWEEN A NON SPS ENGINE FIRING VERSUS AN SPS ENGINE FIRING IS DUE TO DEFLECTIONS OF ENGINE AND VEHICLE STRUCTURE.

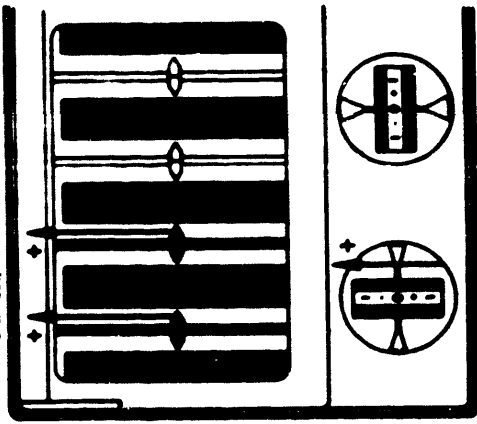
LV TANK PRESSURE/GPI INDICATOR POWER



SPS GIMBALLING

GPI AND TRIM THUMBWHEELS

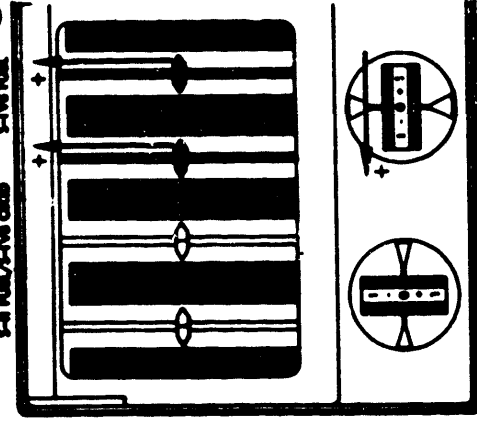
1-4 PUL/PUS TANK PRESS
5-8 PUL/PUS CRSD 9-16 PUL



PITCH

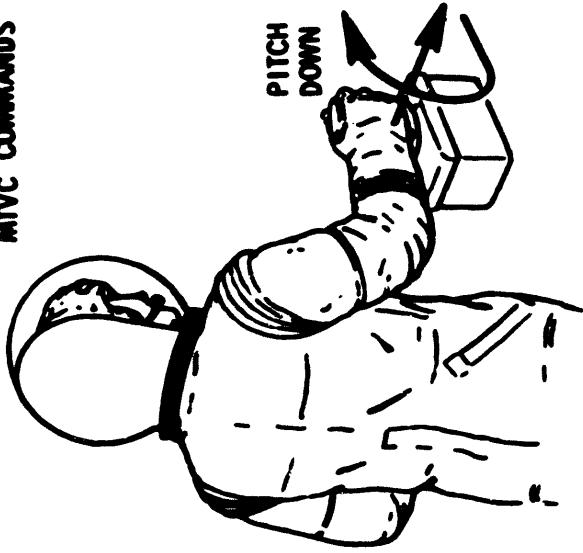
PITCH

1-4 PUL/PUS TANK PRESS
5-8 PUL/PUS CRSD 9-16 PUL

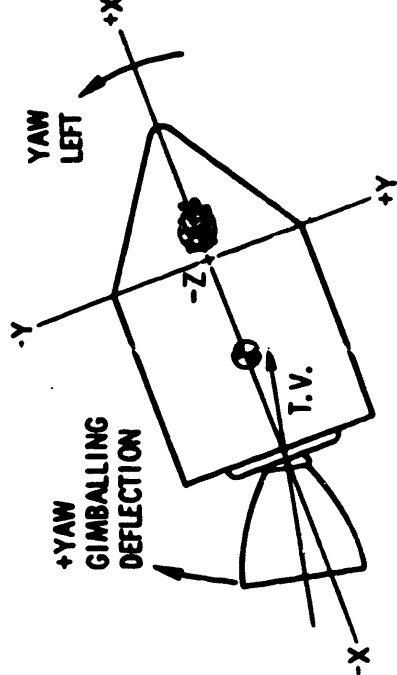
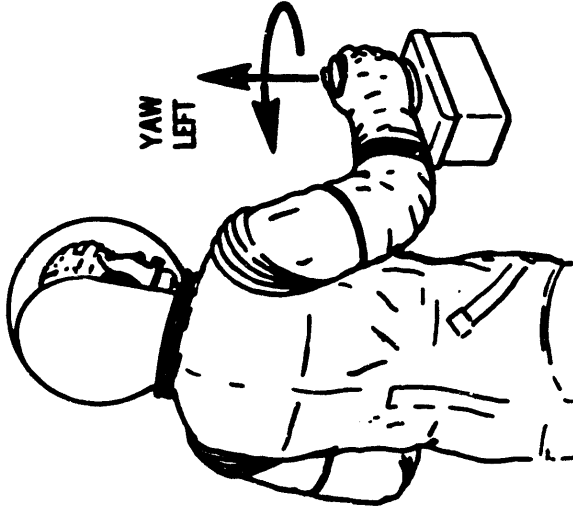
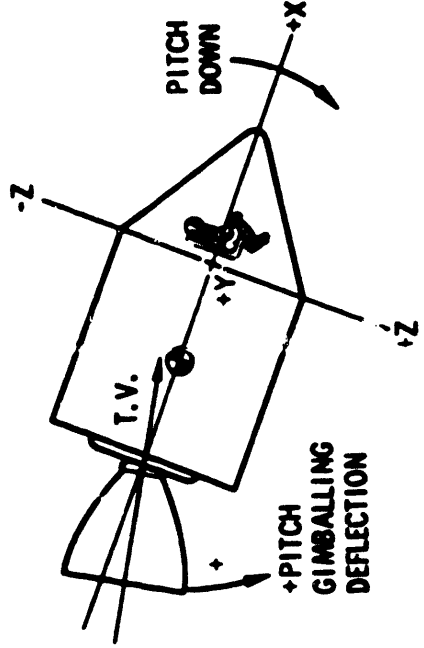


YAW

MTVC COMMANDS

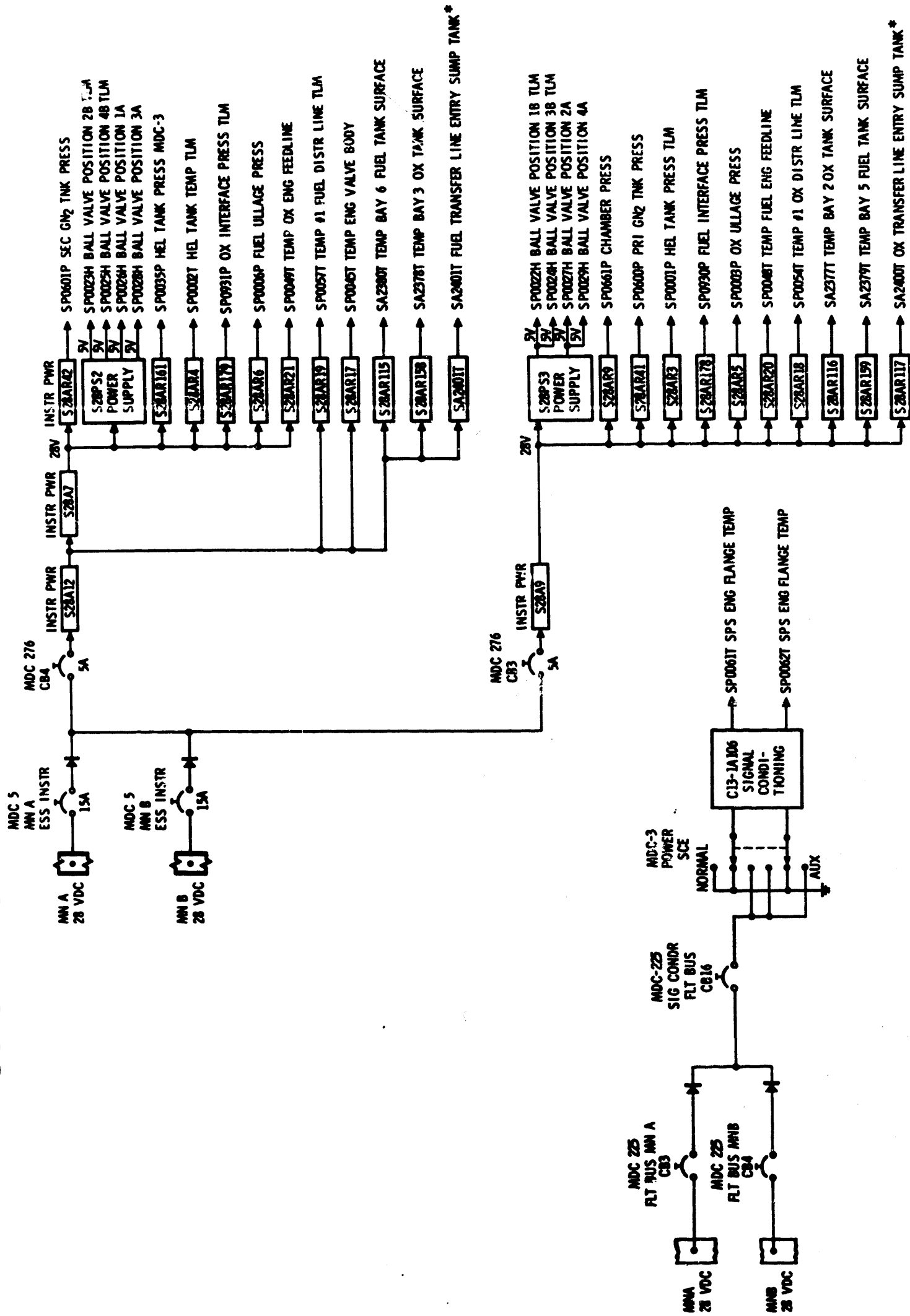


CSM ROTATIONS



SPACE CRAFT AXIS	SPACE CRAFT NOSE MOTION	THUMB WHEEL TRAVEL	TRIM VOLTAGE	GPI INDICATION	MTVC COMMAND ROTATION CONTROL	ENGINE ANGLE COMM AND	ACTUATOR RAM TRAVEL	ENGINE INJECTOR TRAVEL THRUST VECTOR	ENGINE SKIRT TRAVEL
PITCH	(DOWN)	+NUMERAL	+	+	-PITCH (FORWARD)	+	EXTEND	-Z	+Z
	(UP)	-NUMERAL	-	-	+PITCH (AFT)	-	RETRACT	+Z	-Z
YAW	(LEFT)	+NUMERAL	+	+	-Y (CCW)	+	EXTEND	+Y	-Y
	(RIGHT)	-NUMERAL	-	-	+Y (CW)	-	RETRACT	-Y	+Y

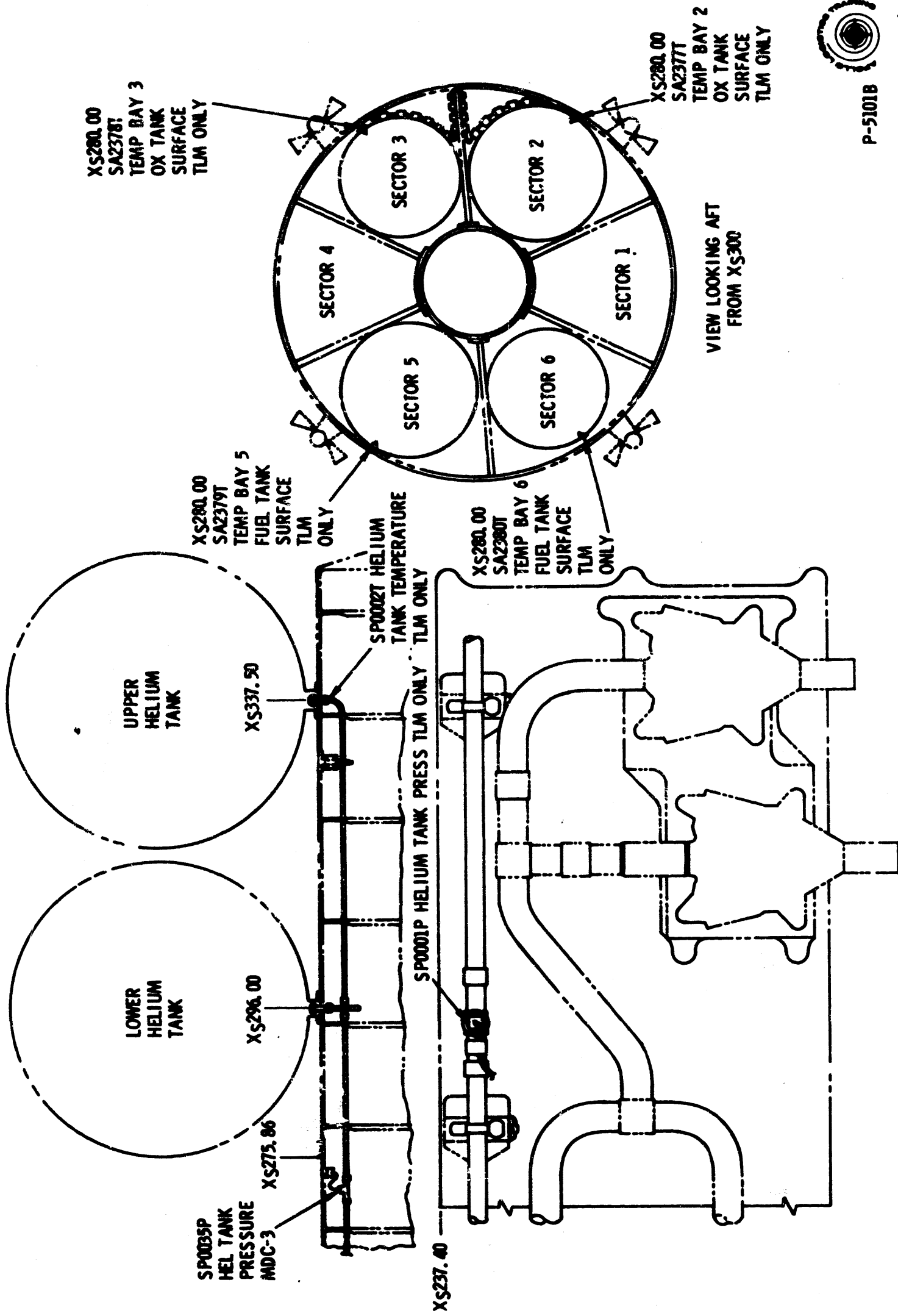
SPS INSTRUMENTATION POWER



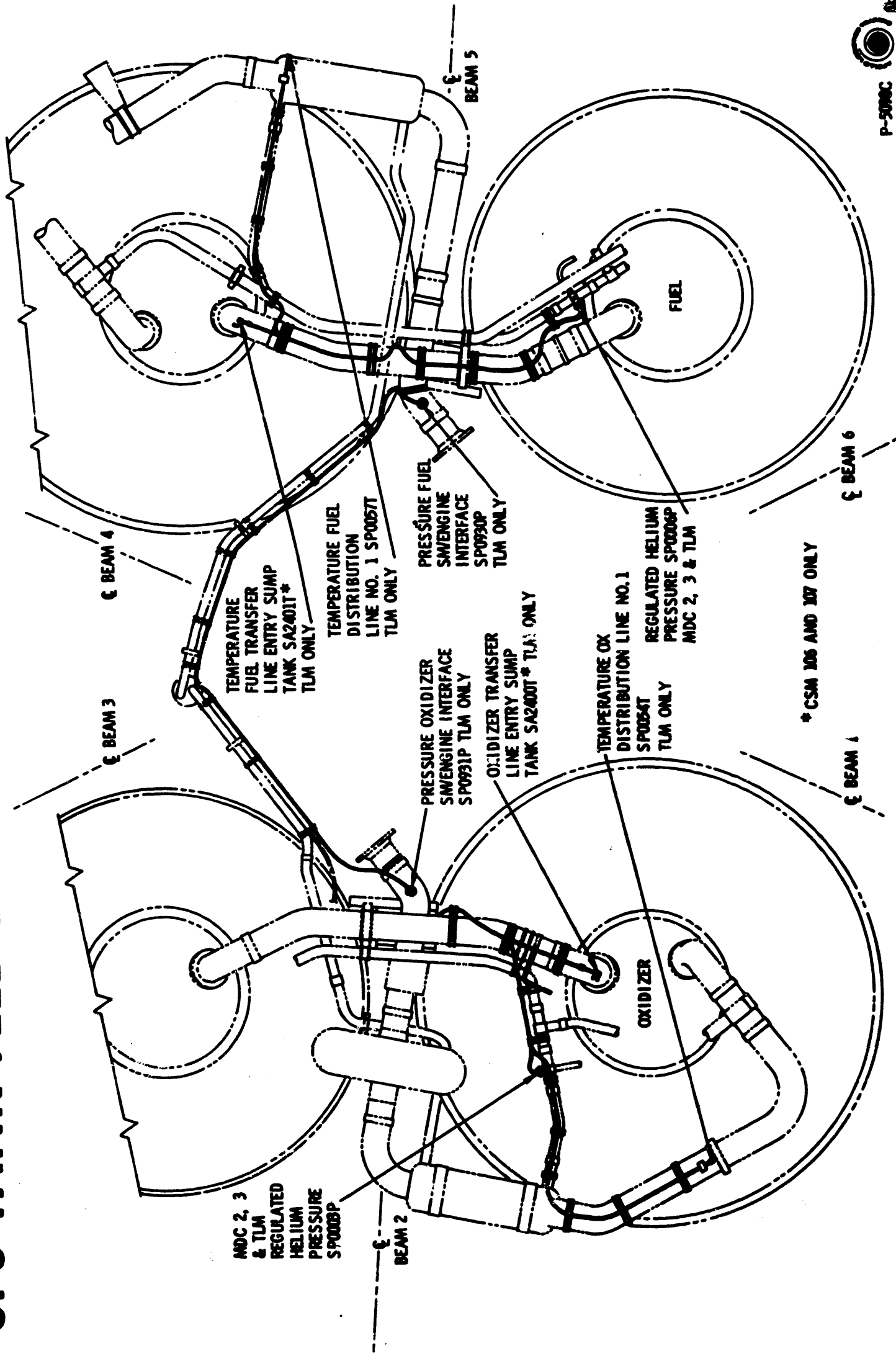
*CSM 106 AND 107 ONLY

SPS INSTRUMENTATION LOCATION

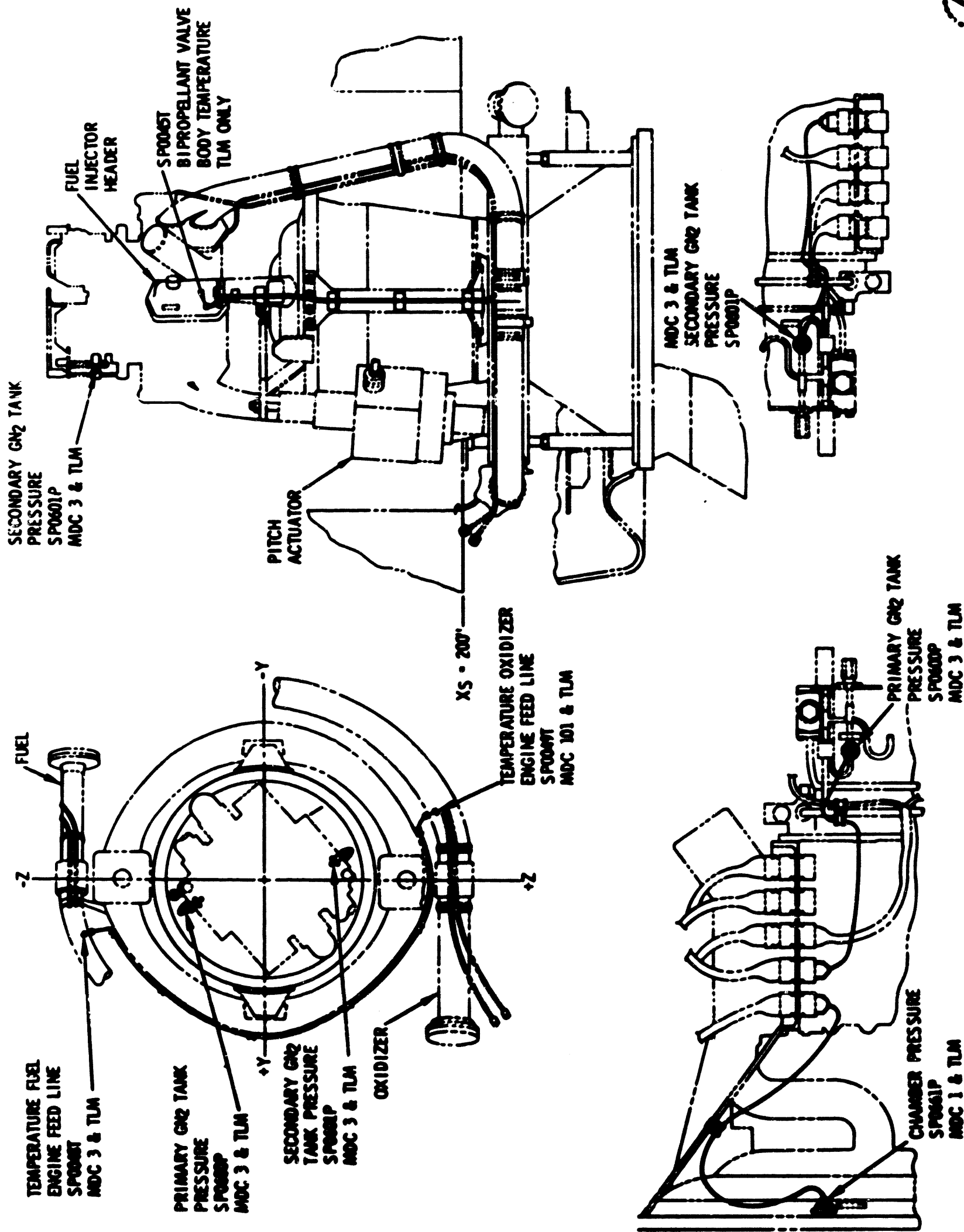
TANK TEMPERATURE, HELIUM TEMPERATURE AND PRESSURE



SPS TANK FEED LINE INSTRUMENTATION LOCATIONS



SPS ENGINE INSTRUMENTATION LOCATIONS



SPS MEASUREMENTS

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
MDC 3	HELIUM TANK PRESSURE	SP0035P	0+5K PSIA	S28AR161			3600±50 PSIA & DECREASE
	HELIUM TANK PRESSURE	SP0001P	0+5K PSIA	S28AR3	11A10	H2	3600±50 PSIA & DECREASE
	HELIUM TANK TEMPERATURE	SP0002T	-100+200°F	S28AR4	10A16	H1	VARIABLE
MDC 2 & 3	FUEL ULLAGE PRESSURE	SP0006P	0+250 PSIA	S28AR6	11A12	H2	170 to 195 PSIA
MDC 2 & 3	OXIDIZER ULLAGE PRESSURE	SP0003P	0+250 PSIA	S28AR5	11A11	H2	
MDC 2 & 3	QUANTITY SENSING SYSTEM SENSOR SELECT IN NORMAL						
	PRIMARY FUEL SUMP #2	SP0658Q	0+60%	C13-1A101	10A31	H1	VARIABLE
	PRIMARY FUEL STORAGE #1	SP0657Q	0+50%	C13-1A101	10A29	H1	
	PRIMARY OXIDIZER SUMP #2	SP0656Q	0+60%	C13-1A101	10A28	H1	
	PRIMARY OXIDIZER STOR. #1	SP0655Q	0+50%	C13-1A101	10A26	H1	

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
SENSOR SELECT IN PRIMARY							
	PRIMARY FUEL SUMP #2	SP0658Q	0+60%	C13-1A101	10A31	H1	PCM
	PRIMARY FUEL STOR. #1	SP0657Q	0+50%	C13-1A101	10A29	H1	PCM
	PRIMARY OX. SUMP #2	SP0656Q	0+60%	C13-1A101	10A28	H1	PCM
	PRIMARY OX. STOR. #1	SP0655Q	0+50%	C13-1A101	10A26	H1	PCM
SENSOR SELECT IN AUXILIARY							
	PRIMARY FUEL SUMP #2	SP0658Q	0+60%	C13-1A101	10A31	H1	PCM
	PRIMARY OX. SUMP #2	SP0656Q	0+60%	C13-1A101	10A28	H1	PCM
	AUX FUEL	SP0657Q	0+100%	C13-1A101	10A29	H1	PCM
	AUX OXIDIZER	SP0655Q	0+100%	C13-1A101	10A26	H1	PCM
	TEMP #1 OX. DISTR. LINE	SP0054T	0+200 F	S28AR18	10A44	H1	PCM
	TEMP #1 FUEL DIST. LINE	SP0057T	0+200°F	S28AR19	10A59	H1	PCM
	FUEL INTERFACE PRESSURE TO ENGINE	SP0930P	0+300 PSIA	S28AR178	11A152	H1	PCM
	OXIDIZER INTERFACE PRESSURE TO ENGINE	SP0931P	0+300 PSIA	S28AR179	11A161	H1	PCM

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
MDC 3	TEMP FUEL ENGINE FEEDLINE	SP0048T	0+200° F	S28AR20	10A117	H2 PCM	+55 to +75° F during translunar coast +70 to +90° F during lunar orbit and transearth coast. Redline markings at +27 and +100° F.
MDC 101	TEMP OX ENGINE FEEDLINE	SP0049T	0+200° F	S28AR21	10A120	H2 PCM	1.4 VDC = 55° F 1.812 VDC = 70° F 1.95 VDC = 75° F 2.362 VDC = 90° F
	TEMP ENGINE VALVE BODY	SP0045T	0+200° F	S28AR17	10A37	HI PCM	+27° F min. +100° F max.
MDC 3	PRIMARY GN ₂ TANK PRESS	SP0600P	0+5K PSIA	S28AR41	10A10	HI PCM	2500+ 50 PSIA AND DECREASE
MDC 3	SECONDARY GN ₂ TANK PRESS	SP0601P	0+5K PSIA	S28AR42	10A11	HI PCM	
	BALL VALVE POSITION 1B	SP0022H	0+90 DEG	CI3-1A101	11A31	HI PCM	OPEN OR CLOSED (0° or 90°)
MDC 3	BALL VALVE POSITION 1A	SP0026H	0+90 DEG				
	BALL VALVE POSITION 2B	SP0023H	0+90 DEG	CI3-1A101	11A32	HI PCM	
MDC 3	BALL VALVE POSITION 2A	SP0027H	0+90 DEG				
	BALL VALVE POSITION 3B	SP0024H	0+90 DEG	CI3-1A101	11A33	HI PCM	

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
MDC 3	BALL VALVE POSITION 3A	SP0028H	0+90 DEG				OPEN OR CLOSED (0° or 90°)
	BALL VALVE POSITION 4B	SP0025H	0+90 DEG	CI3-1A10I	11A34	HI	
	BALL VALVE POSITION 4A	SP0029H	0+90 DEG				
MDC 1	CHAMBER PRESSURE	SP0661P	0+150 PSIA	S28AR9	12A14	HI	95 to 105 PSIA during burn.
MDC 2	ENGINE INJECTOR FLANGE TEMP. NO. 1	SP0061T	0+600° F	CI3-1A106	10A47	HI	ILLUMI- NATES SPS FLANGE TEMP HI ON MDC 2 AT 480° F = 4 VDC
	ENGINE INJECTOR FLANGE TEMP. NO. 2	SP0062T	0+600° F	CI3-1A106	10A49	HI	
	GIMBAL POSITION PITCH 1 OR 2	CH3517H	-5.0 to +5.0 DEG	SCS	12A8	HI	VARIABLE
	GIMBAL POSITION YAW 1 OR 2	CH3518H	-5.0 to +5.0 DEG	SCS	12A10	HI	
	YAW DIFFERENTIAL CURRENT	CH3667C	-0.835 to +0.835 AMP	SCS	12A16	HI	
	PITCH DIFFERENTIAL CURRENT	CH3666C	-0.835 to +0.835 AMP	SCS	22A4	HI	
	SPS SOLENOID DRIVER #1	CH3640X	EVENT	SCS	11E4-4	E2	OFF ON EVENT
	SPS SOLENOID DRIVER #2	CH3605X	EVENT	SCS	11E9-4	E2	

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
	TEMP BAY 2 OX TANK SURFACE	SA2377T	-100+200°F	S28ARI16	I0A142	HI	PCM
	TEMP BAY 3 OX TANK SURFACE	SA2378T	-100+200°F	S28ARI58	I0A137	HI	PCM
	TEMP BAY 5 FUEL TANK SURFACE	SA2379T	-100+200°F	S28ARI59	I0A139	HI	PCM
	TANK BAY 6 FUEL TANK SURFACE	SA2380T	-100+200°F	S28ARI15	I0A140	HI	PCM
	*OX TRANSFER LINE ENTRY SUMP TANK	SA2400T	-100+200°F	S28ARI17	I0A143	HI	PCM
	*FUEL TRANSFER ENTRY SUMP TANK	SA2401T	-100+200°F	S28ARI18	I0A145	HI	PCM

*CSM 106 and 107 only

SMRCS DATA

HELIUM TANKS (4)

4150±50 PSIG at 70±5° during servicing, after servicing setting on launch pad 70±10°F.

Capacity 1.35 lbs.

Internal volume of 910±5 cubic inches

Weight 11.5 lbs. wall thickness, 0.135 inch.

Diameter 12.37"

REGULATOR UNITS (8)

Filter 25 micron nominal, 40 micron absolute

Primary 181±3 PSIG with a normal lock-up of 183±5 PSIG.

Secondary lock-up of 187±5 PSIG. From lock-up pressure do not drop below 182 PSIG or rise above 188 PSIG.

SECONDARY FUEL PRESSURE TRANSDUCERS (4)

Illuminate Caution and Warning light on MDC-2 (SMRCS A, B, C, D).

Under pressure 145 PSIA nominal

Over pressure 215 PSIA nominal

CHECK VALVES - FILTERS

40 micron nominal, 74 micron absolute, one at inlet to check valve assembly, one at each test port.

HELIUM RELIEF VALVES (8)

Diaphragm rupture at 228±8 PSIG

Filter, 10 micron nominal, 25 micron absolute

Relief valve relieves at 236.5±1.5 PSIG.

Relief valve reseats at not less than 220 PSIG.

Flow capacity .3 lbs/minute at 248 PSIG at 60°F.

Bleed device closes when increasing pressure reaches no more than 179 PSIG

in the cavity and a helium flow of less than 20 standard cubic centimeters per hour across the bleed device and relief valve assembly combined.

The bleed device shall reopen when decreasing pressure has reached no less than 20 PSIG.

PRIMARY FUEL TANK (4)

Combined propellant and ullage volume of 69.1 lbs, initially at 150 PSIG, resulting in a tank pressure of no more than 215 PSIA when heated to 85°F.

Outside diameter 12.62" maximum

Length 23.717" + .060" - .000"

Wall thickness .017" to .022"

Helium inlet port 1/4", fill and drain port 1/2"

SMRCS DATA (Cont)

PRIMARY OXIDIZER TANK (4)

Combined propellant and ullage volume of 137.0 lbs., initially at 65°F at 150 PSIG, resulting in a tank pressure of no more than 215 PSIA when heated to 85°F.

Outside diameter 12.62" maximum

Length 28.558" + .060" - .000"

Wall thickness .017" to .022"

SECONDARY FUEL TANK (4)

Combined propellant and ullage volume of 45.2 lbs., initially at 65°F at 150 PSIG, resulting in a tank pressure of no more than 205 PSIA when heated to 105°F.

Outside diameter 12.62" maximum

Length 17.329" + .040" - .000"

Wall thickness .022" to .027"

SECONDARY OXIDIZER TANK (4)

Combined propellant and ullage volume of 89.2 lbs., initially at 65°F at 150 PSIG, resulting in a tank pressure of no more than 205 PSIA when heated to 105°F.

Outside diameter 12.65" maximum

Length 19.907" + .040" - .000"

Wall thickness .022" to .027"

IN LINE FILTERS (8)

5 micron nominal, 15 micron absolute

ENGINE (16)

1000 second service life, 750 seconds continuous CSM 103 and subs, capable of 10,000 operational cycles.

Expansion ratio 40 to 1 at nozzle exit

Cooling - Film and radiation

Injector - Pre-mix ignitor one on one unlike impingement, 8 fuel annulus for film cooling of pre-mix ignitor, main chamber 8 on 8 unlike impingement, 8 fuel for film cooling of combustion chamber wall

Nozzle extension - L 605 cobalt base

Nozzle exit diameter - 5.6"

Fuel lead

Automatic coils - Connected in parallel

Direct coils - Connected in series

Weight - 4.99 lbs.

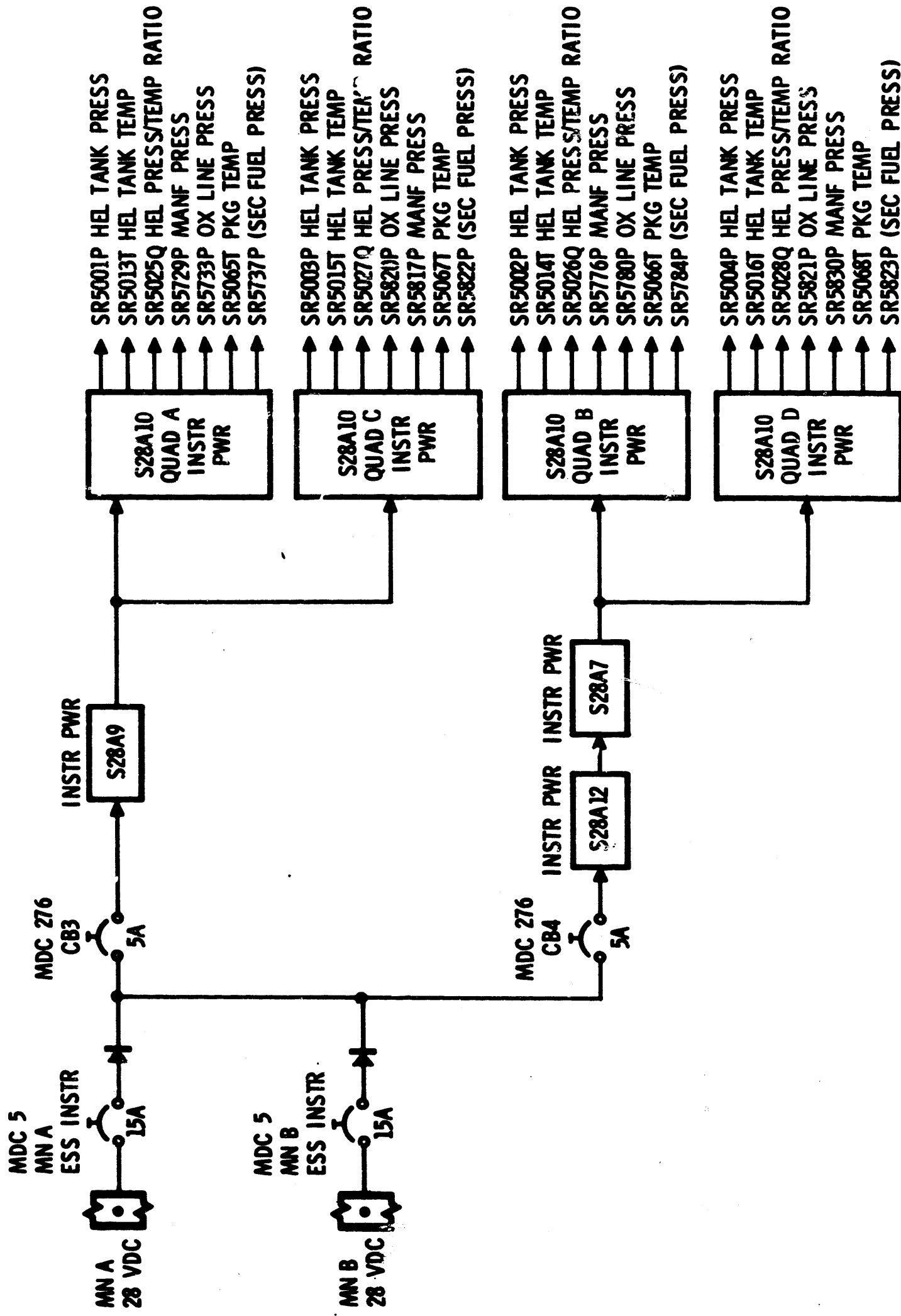
Length - 13.400" maximum

Filters - Each injector valve inlet, 100 micron nominal, 250 micron absolute

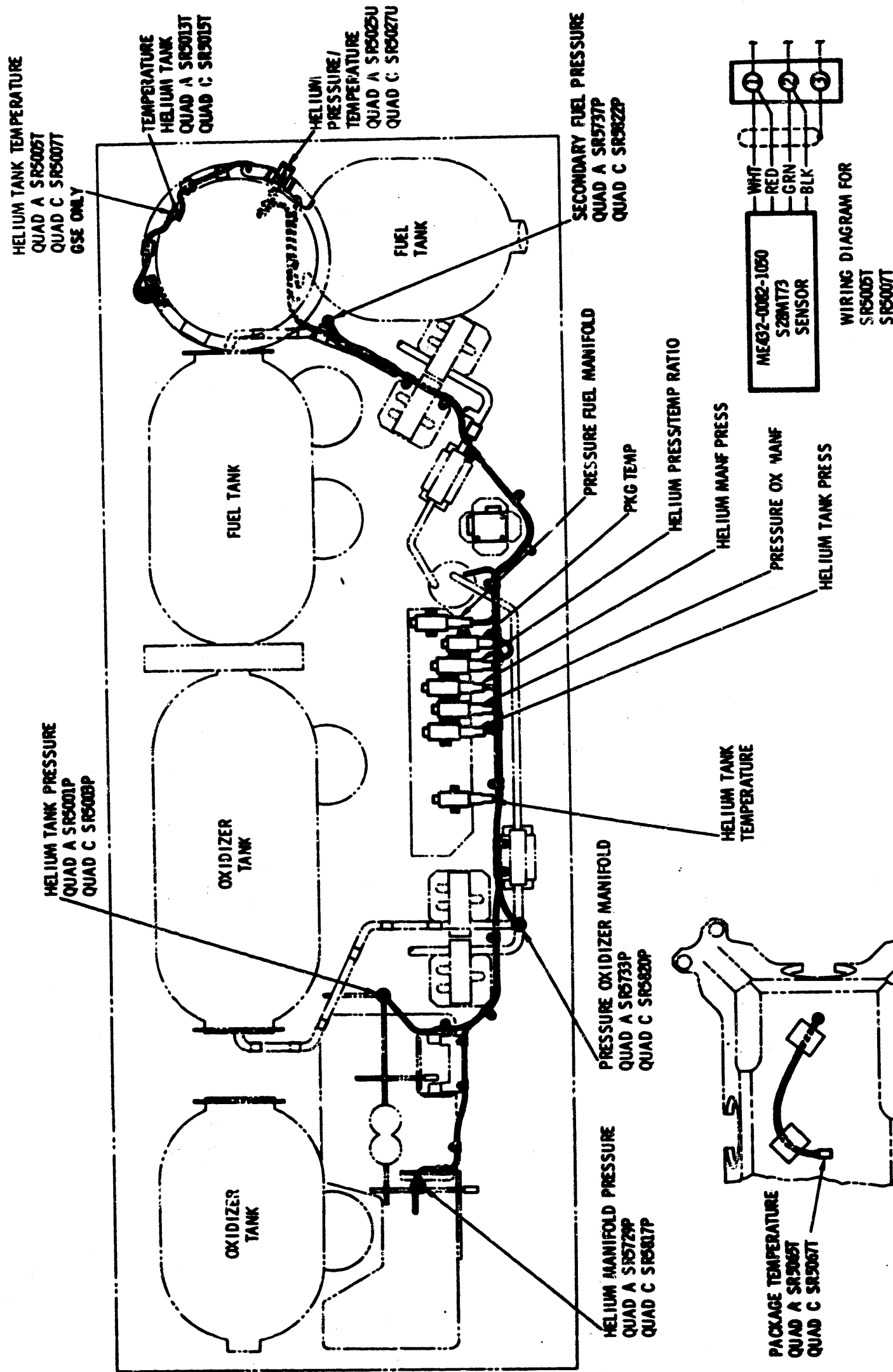
SMRCS DATA (Cont)

PACKAGE TEMPERATURE TRANSDUCER (4)	Illuminate Caution and Warning light on MDC-2 (SMRCS A, B, C, D) 75°F nominal
HEATER THERMO SWITCH CSM 106 and Subs	205° nominal Open at 134°F Close at 115°F Minimum Spread 9°F 36±3.6 watts per element nominal Two per quad

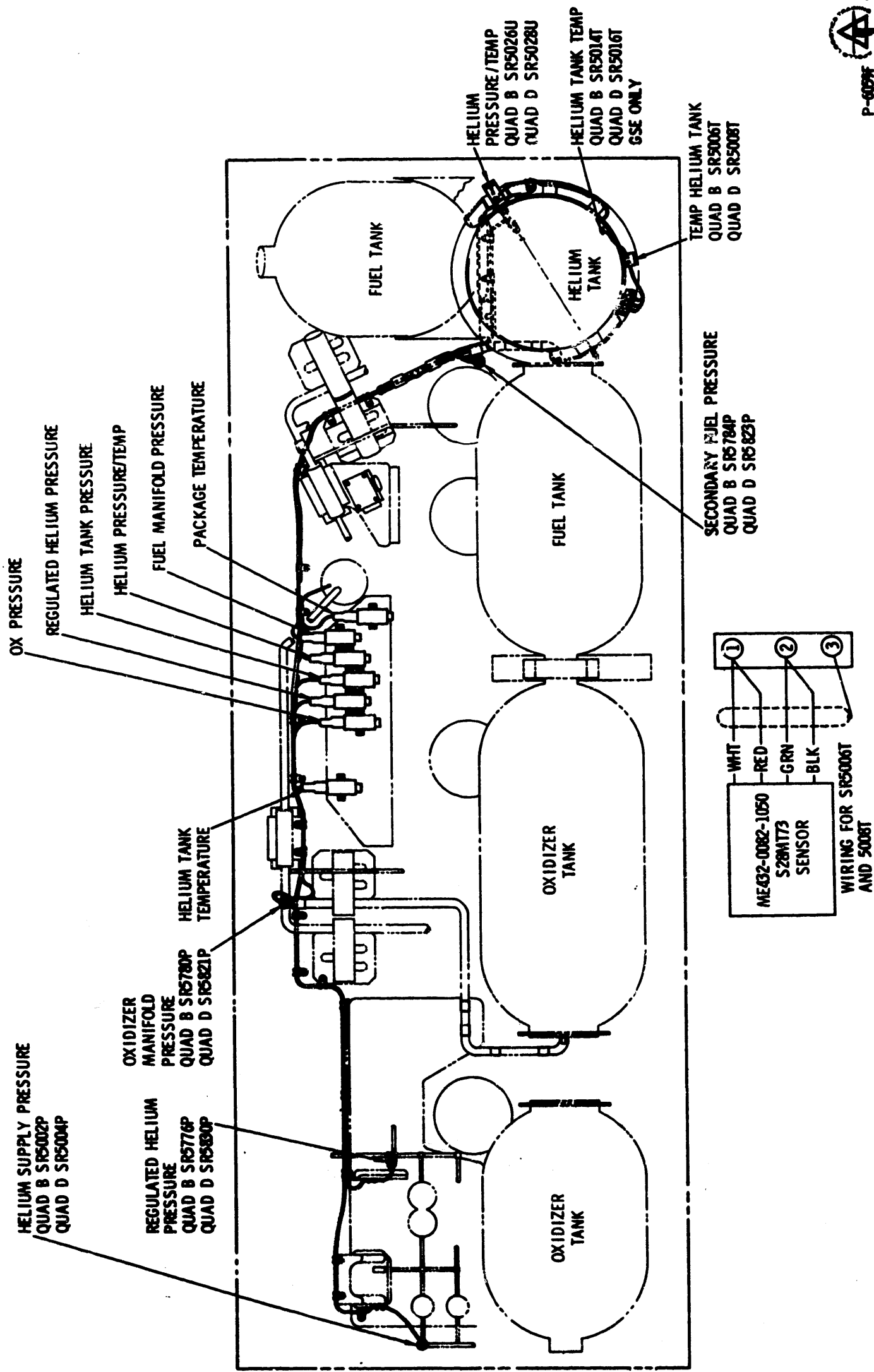
SM RCS INSTRUMENTATION POWER



SM RCS QUAD A&C INSTRUMENTATION



SM RCS QUAD B&D INSTRUMENTATION



SMRCS MEASUREMENTS

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
MDC-2	Helium Tank Pressure A	SR5001P	0+5K PSIA	S28AR36(SI9A1)	10A24	H2	PCM
MDC-2	Helium Tank Pressure B	SR5002P	0+5K PSIA	S28AR36(SI9A2)	10A27	H2	PCM
MDC-2	Helium Tank Pressure C	SR5003P	0+5K PSIA	S28AR36(SI9A3)	10A30	H2	4150 ⁺ -50 PSIA and Decrease
MDC-2	Helium Tank Pressure D	SR5004P	0+5K PSIA	S28AR36(SI9A4)	10A33	H2	PCM
MDC-2	Helium Tank Temperature A	SR5013T	0+100°F	S28AR164(SI9A1)	11A163	H2	PCM
MDC-2	Helium Tank Temperature B	SR5014T	0+100°F	S28AR164(SI9A2)	11A164	H2	VARIABLE
MDC-2	Helium Tank Temperature C	SR5015T	0+100°F	S28AR164(SI9A3)	11A165	H2	PCM
MDC-2	Helium Tank Temperature D	SR5016T	0+100°F	S28AR164(SI9A4)	11A166	H2	PCM
MDC-2	Helium Press/Temp Ratio System A	SR5025Q	0+100%	S28AR174(SI9A1)	10A93	H2	PCM
MDC-2	Helium Press/Temp Ratio System B	SR5026Q	0+100%	S28AR174(SI9A2)	10A99	H2	VARIABLE
MDC-2	Helium Press/Temp Ratio System C	SR5027Q	0+100%	S28AR174(SI9A3)	10A102	H2	PCM
MDC-2	Helium Press/Temp Ratio System D	SR5028Q	0+100%	S28AR174(SI9A4)	10A105	H2	PCM
	Manifold Pressure A	SR5729P	0+400 PSIA	S28AR44(SI9A1)	11A39	H2	PCM 178
	Manifold Pressure B	SR5776P	0+400 PSIA	S28AR44(SI9A2)	11A40	H2	PCM to
	Manifold Pressure C	SR5817P	0+400 PSIA	S28AR44(SI9A3)	11A46	H2	PCM 192
	Manifold Pressure D	SR5830P	0+400 PSIA	S28AR44(SI9A4)	11A130	H2	PCM PSIA

SMRCS MEASUREMENTS

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
MDC-2	Oxidizer Line Pressure A	SR5733P	0+300 PSIA	S28ARI70(SI9A1)	11A23	H1	PCM
	Secondary Fuel Pressure A	SR5737P	0+400 PSIA	S28ARI71(SI9A1)	11A92	H2	PCM
	Oxidizer Line Pressure B	SR5780P	0+300 PSIA	S28ARI70(SI9A2)	11A24	H1	PCM 178
	Secondary Fuel Pressure B	SR5784P	0+400 PSIA	S28ARI71(SI9A2)	11A94	H2	PCM to
MDC-2	Oxidizer Line Pressure C	SR5820P	0+300 PSIA	S28ARI70(SI9A3)	11A27	H1	PCM 192
	Secondary Fuel Pressure C	SR5822P	0+400 PSIA	S28ARI71(SI9A3)	11A111	H2	PCM PSIA
	Oxidizer Line Pressure D	SR5821P	0+300 PSIA	S28ARI70(SI9A4)	11A28	H1	PCM
	Secondary Fuel Pressure D	SR5823P	0+400 PSIA	S28ARI71(SI9A4)	11A112	H2	PCM
MDC-2	A Package Temperature	SR5065T	0+300° F	S28AR40(SI9A1)	10A136	H1	PCM 75° F NOMINAL
MDC-2	B Package Temperature	SR5066T	0+300° F	S28AR40(SI9A2)	10A17	H1	PCM to
MDC-2	C Package Temperature	SR5067T	0+300° F	S28AR40(SI9A3)	10A19	H1	PCM 205° F NOMINAL
MDC-2	D Package Temperature	SR5068T	0+300° F	S28AR40(SI9A4)	10A20	H1	PCM
	Direct RCS Switch No. 1 Enable	CH3601X	EVENT	SCS	11E24-3	E2	PCM OFF ON EVENT
	Direct RCS Switch No. 2 Enable	CH3602X	EVENT	SCS	11E24-4	E2	PCM

CMRCS DATA

HELIUM TANKS (2)

4150±50 PSIG at 70±5° during servicing, setting on launch pad 70±10°F
Capacity .57 lbs.
Inside diameter 8.84"
Wall thickness .105".
Internal volume of 365±5 cubic inches.
Weight 5.25 lbs.

HELIUM ISOLATION SQUIB VALVE FILTER

Remove 98% of all particles whose two smallest dimensions are greater than 40 microns. Remove 100% of all particles whose two smallest dimensions are greater than 74 microns.

REGULATOR UNITS (4)

Filter 25 micron nominal, 40 micron absolute
Primary 291±4 PSIG, lockup pressure minimum of 287 PSIG and not to exceed 302 PSIG.
Secondary lockup 287 to 302 PSIG

CHECK VALVE FILTERS

40 micron nominal, 74 micron absolute, one at inlet to check valve assembly, one each at test port

HELIUM RELIEF VALVES (4)

Diaphragm rupture at 340±8 PSIG
Filter, 10 micron nominal, 25 micron absolute
Relief valve relieves at 346±14 PSIG
Relief valve reseats at no less than 327 PSIG
Flow capacity .3 lb/minute at 60°F and 346±14 PSIG
Bleed device closes when increasing pressure has reached no more than 179 PSIG in the cavity and a helium flow of less than 20 standard cubic centimeters per hour across the bleed device and relief valve assembly combined. Bleed device shall reopen when decreasing pressure has reached no less than 20 PSIG.

HELIUM MANIFOLD PRESSURE TRANSDUCERS (4)

Illuminates Caution and Warning light on MDC-2 (CMRCS 1 or 2)
after helium isolation squib valve actuation.
Under pressure 260 PSIA nominal
Over pressure 330 PSIA nominal

FUEL TANKS (2)

See SMRCS secondary fuel tank

OXIDIZER TANKS (2)

See SMRCS secondary oxidizer tank

CMRCS DATA

VALVE ISOLATION (4) BURST DIAPHRAGM

Rupture at 241 ± 14 PSIG within 2 seconds after rupture pressure is reached at any temperature between 40 to 105°F
Filter, 75 micron nominal, 100 micron absolute

ENGINE (12)

200 second service life, 3000 operational cycles

Expansion ratio 9 to 1

Cooling - Ablation

Injector - 16 on 16 splash plate

Combustion chamber - Refrasil ablative sleeve and graphite base throat insert.

Automatic and direct coils - Connected in parallel

Weight 8.3 lbs.

Length 11.65" maximum

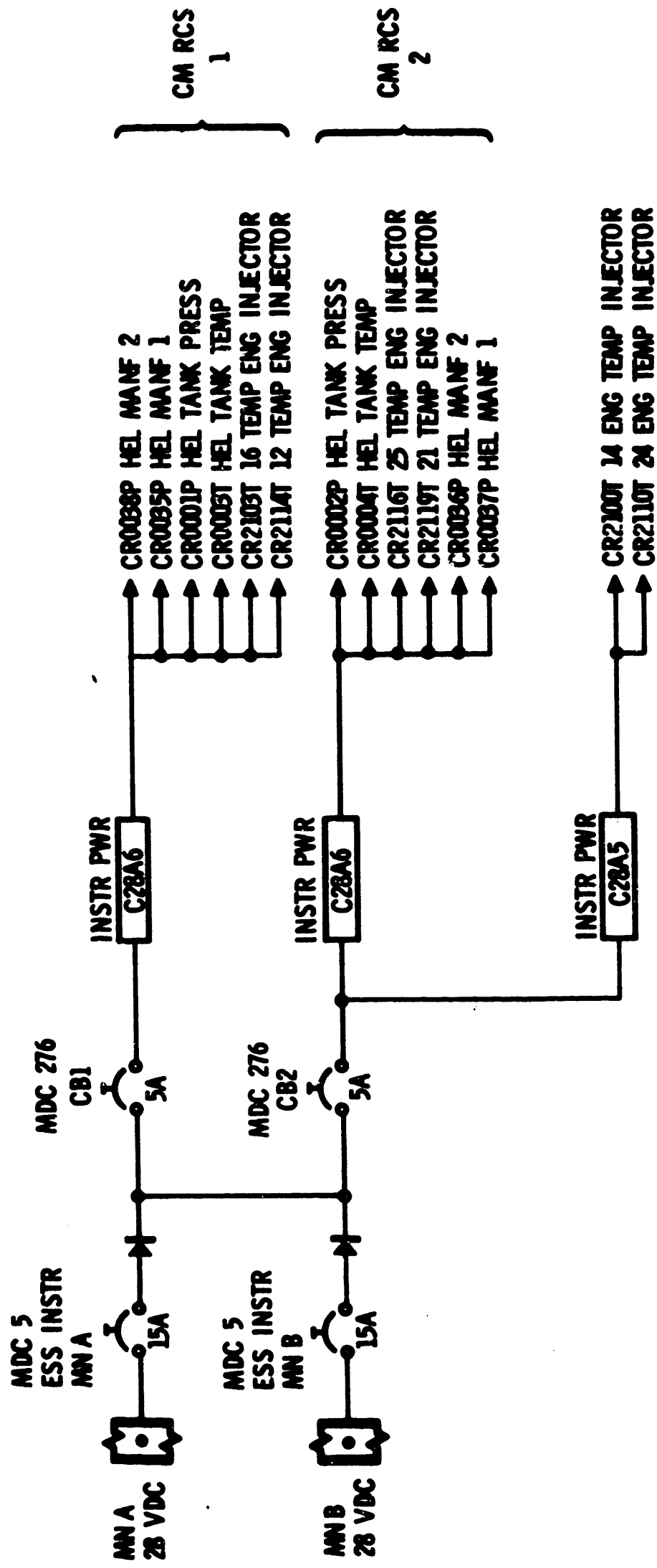
Nozzle exit diameter 2.13"

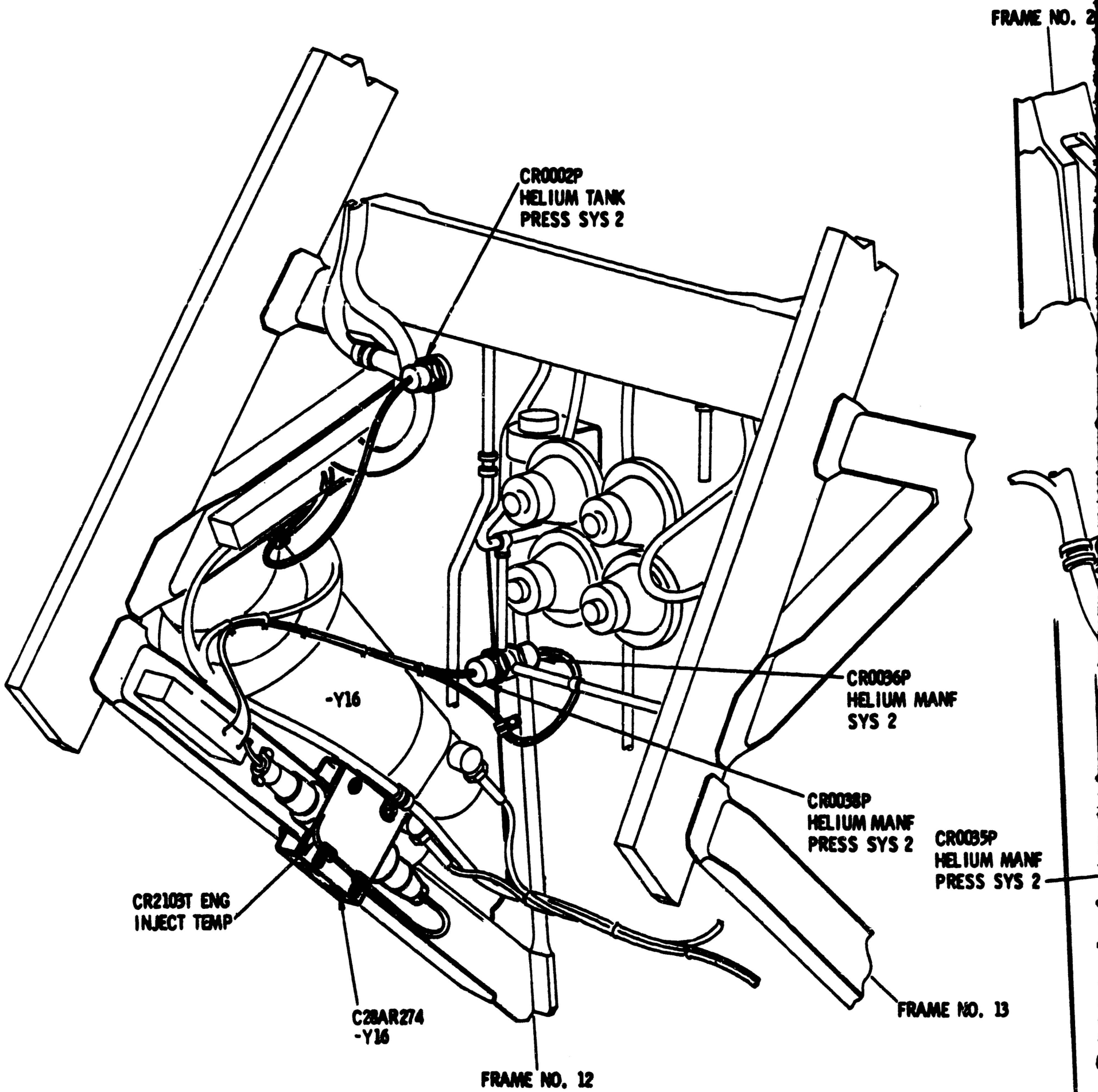
Nozzle extension - Ablative refracil

OXIDIZER AND FUEL BLOWOUT PLUG

Pin shear at approximately 200 PSI.

CM RCS INSTRUMENTATION POWER





FRAME NO. 1

FRAME NO. 24

FRAME

CR0001P
HELIUM TANK
PRESS SYS 1

CR0005P
HELIUM MANF
PRESS SYS 2

CR0007P
HELIUM MANF
SYS 1

+Y25

CR2116T ENG
INJECT TEMP

C28AR272
+Y25

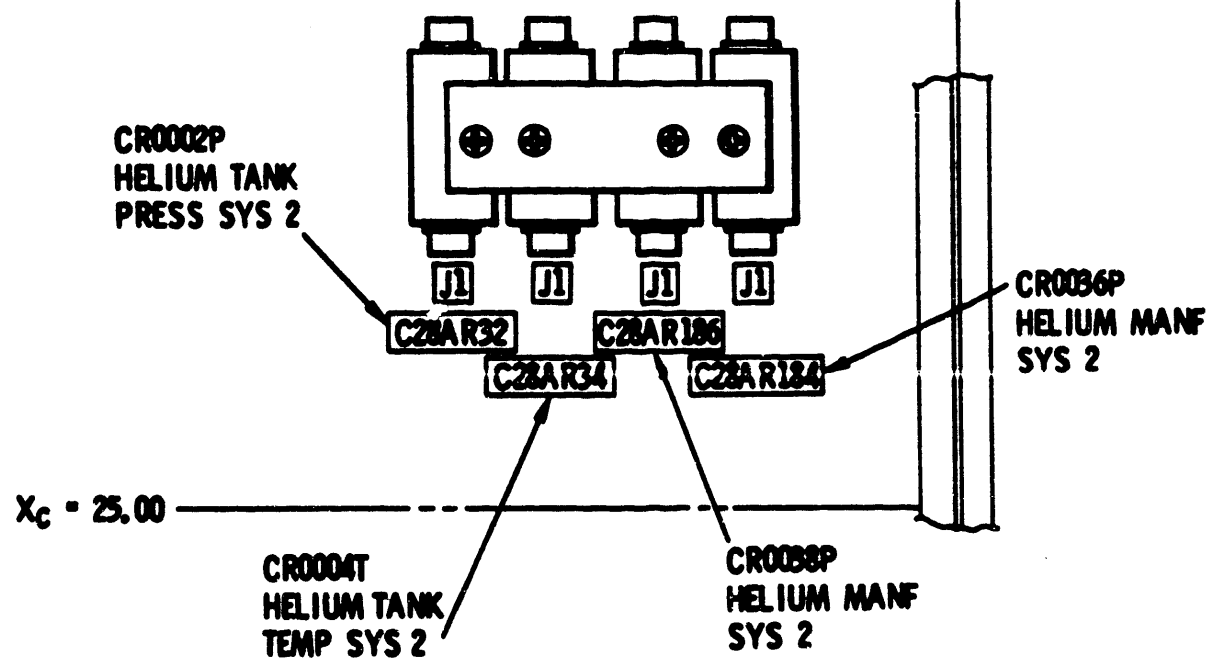
ME NO. 13

Xc = 3

Xc

FRAME NO. 16

FRAME NO. 17

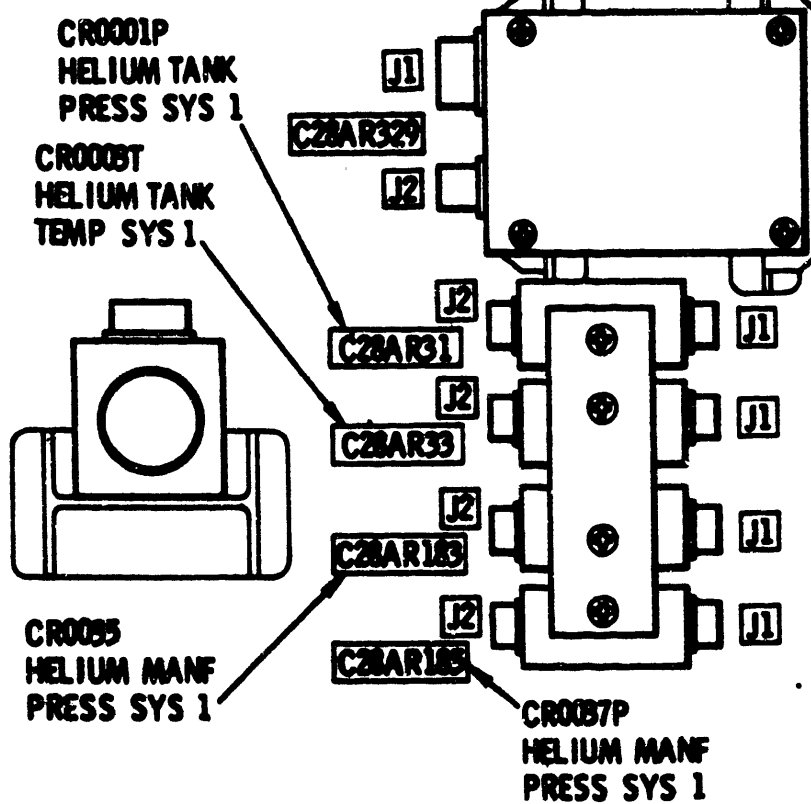


FRAME NO. 16

FRAME NO. 22

FRAME NO. 23

$X_c = 32.00$



$X_c = 22.60$

FRAME NO. 12

NO. 17

CR0036P
HELIUM MANF
SYS 2

FRAME NO. 16

CR2119T ENG
INJECT TEMP

CWR
21

C28AR404

CWR 21

NO. 23

CR0004T
HELIUM TANK
TEMP SYS 2

$X_c = 22.60$

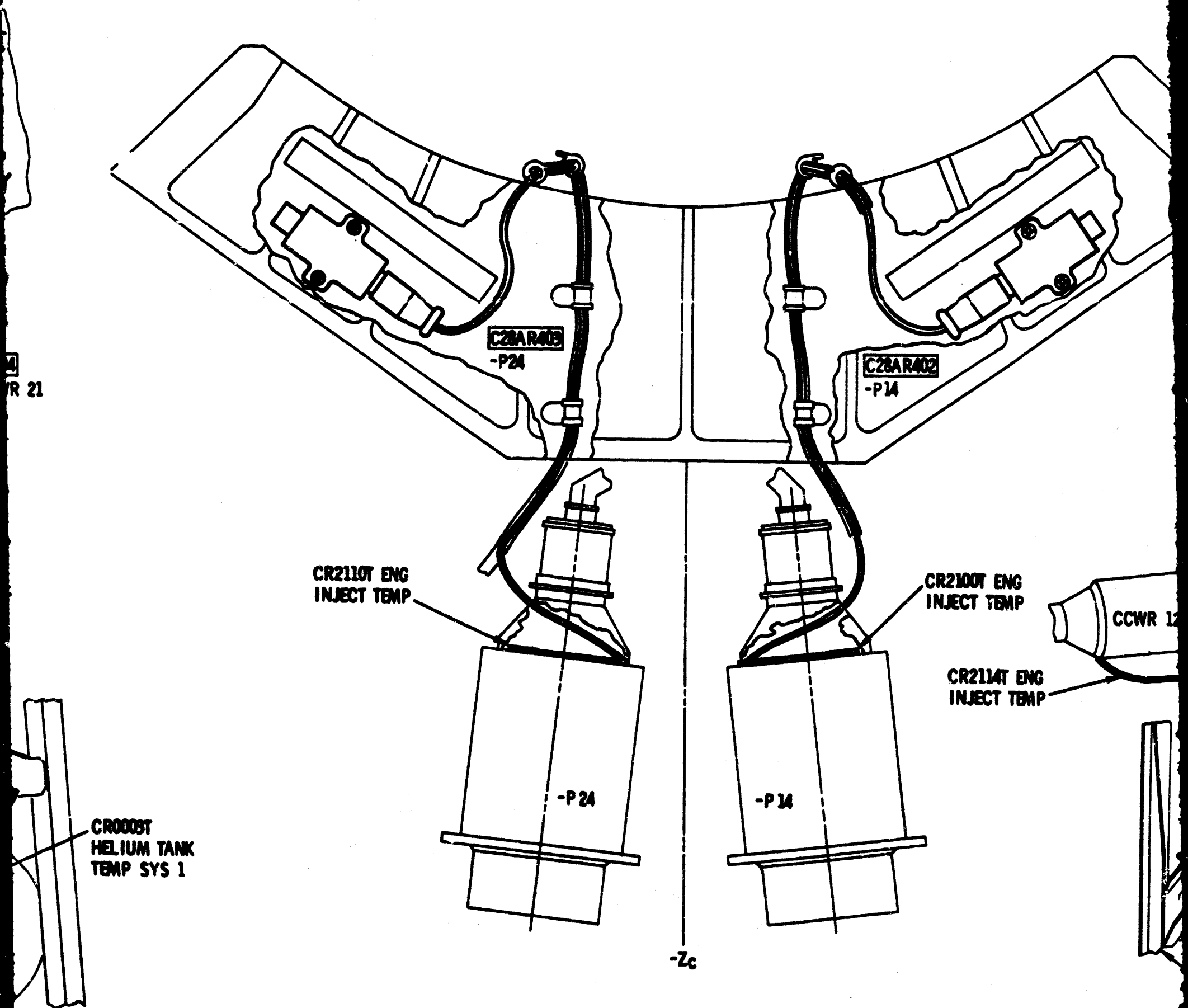
FRAME NO. 12

CR0003T
HELIUM TANK
TEMP SYS 1

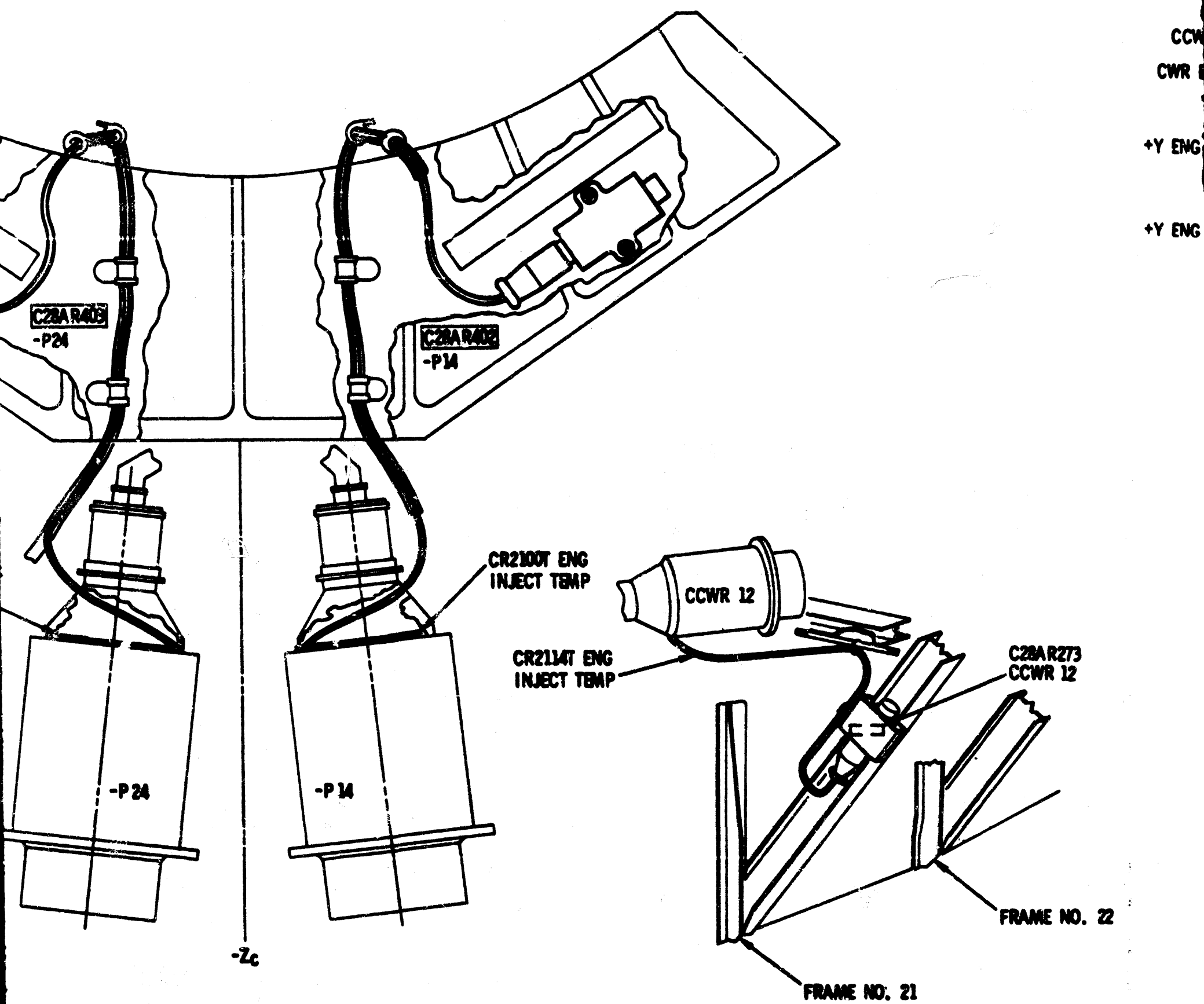
$X_c = 22.60$

FRAME NO. 3

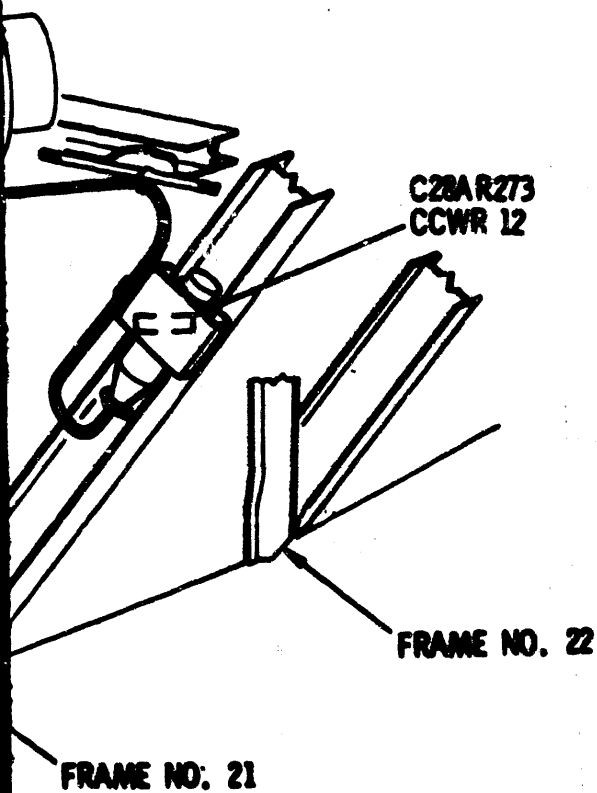
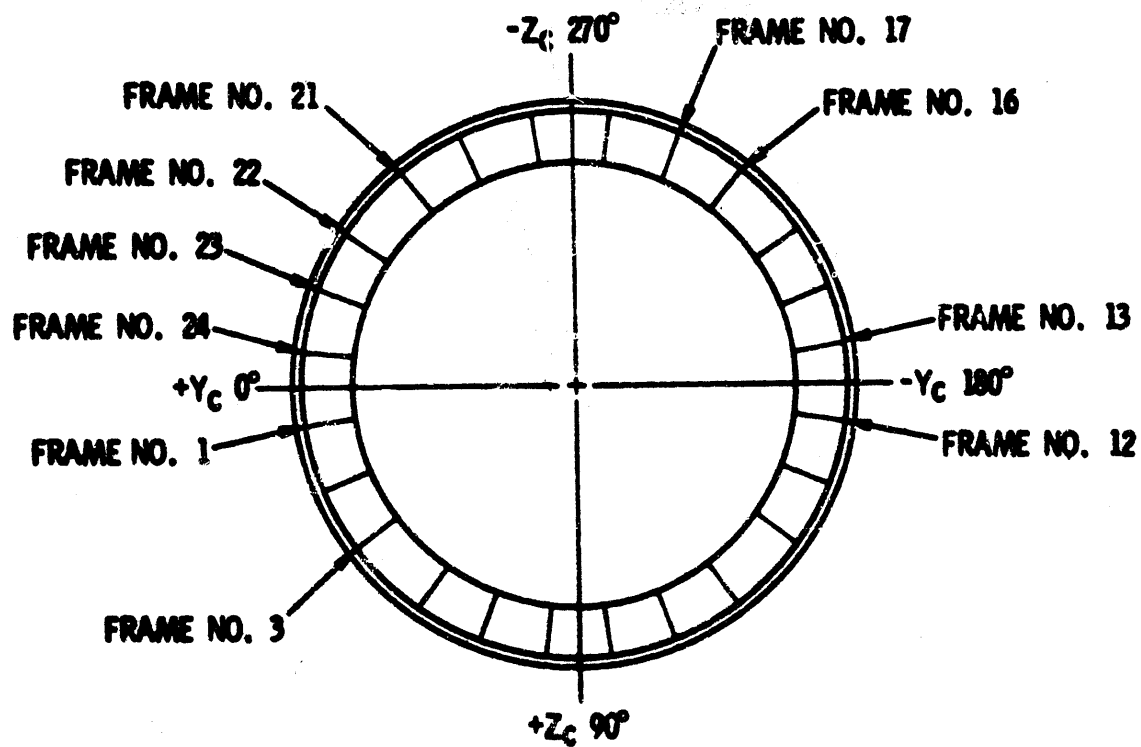
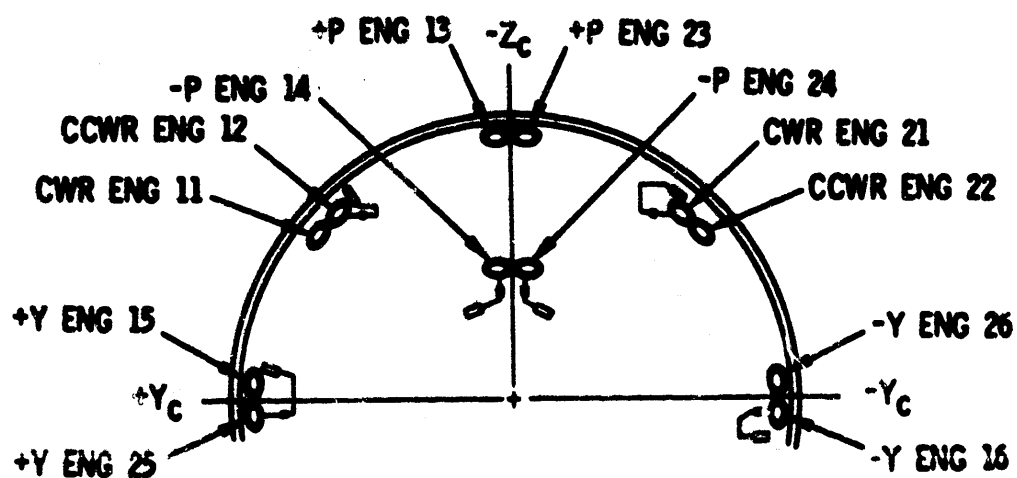
4
R 21



CM RCS INSTRU



CM RCS INSTRUMENTATION LOCATIONS ₂



CMRCS MEASUREMENTS

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
MDC-2	1 Helium Tank Pressure	CR0001P	0+5K PSIA	C28AR31	10A18	H2	4150 $\pm$ 50 PSIA and DECREASE
MDC-2	2 Helium Tank Pressure	CR0002P	0+5K PSIA	C28AR32	10A21	H2	PCM
MDC-2	1 Helium Tank Temperature	CR0003T	0+300°F	C28AR33	11A155	H2	VARIABLE
MDC-2	2 Helium Tank Temperature	CR0004T	0+300°F	C28AR34	11A156	H2	PCM
MDC-2	1 Helium Manifold Press	CR0037P	0+400 PSIA	C28AR185			
	1 Helium Manifold Press	CR0035P	0+400 PSIA	C28AR183	11A21	H2	285 to
MDC-2	2 Helium Manifold Press	CR0038P	0+400 PSIA	C28AR186			302
	2 Helium Manifold Press	CR0036P	0+400 PSIA	C28AR184	11A22	H2	PSIA
MDC-101	1-P Temperature Engine #14 Injector	CR2100T	-50+50°F	C28AR402			
MDC-101	1-Y Temperature Engine #16 Injector	CR2103T	-50+50°F	C28AR274			
MDC-101	1 CCWR Temperature Engine #12 Injector	CR2114T	-50+50°F	C28AR273			
MDC-101	2-P Temperature Engine #24 Injector	CR2110T	-50+50°F	C28AR403			
MDC-101	2+Y Temperature Engine #25 Injector	CR2116T	-50+50°F	C28AR272			
MDC-101	2CWR Temperature Engine #21 Injector	CR2119T	-50+50°F	C28AR404			

Preheat required if below +28°F 3.9 vdc. If heating required heat for 20 minutes.

CMRCS MEASUREMENTS

Crew Display	Measurement	Measurement Number	Range	Signal Conditioner	Channel Code	Bit Rate	Operating Range
	RCS Activate Signal A Z19	CD0170X	EVENT	Seq.	11E22-3	E2	PCM
	RCS Activate Signal B Z19	CD0171X	EVENT	Seq.	11E23-3	E2	PCM
	CMRCS Pressurize Signal A Z7	CD0173X	EVENT	Seq.	11E22-7	E2	PCM
	CMRCS Pressurize Signal B Z7	CD0174X	EVENT	Seq.	11E23-5	E2	PCM
	DIRECT RCS SWITCH NO. 1 ENABLE	CH3601X	EVENT	SCS	11E24-3	E2	PCM
	DIRECT RCS SW SWITCH NO. 2 ENABLE	CH3602X	EVENT	SCS	11E24-4	E2	PCM

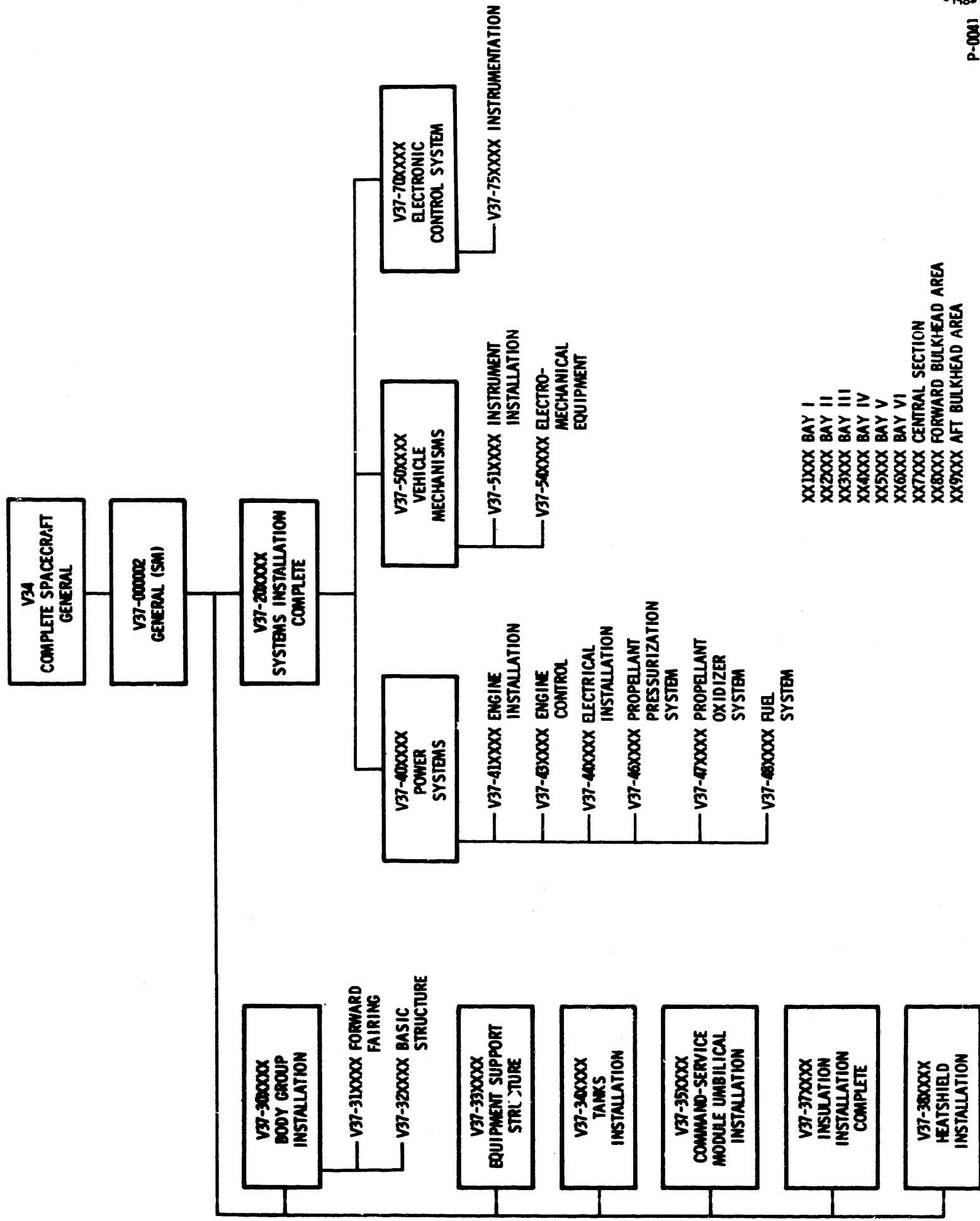
ABBREVIATIONS

AC	ALTERNATING CURRENT	ISO	ISOLATION
ACCEL	ACCELERATION	ISOL	ISOLATION
ACK	ACKNOWLEDGE	JETT	JETTISON
AMP	AMPLIFIER	LBS	POUNDS
AUTO	AUTOMATIC	LDEC	LUNAR DOCKING EVENT CONTROLLER
AUX	AUXILIARY	LES	LAUNCH ESCAPE SYSTEM
BAT	BATTERY	LO	LOW
CAUT/WARN	CAUTION/WARNING	LV	LAUNCH VEHICLE
CCW	COUNTERCLOCKWISE	MAX	MAXIMUM
CL	CENTERLINE	MESC	MASTER EVENT SEQUENCE CONTROLLER
CM OR C/M	COMMAND MODULE	MDC	MAIN DISPLAY CONSOLE
CMC	COMMAND MODULE COMPUTER	MIN	MINIMUM
CMD	COMMAND	MN	MAIN
CONT	CONTROL	MOT	MOTOR
CW	CLOCKWISE	M1	MOTOR 1
DC	DIRECT CURRENT	M2	MOTOR 2
DEC	DECREASE	N	NORMAL
DEM	DEMODULATOR	NEG	NEGATIVE
ECO	ENGINE CUTOFF	NO.	NUMBER
EDS	EMERGENCY DETECTION SYSTEM	NORM	NORMAL
EMS	ENTRY MONITOR SYSTEM	OX	OXIDIZER
EPS	ELECTRICAL POWER SUBSYSTEM	OXID	OXIDIZER
FCSM	FLIGHT COMBUSTION STABILITY MONITOR	POT	POTENTIOMETER
FT	FEET	PRI	PRIMARY
GN2	GASEOUS NITROGEN	PRIM	PRIMARY
GPI	GIMBAL POSITION INDICATOR	PROP	PROPELLANT
GRD	GROUND	PSI	POUNDS PER SQUARE INCH
GSE	GROUND SUPPORT EQUIPMENT	PSIA	POUNDS PER SQUARE INCH ABSOLUTE
He	HELIUM	PSIG	POUNDS PER SQUARE INCH GAUGE
HI	HIGH	PU	PROPELLANT UTILIZATION
HR	HEATER	PUG	PROPELLANT UTILIZATION GAUGING
IMU	INERTIAL MEASUREMENT UNIT	PWR	POWER
INC	INCREASE	RCS	REACTION CONTROL SUBSYSTEM
IND	INDICATOR	ROT	ROTATION
		S	SWITCH

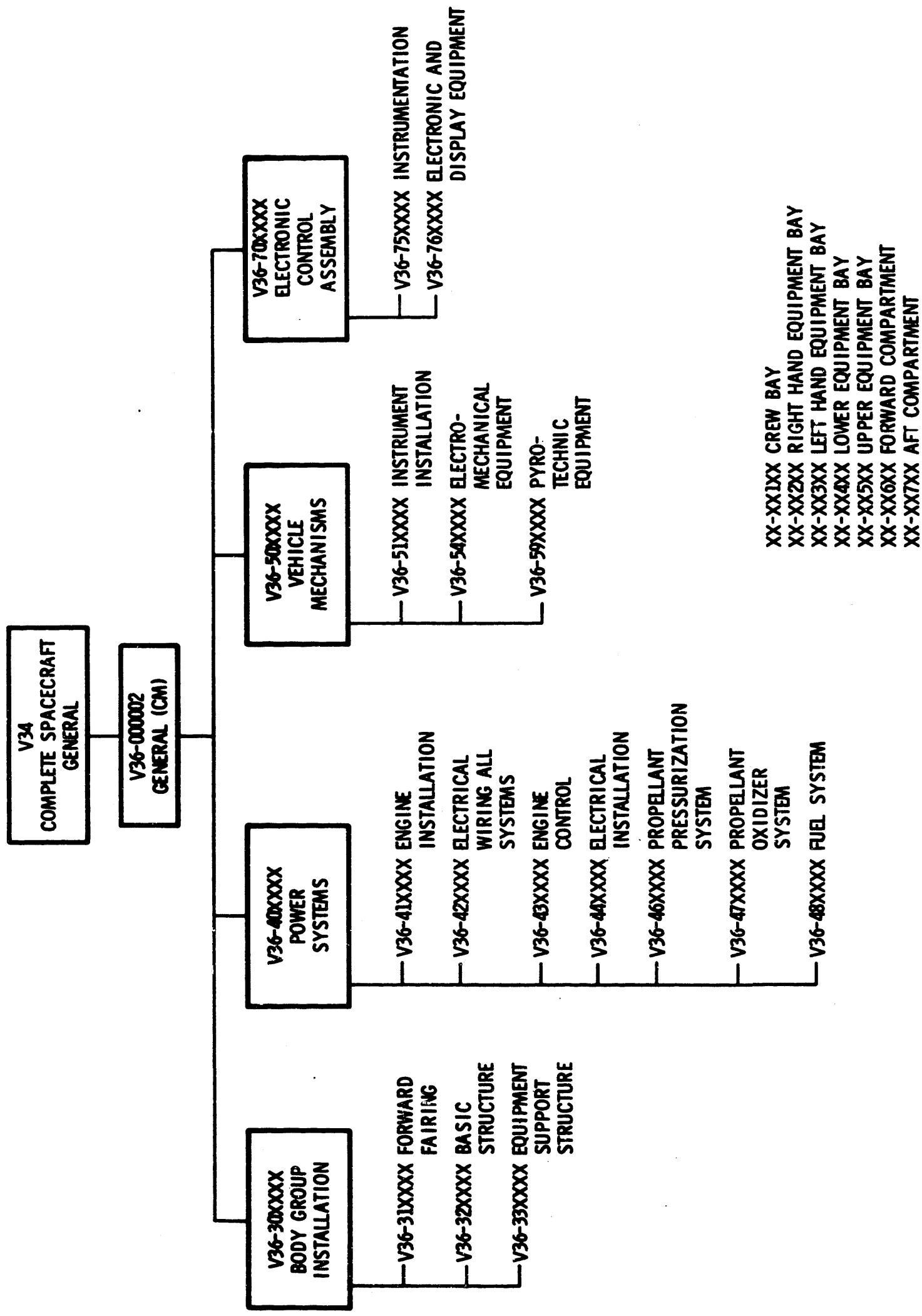
ABBREVIATIONS (Cont'd)

S/C	SPACECRAFT	T.J.	TOWER JETTISON
SCS	STABILIZATION CONTROL SYSTEM	TLM	TELEMETRY
SEC	SECOND	TP	TEST PORT
SEC	SECONDARY	TV	THRUST VECTOR
SEP	SEPARATION	TVC	THRUST VECTOR CONTROL
SEQ	SEQUENCER	UDMH	UNSYMMETRICAL DIAMETHYL HYDRAZINE
SM OR S/M	SERVICE MODULE	V	VOLTS
SPS	SERVICE PROPULSION SUBSYSTEM	X _C	COMMAND MODULE STATION
SW	SWITCH	XPDNR	TRANSPONDER
TEMP	TEMPERATURE	X _S	SERVICE MODULE STATION

APOLLO SERVICE MODULE DRAWING SYSTEM (V37-) SERVICE MODULE PROPULSION SYSTEMS

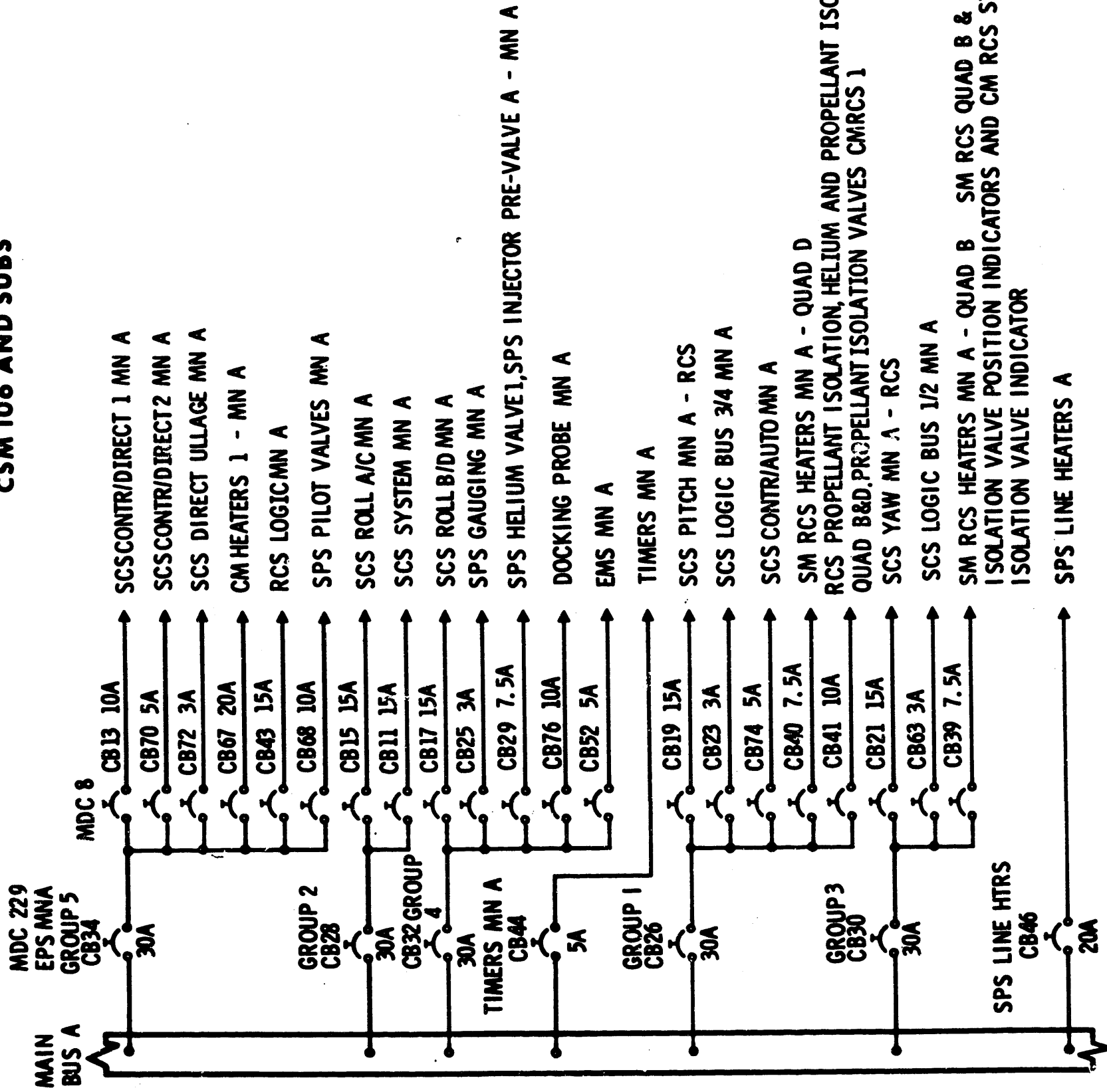


APOLLO COMMAND MODULE DRAWING SYSTEM(V36-) COMMAND MODULE PROPULSION SYSTEMS



MAIN A DC POWER

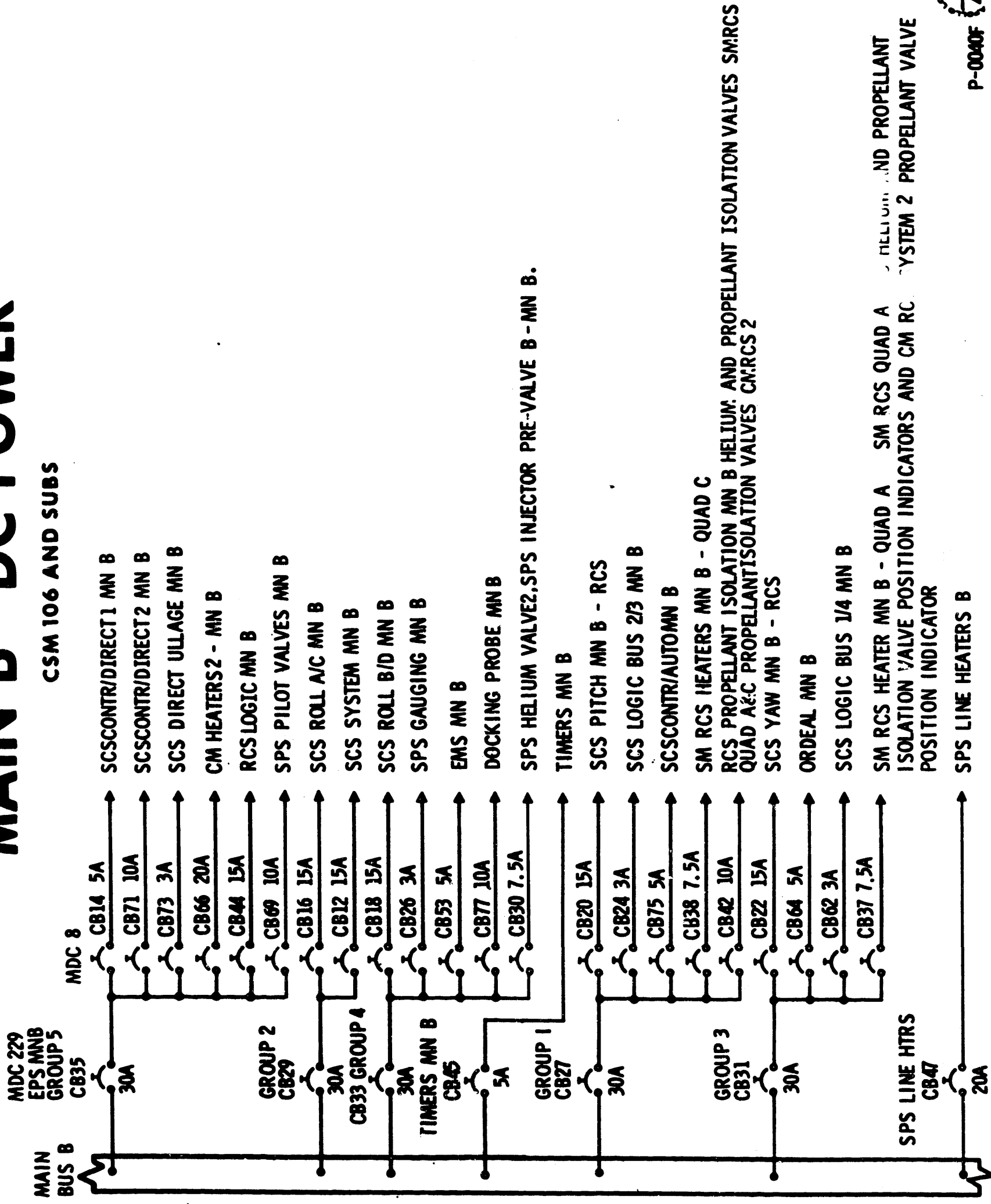
CSM 106 AND SUBS



P-0039F

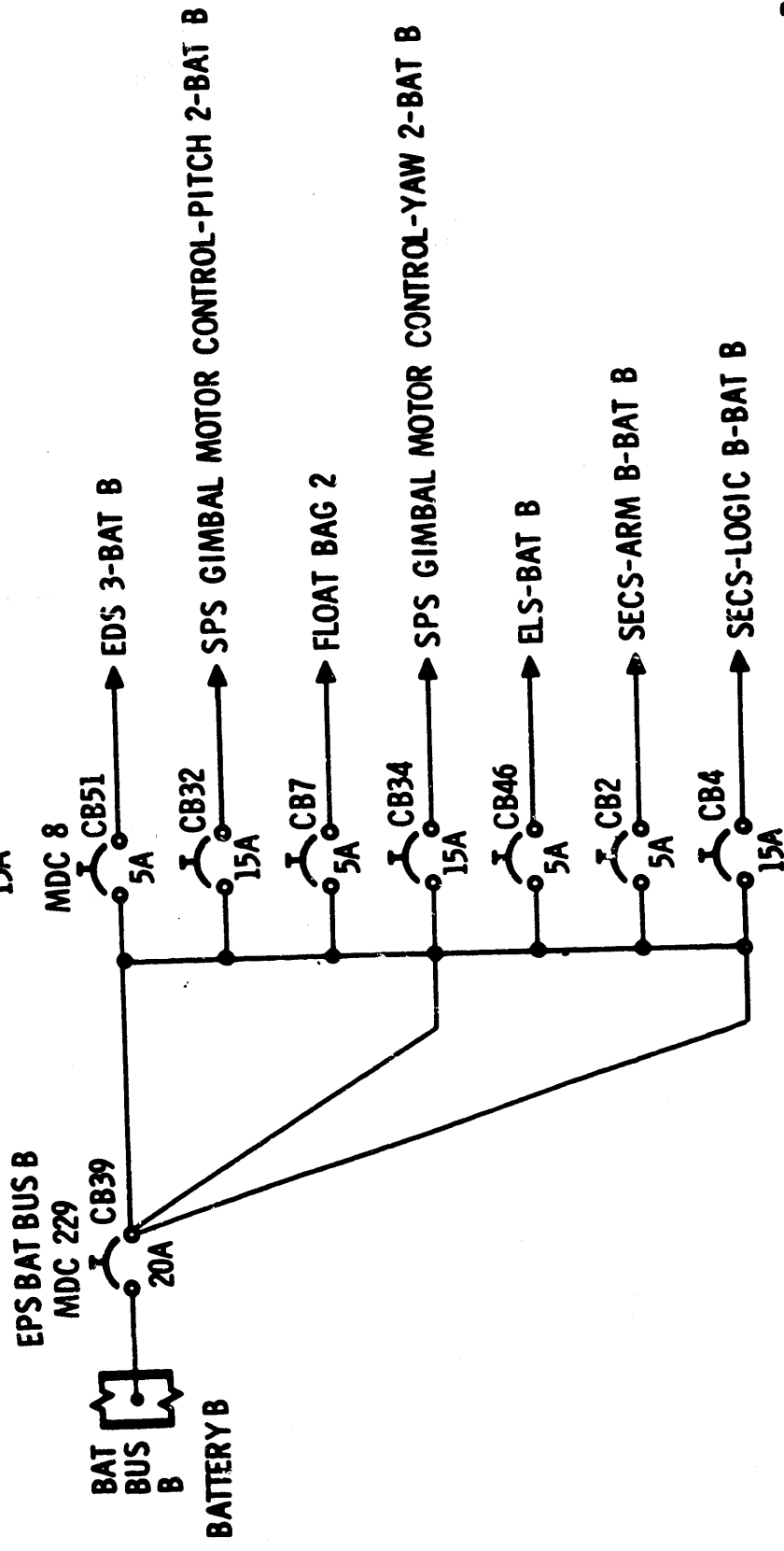
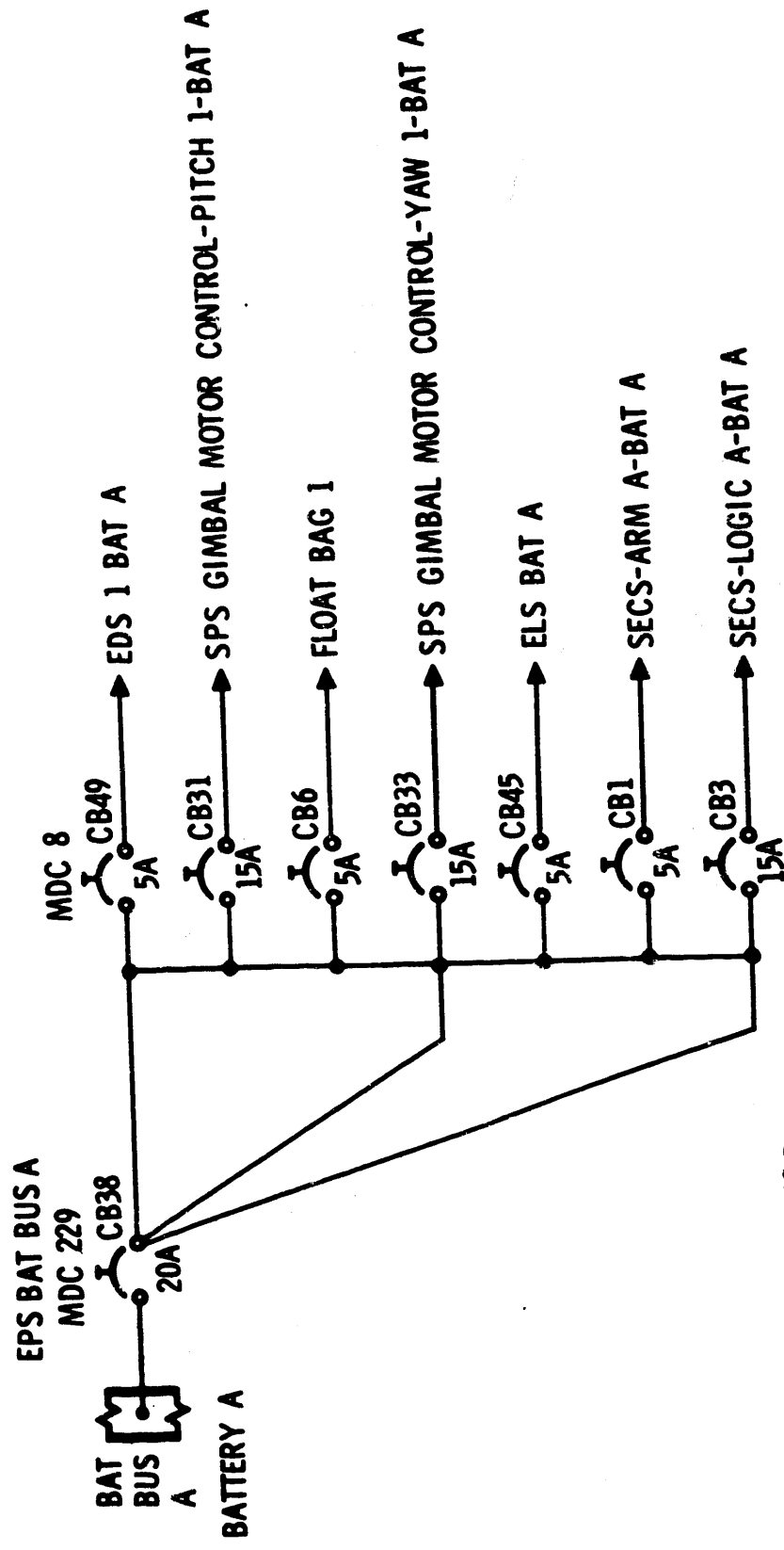
MAIN B DC POWER

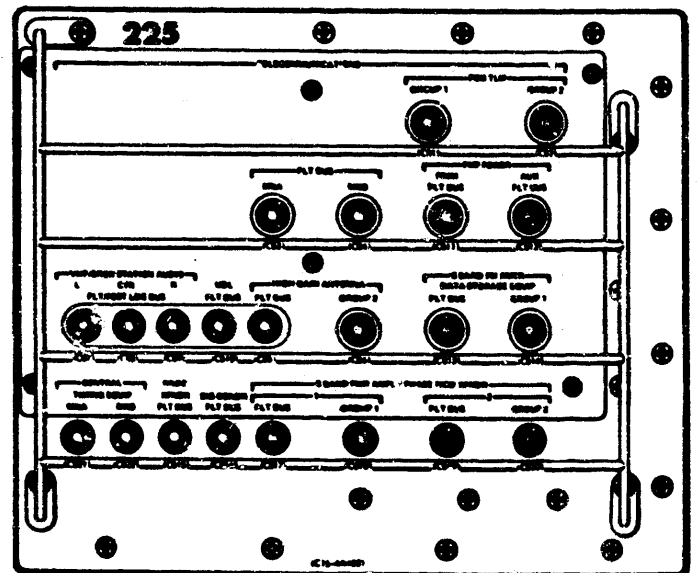
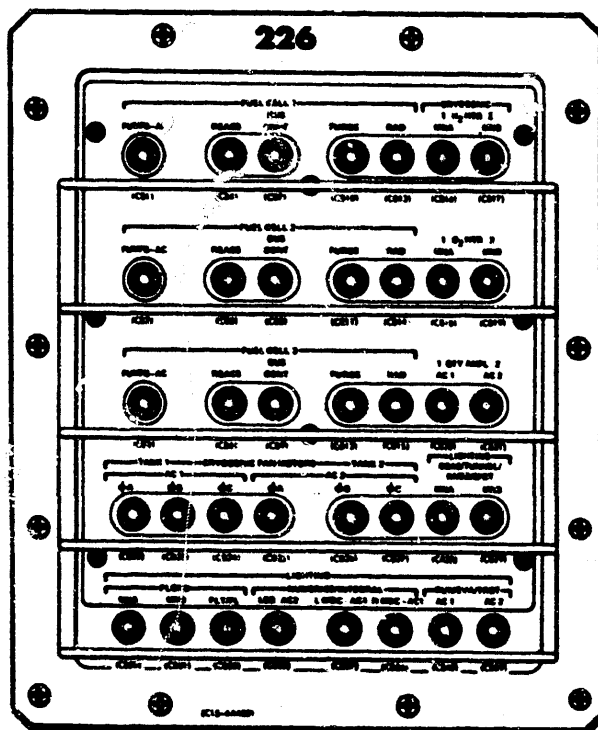
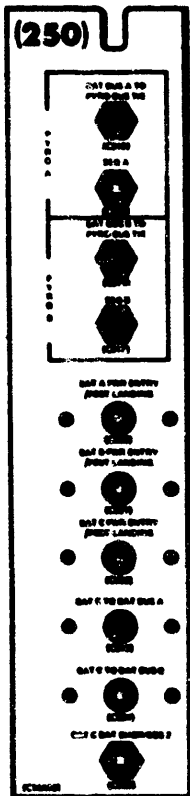
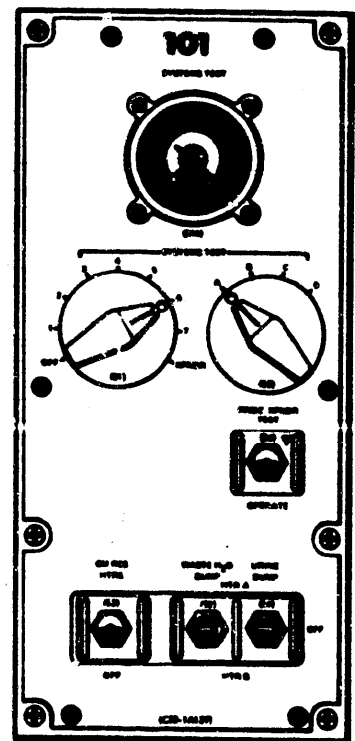
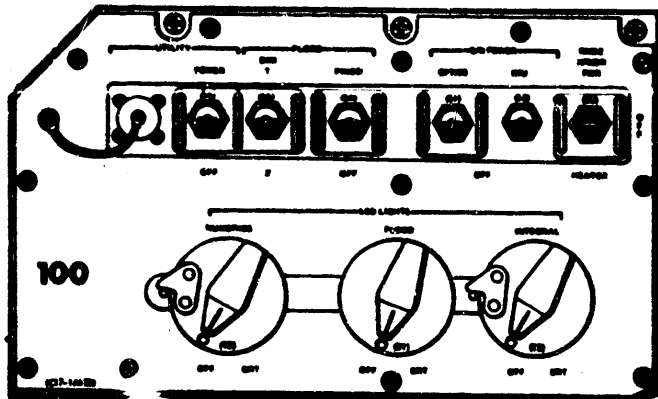
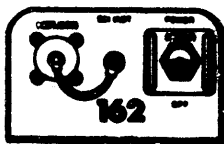
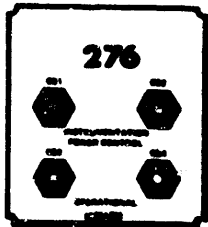
CSM 106 AND SUBS

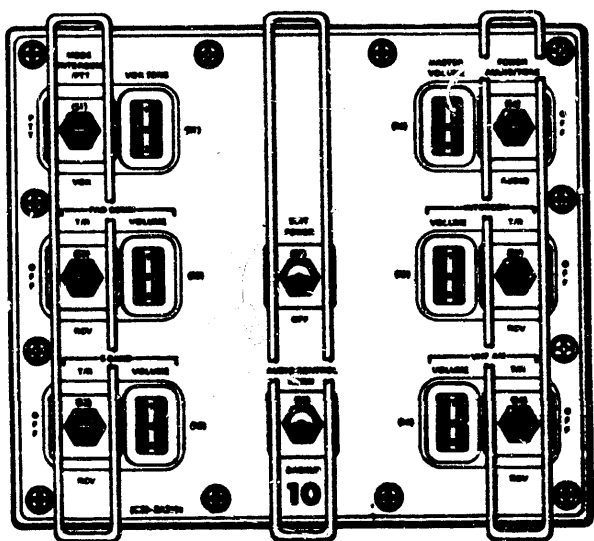
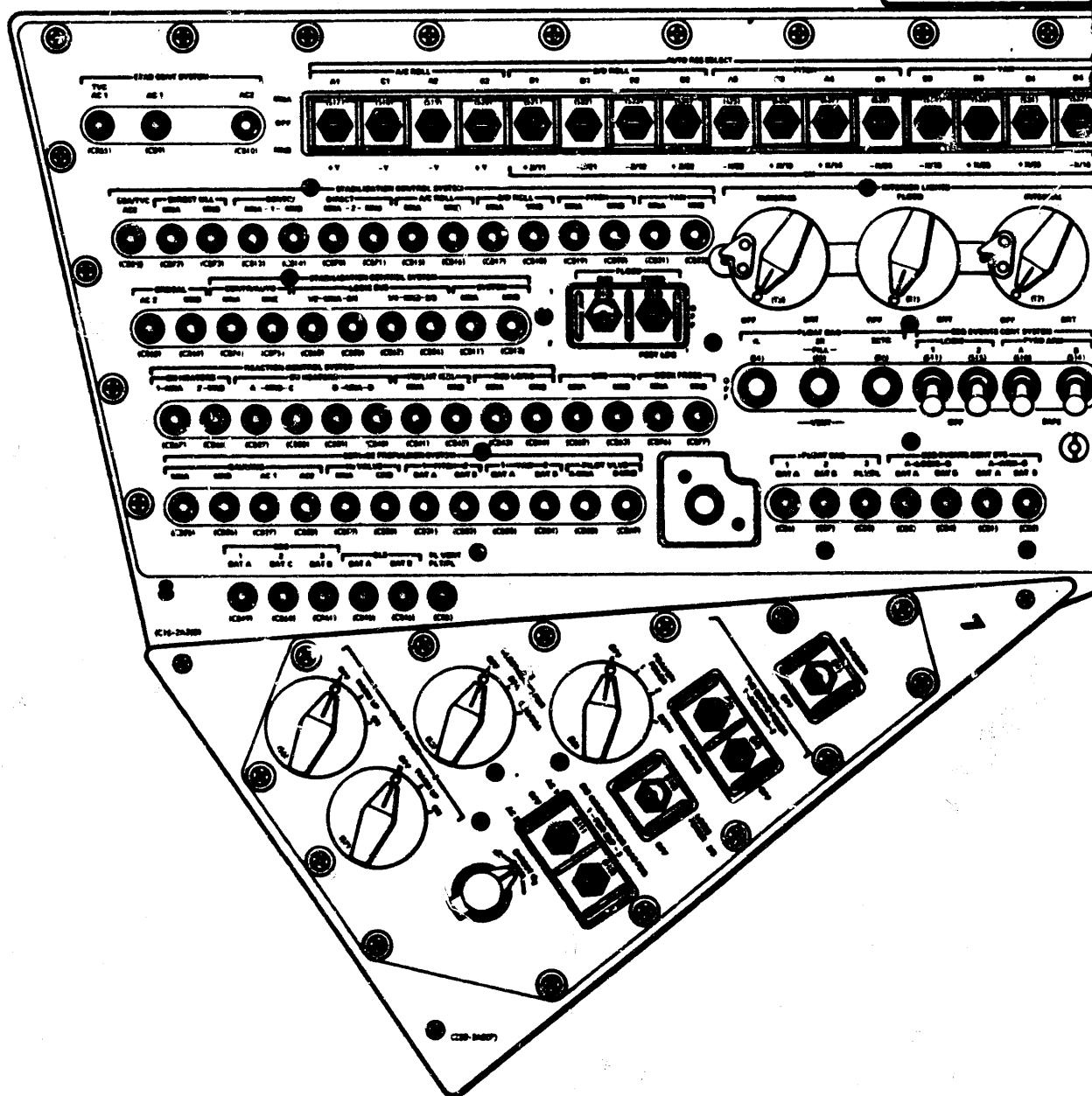
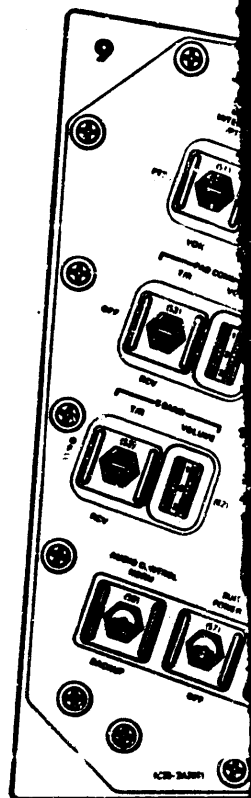
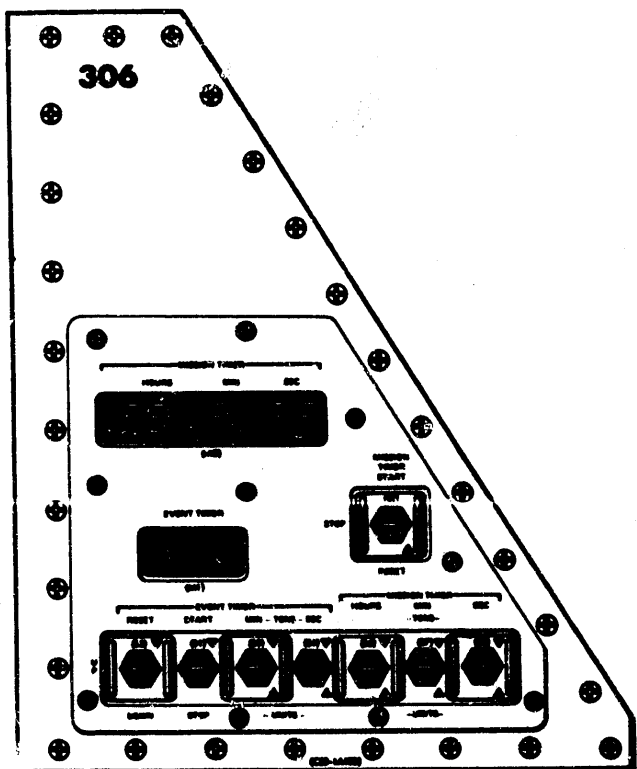


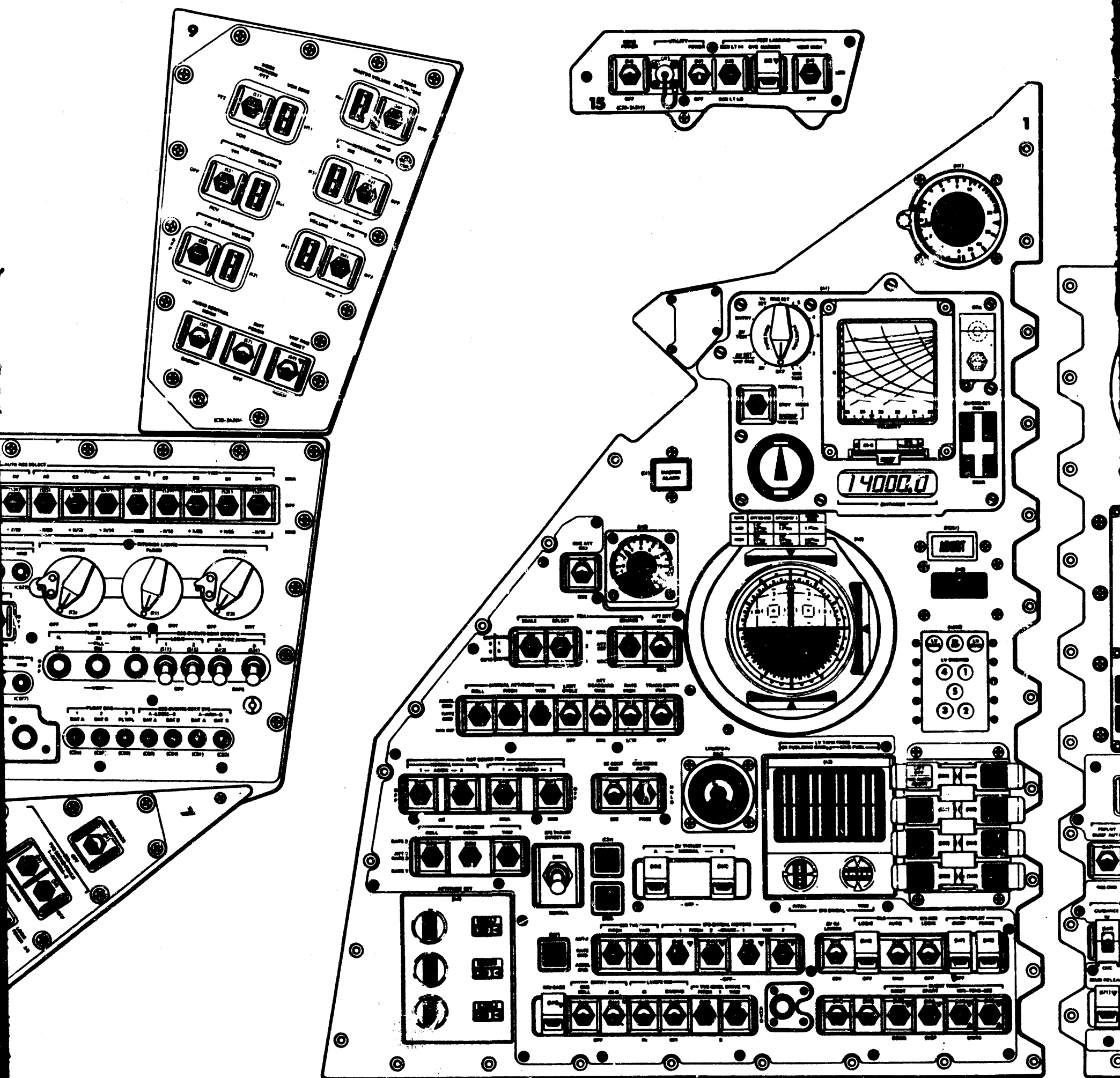
BATTERY A&B POWER

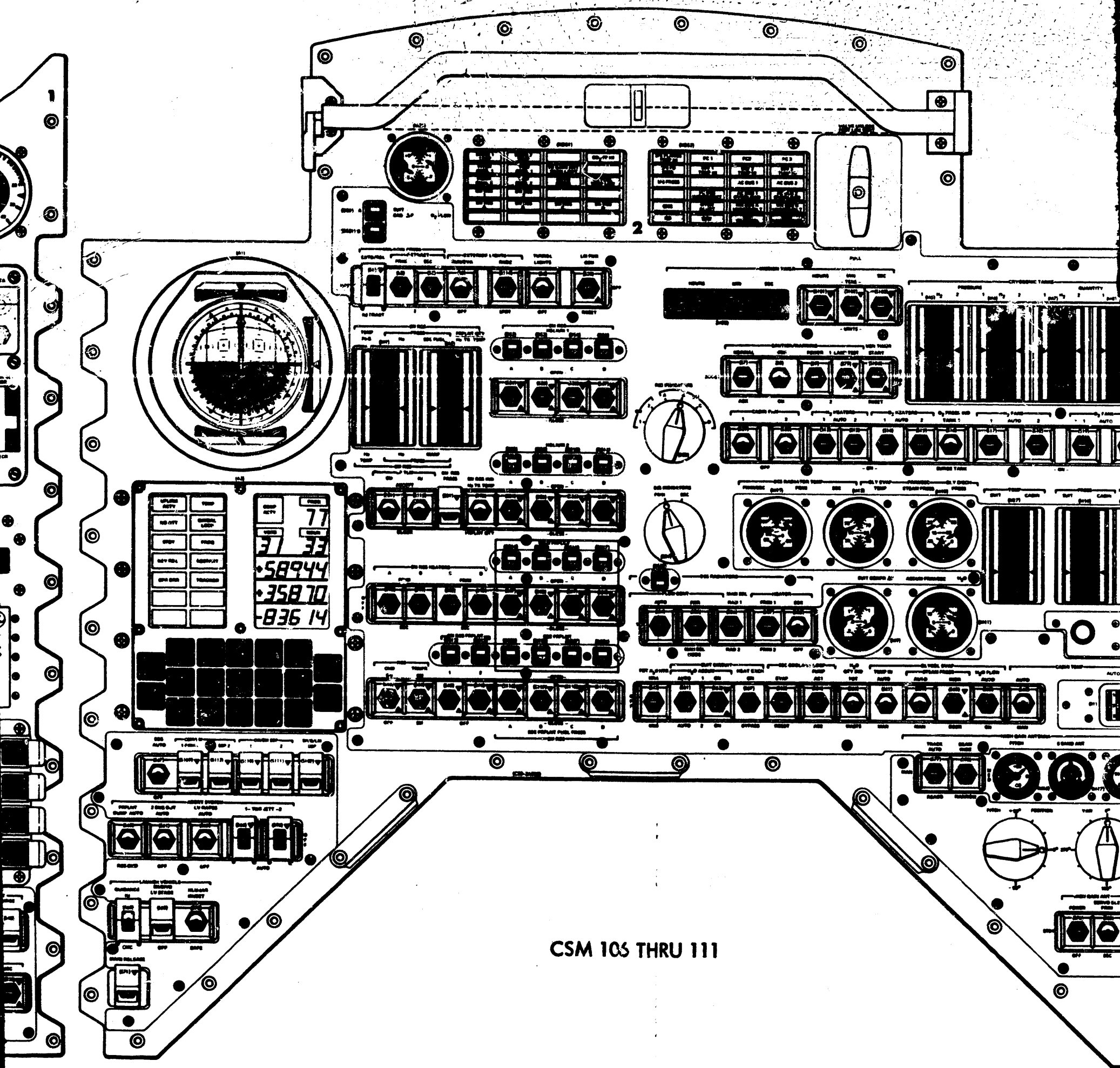
CSM 101 & SUBS



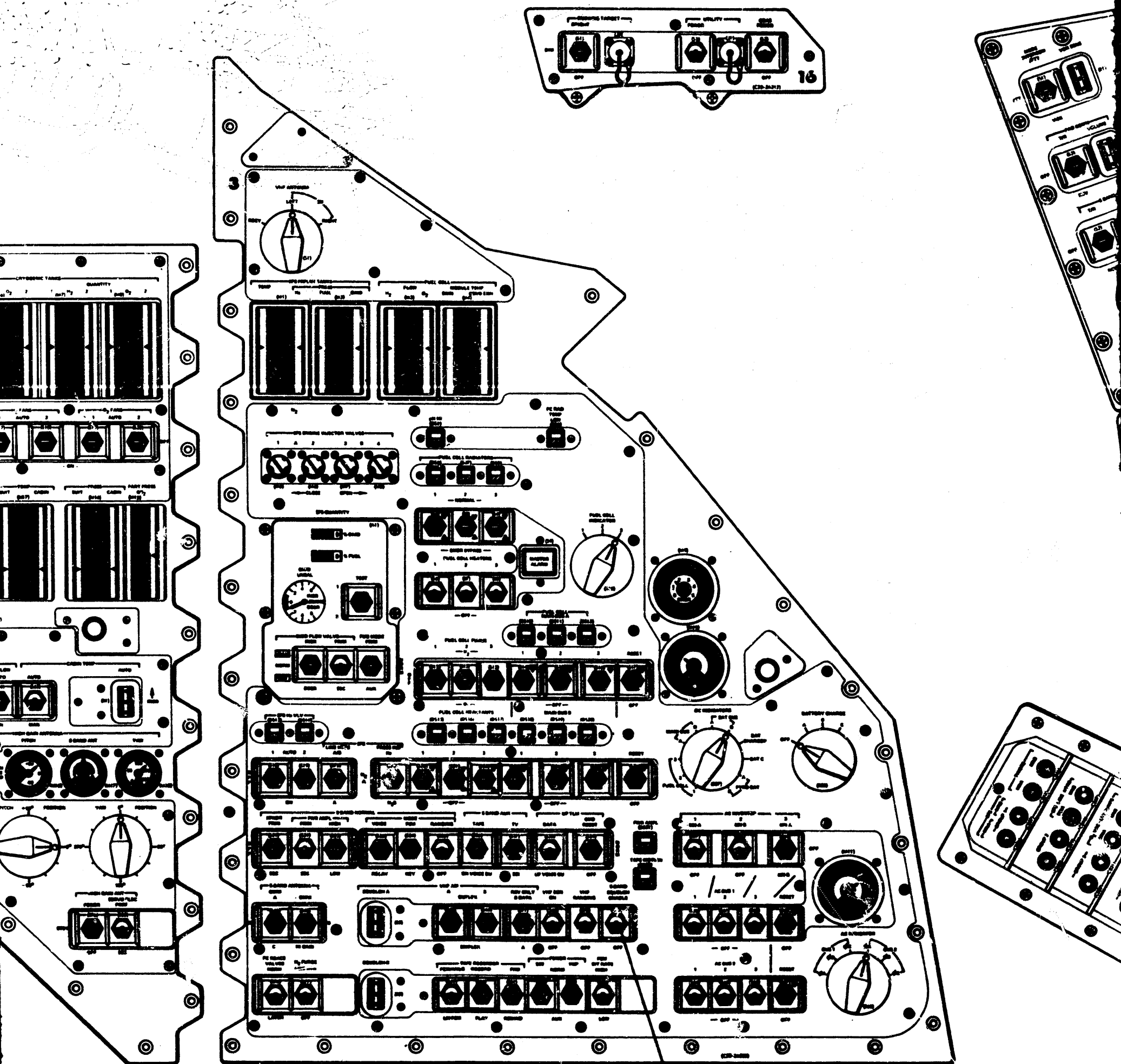




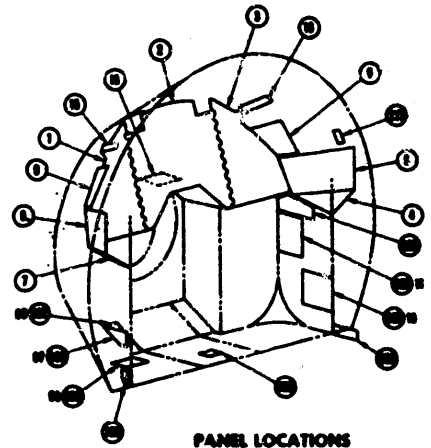
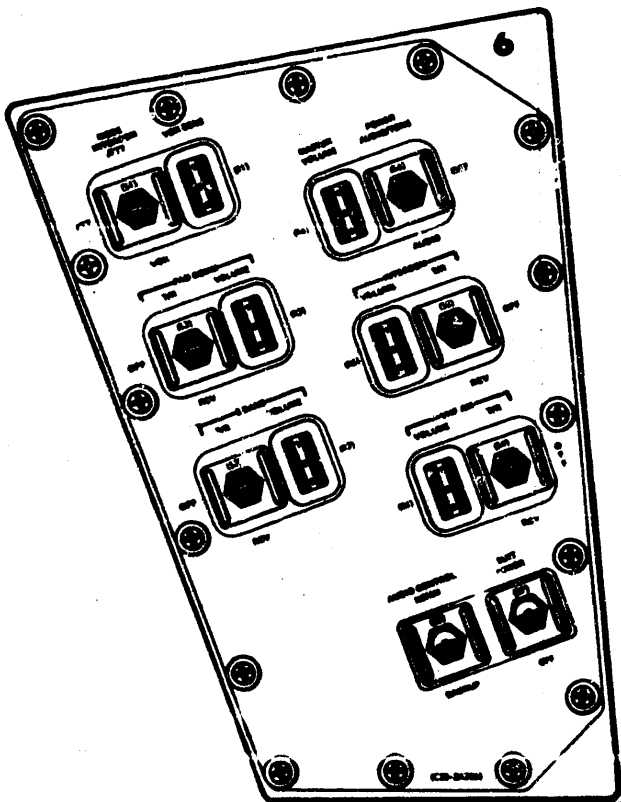




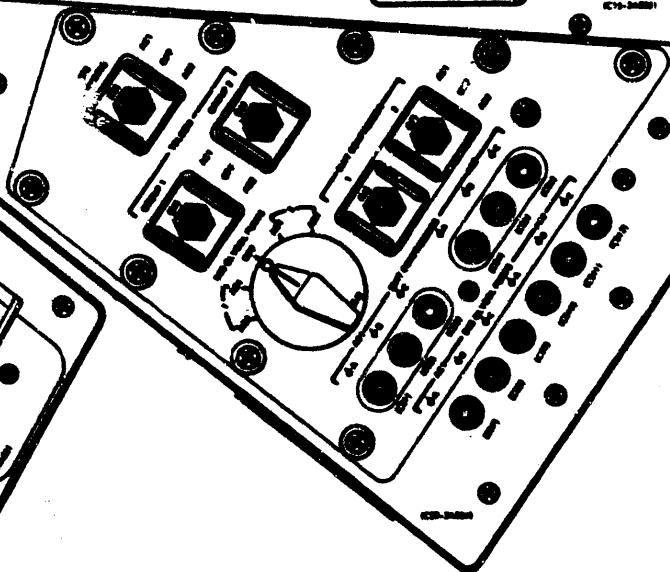
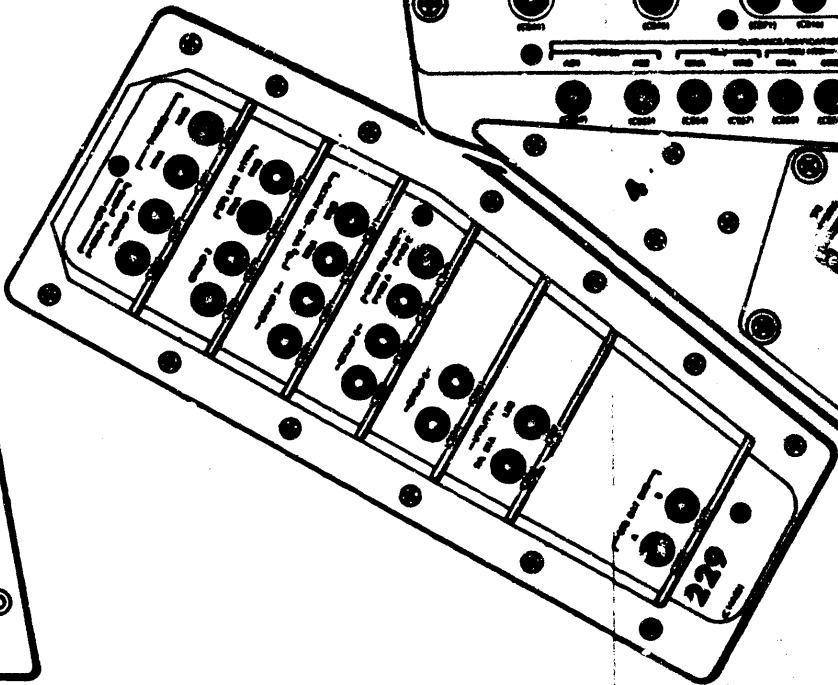
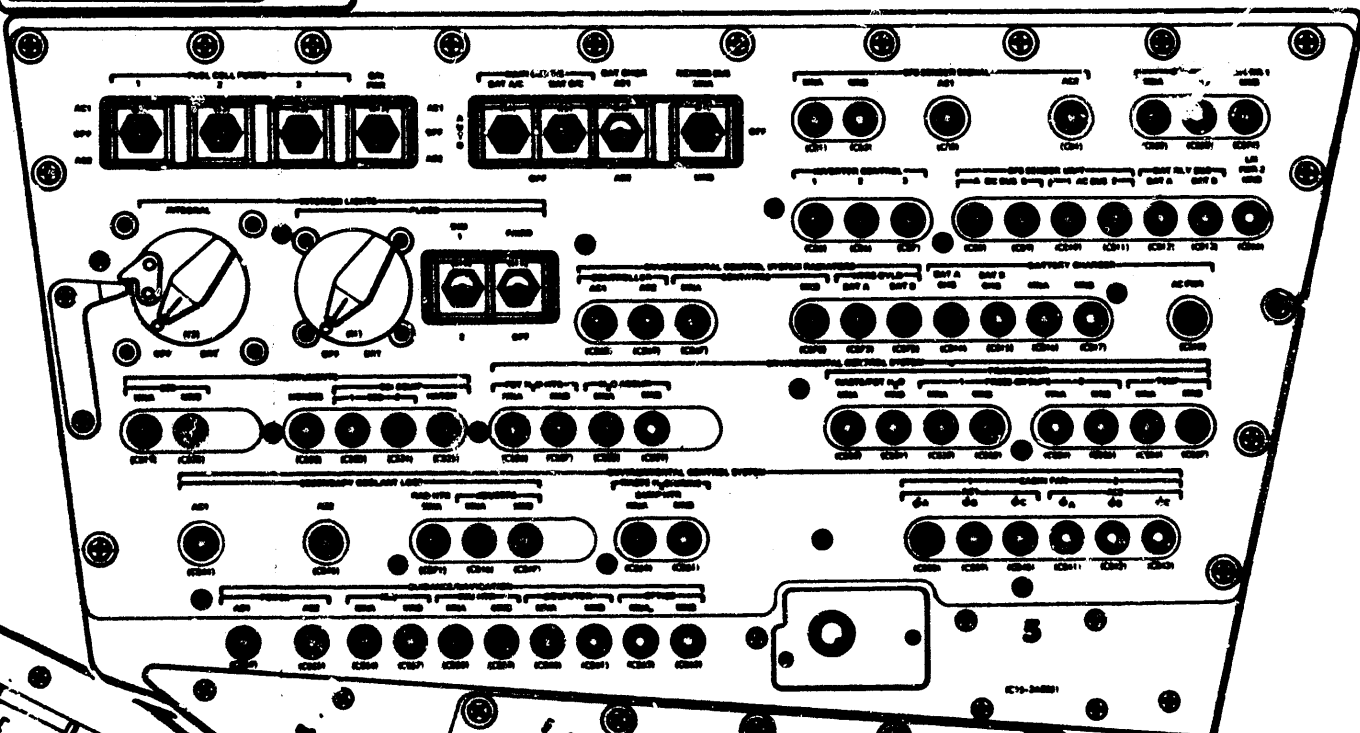
CSM 103 THRU 111



CSM 108 AND SUBS

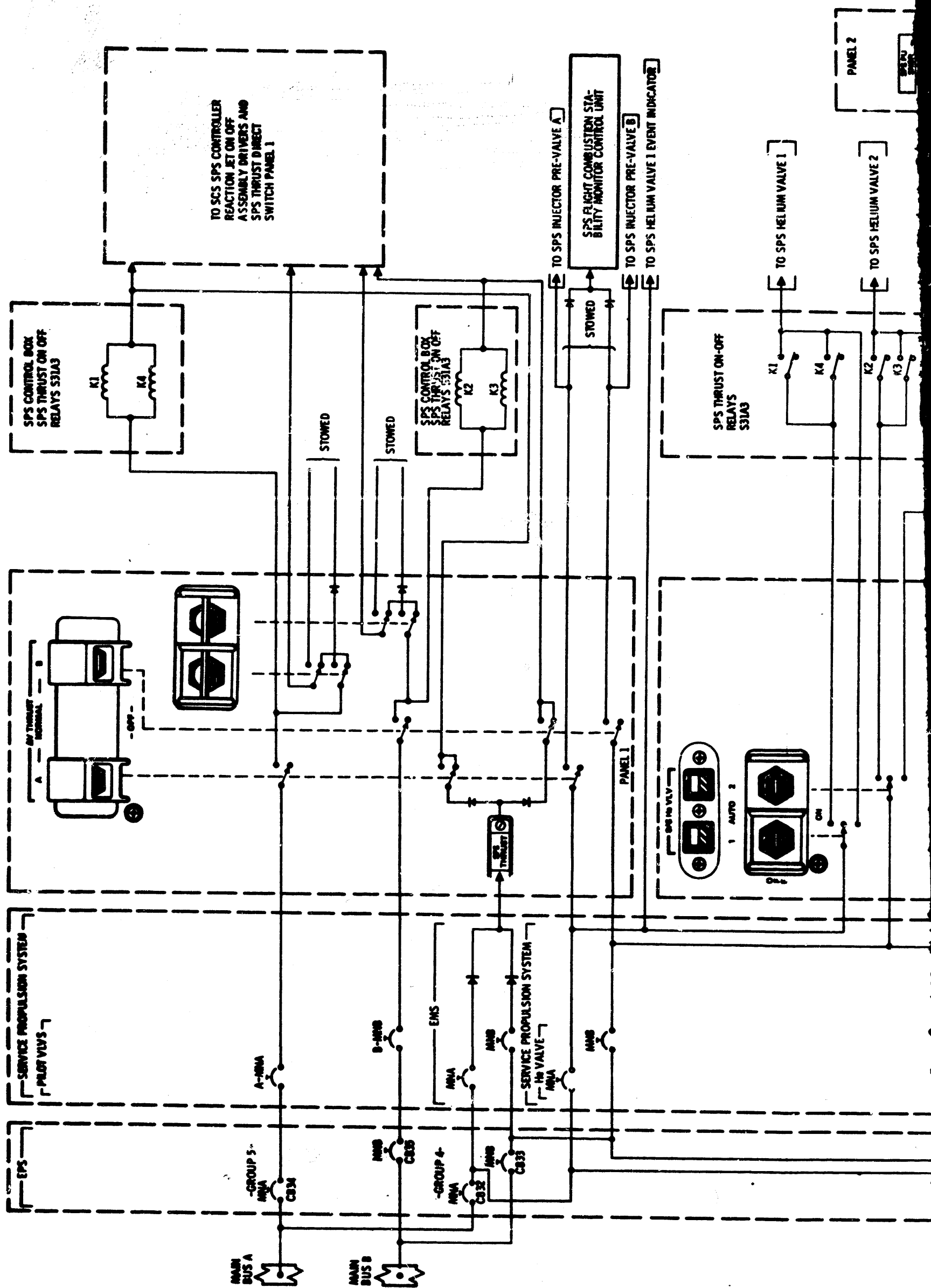


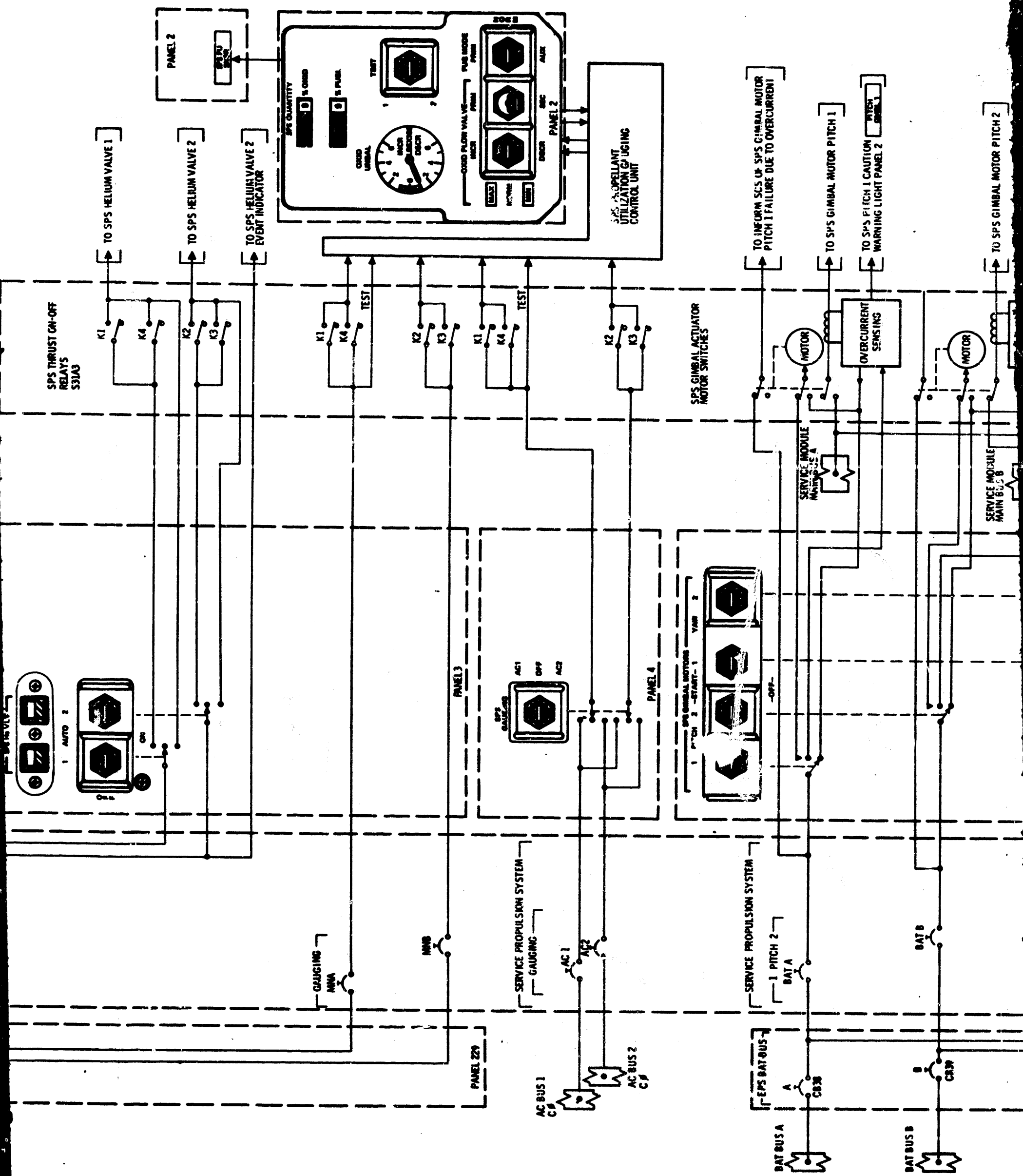
PANEL LOCATIONS

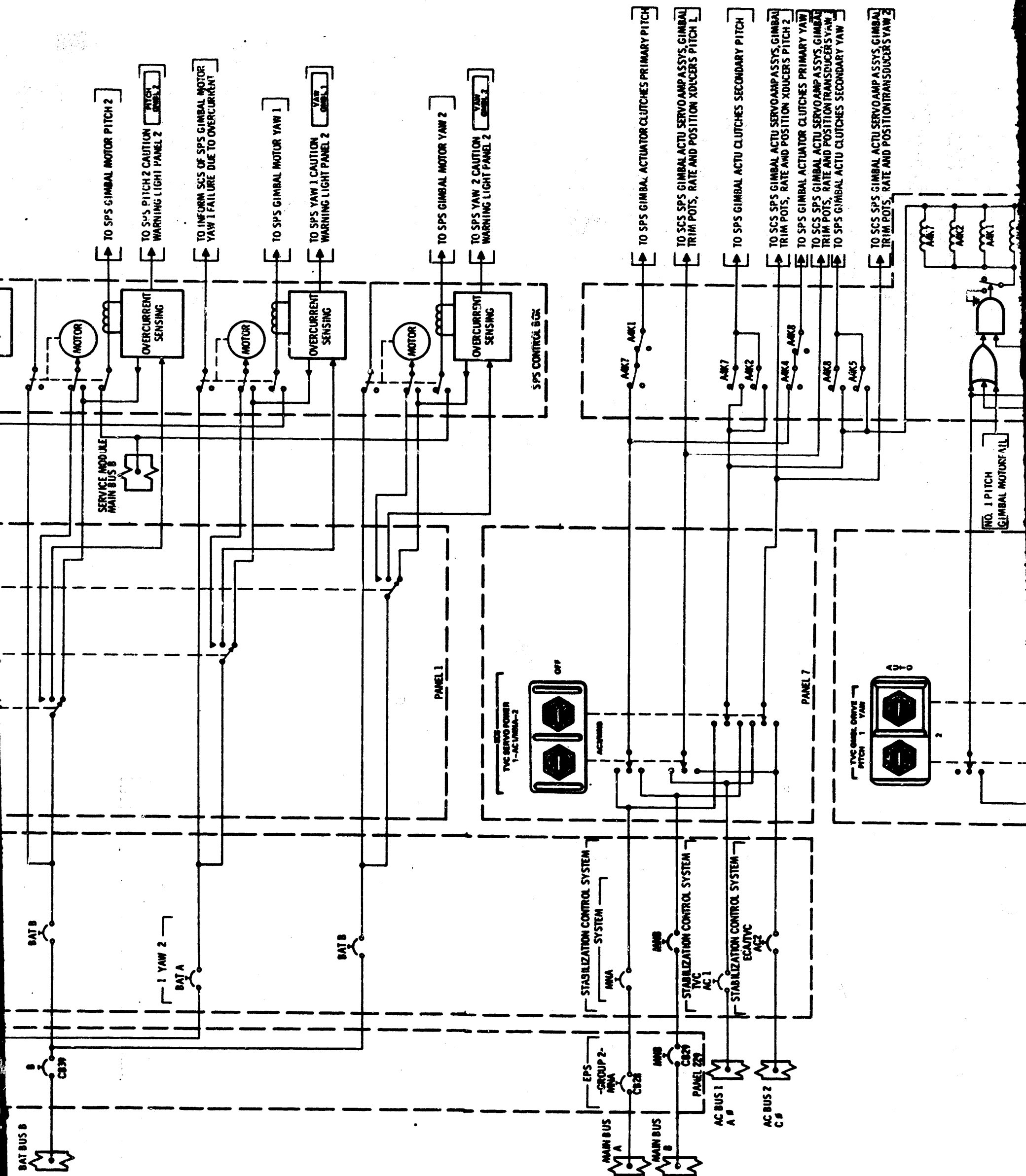


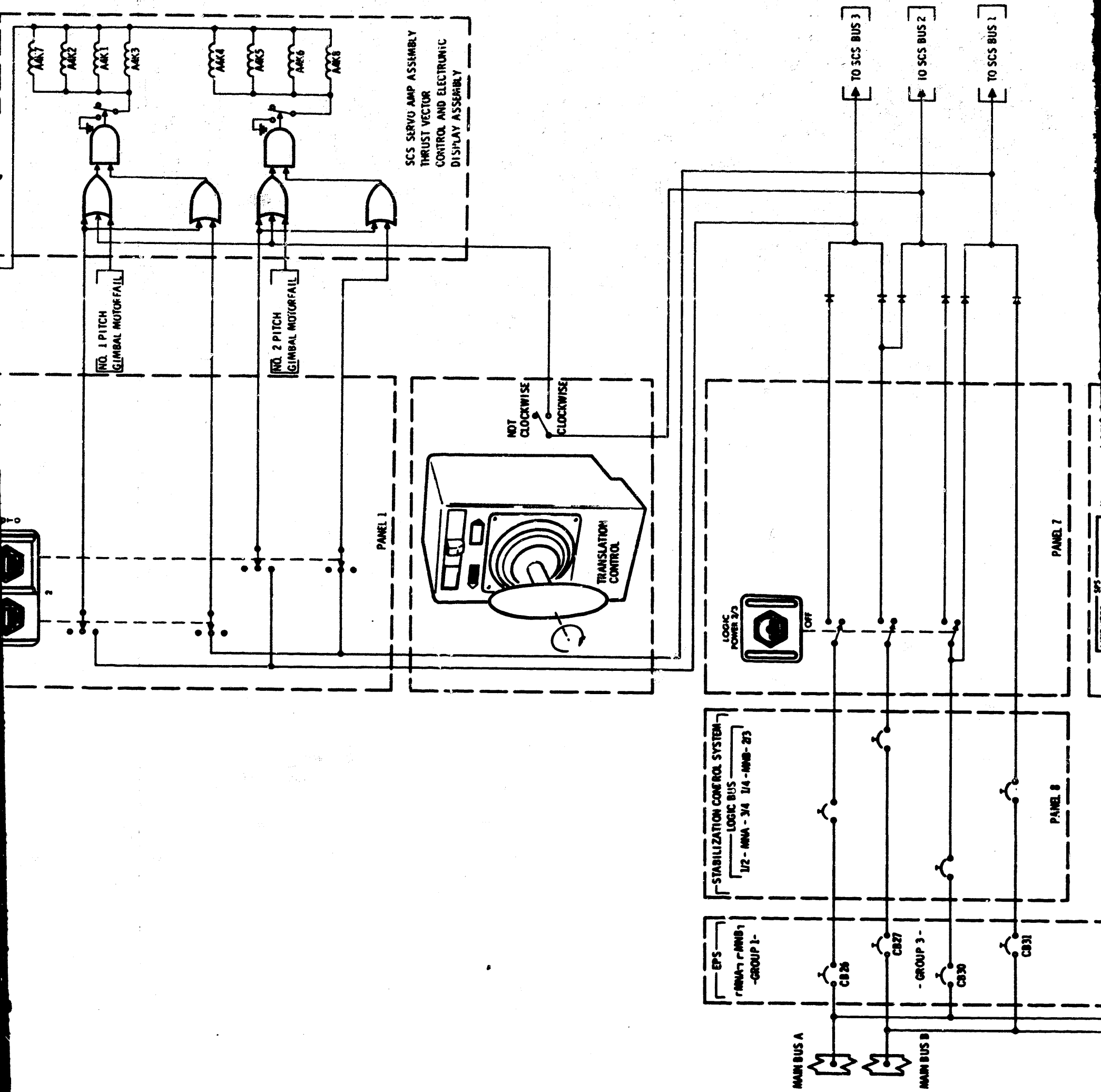
1. ALL NUMBERS AND LETTERS ENCLOSED WITH PARAGRAPHS DO NOT APPEAR ON PANELS.
2. SWITCH POSITIONS ARE SHOWN FOR CONVENIENCE ONLY AND DO NOT NECESSARILY REFLECT OPERATING POSITIONS AND/OR SWITCH PHASES. TWO POSITION SWITCHES ARE SHOWN IN THE DOWN POSITION. THREE POSITION SWITCHES ARE SHOWN IN CENTER POSITION.
3. THIS IS AN ARRANGEMENT AND NOMENCLATURE CONTROL DUG FOR DESIGN REQUIREMENTS AND FOR USE BY OTHER DEPTS AS A GUIDE. ONLY RELEASED DESIGN DRAWINGS AND DOCUMENTS SHALL BE USED FOR APL, MOCK-UP, SIMULATOR, EVALUATOR, TRAINING OR CONSTRUCTION PANELS WHICH ARE SUBJECT TO IMPACT CURRENT DESIGN.
4. THIS DUG REFLECTS A CONFIGURATION AS DEFINED BY APPLICABLE SCPT'S. ASSEMBLY DUGS AND MARKING DUGS AS OF THE DATE OF RELEASE WITH RESPECT TO THE AFFECTED VEHICLES.

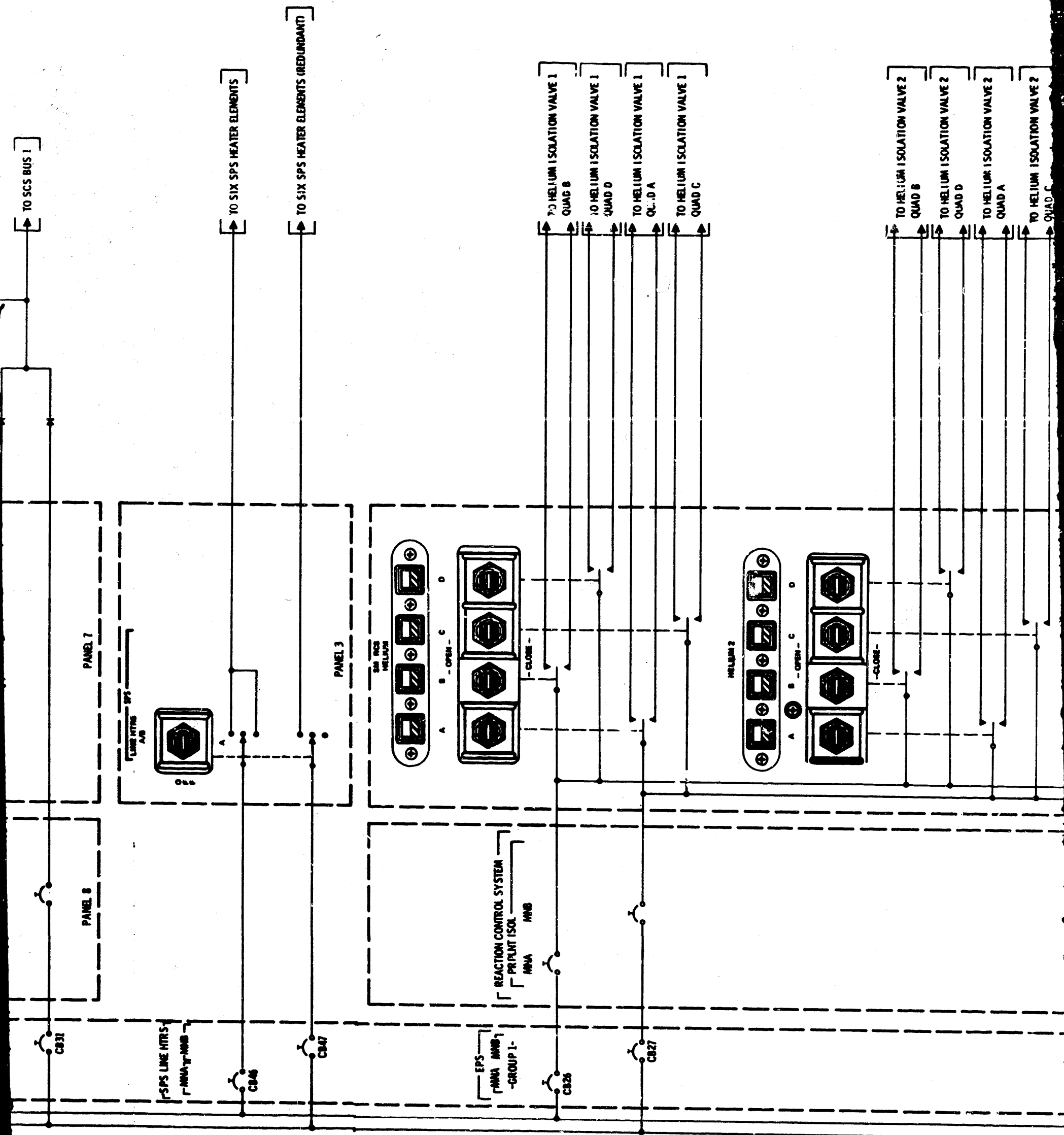








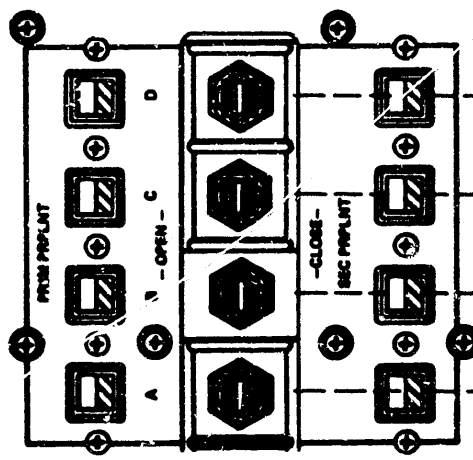




TO HELIUM ISOLATION VALVE 2
QUAD D

TO HELIUM ISOLATION VALVE 2
QUAD A

TO HELIUM ISOLATION VALVE 2
QUAD C

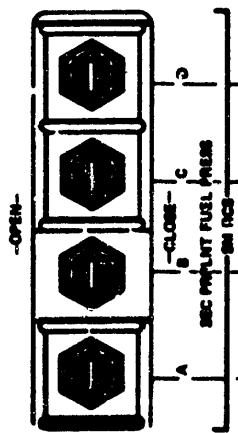


TO PROPELLANT ISOLATION VALVES
(2 FUEL AND 2 OXIDIZER) QUAD B

TO PROPELLANT ISOLATION VALVES
(2 FUEL AND 2 OXIDIZER) QUAD D

TO PROPELLANT ISOLATION VALVES
(2 FUEL AND 2 OXIDIZER) QUAD A

TO PROPELLANT ISOLATION VALVES
(2 FUEL AND 2 OXIDIZER) QUAD C

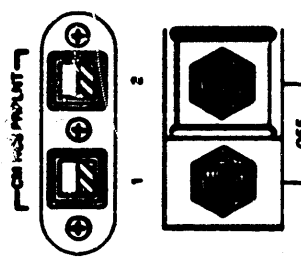


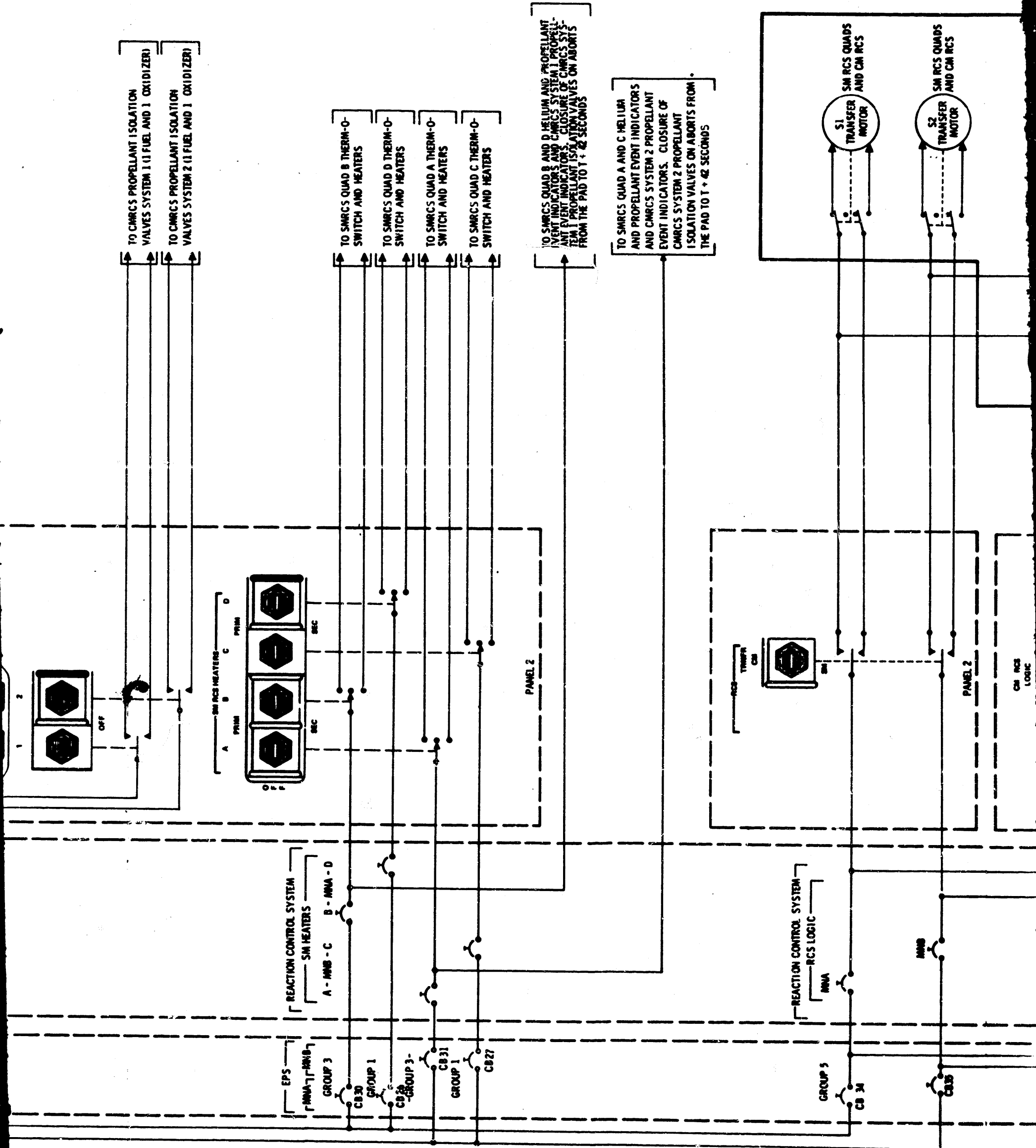
TO SECONDARY PROPELLANT FUEL
PRESSURE VALVE QUAD B

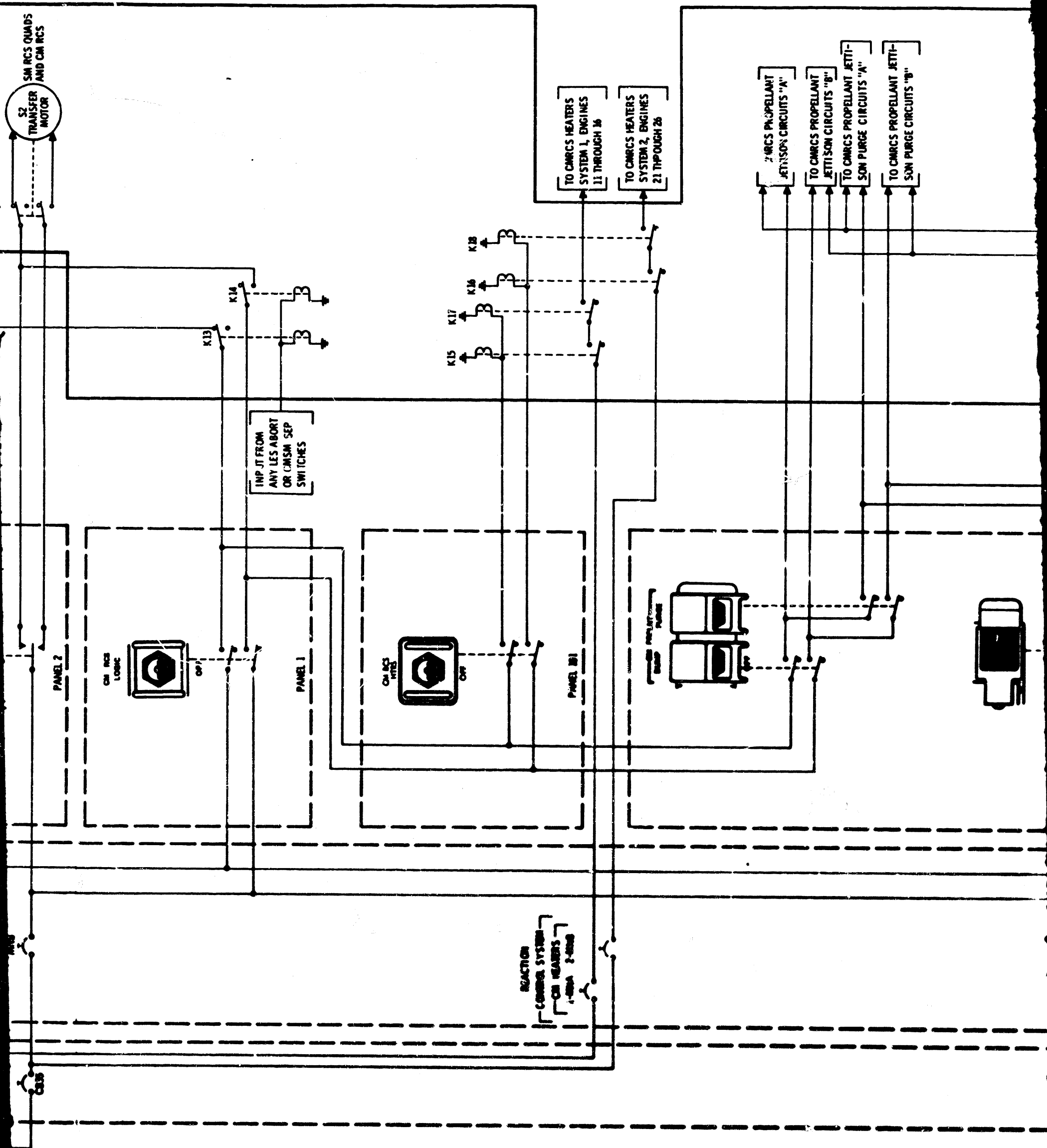
TO SECONDARY PROPELLANT FUEL
PRESSURE VALVE QUAD D

TO SECONDARY PROPELLANT FUEL
PRESSURE VALVE QUAD A

TO SECONDARY PROPELLANT FUEL
PRESSURE VALVE QUAD C







SON PURGE CIRCUITS "B"

HELIUM ISOLATION
PRESSURIZATION
SQUIB VALVES
CMRCS SYSTEM
1 AND 2

LUNAR
DOCKING
EVENTS
CONTROLLER
AND
MASTER
EVENT
SEQUENCE
CONTROLLER

IN PUT
FROM
LES
ABORTS
AND
CMRCS
SEP
SONIC RES

PAGE 1

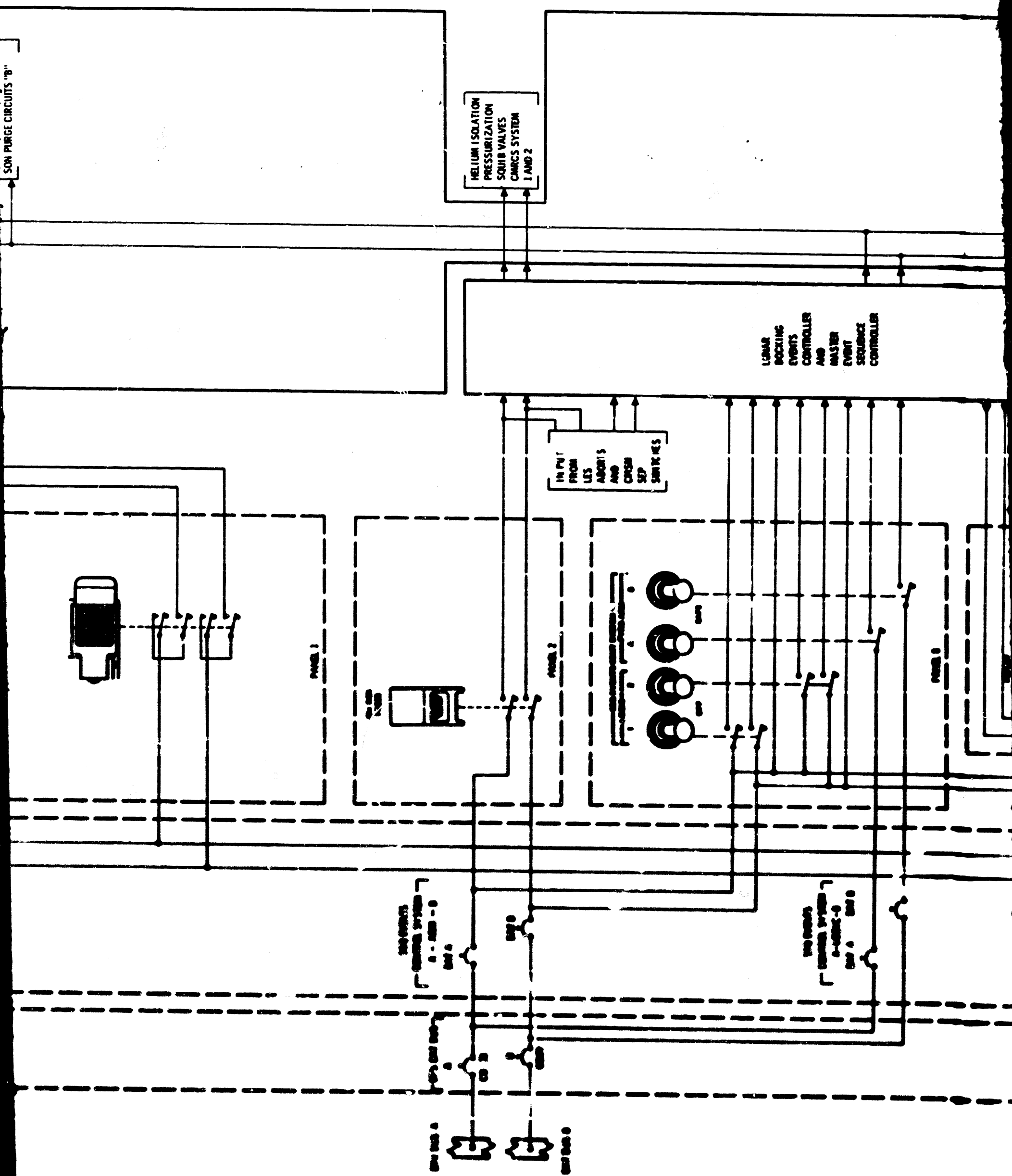
PAGE 2

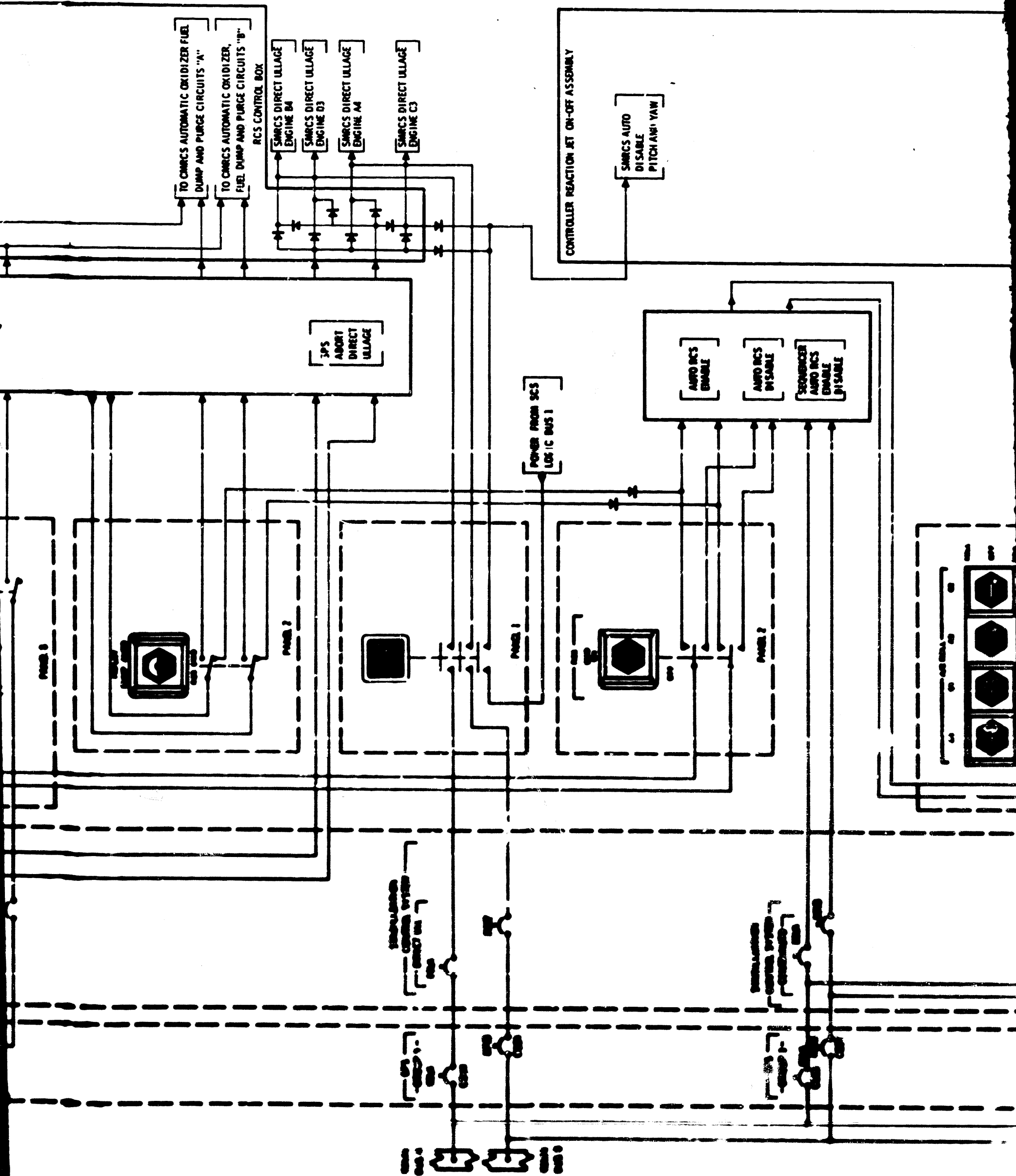
PAGE 3

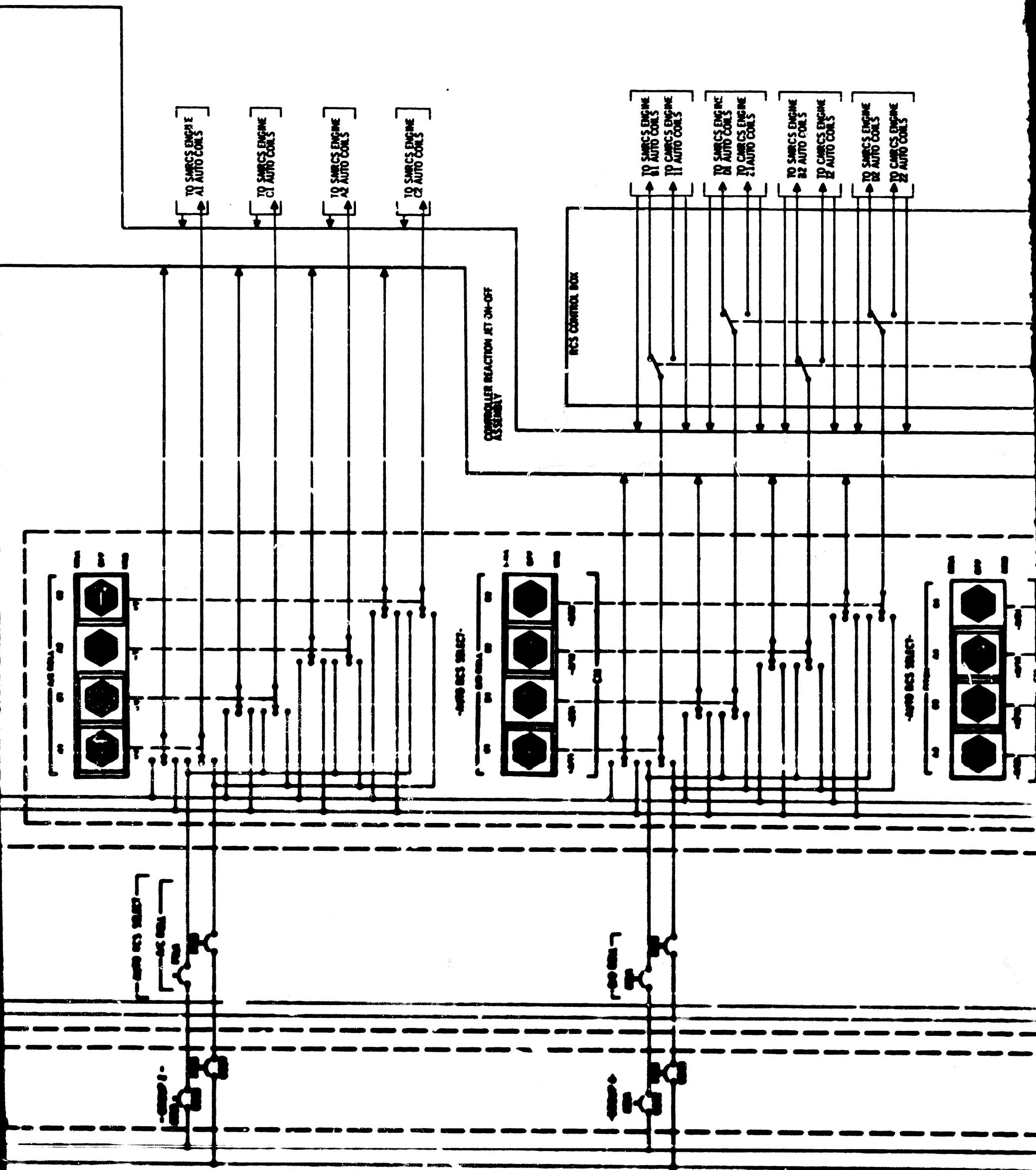
SONIC RES
CMRCS SYSTEM
A - 1000 - 0
SEP 4

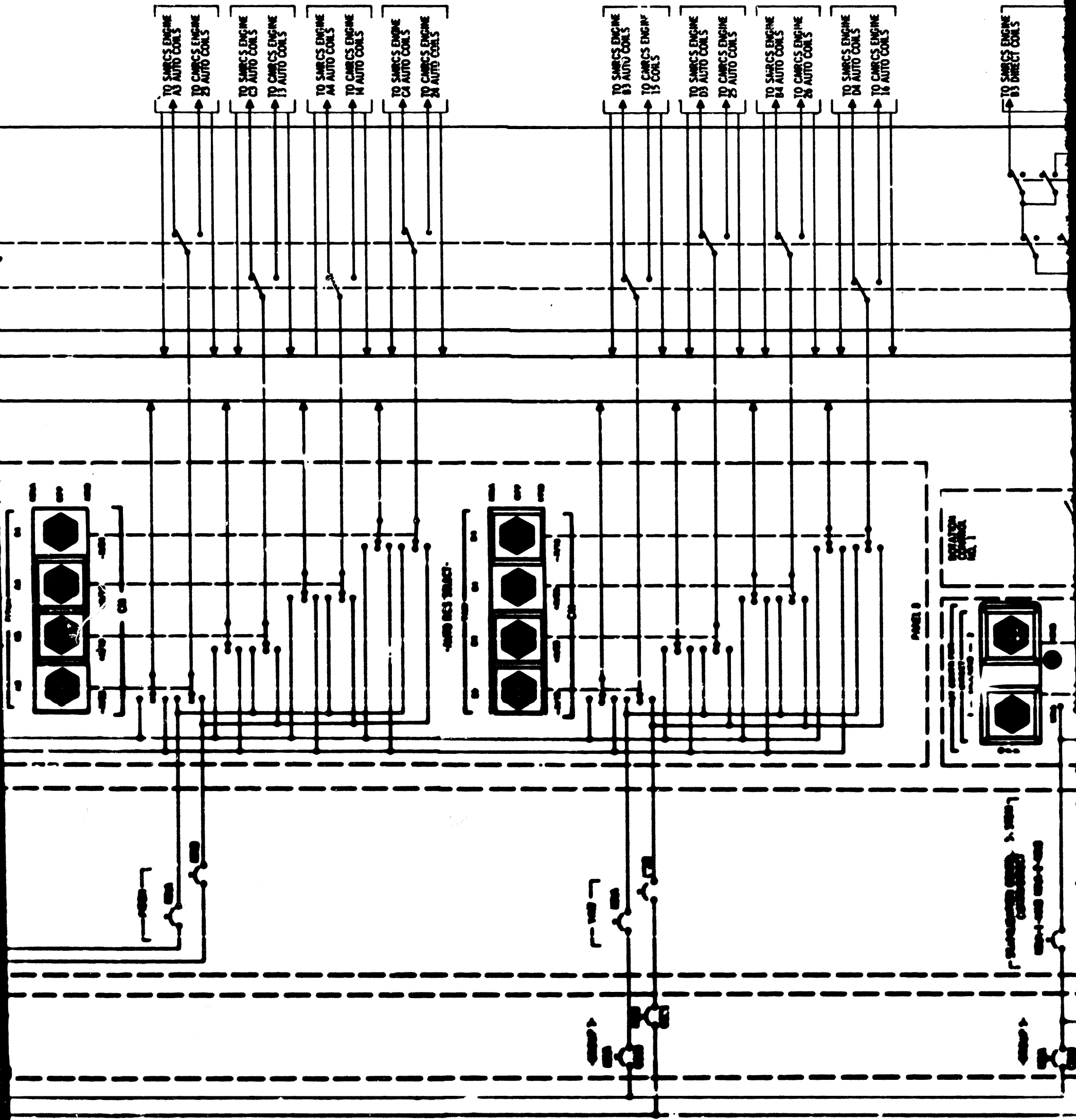
SONIC RES
CMRCS SYSTEM
A - 1000 - 0
SEP 4

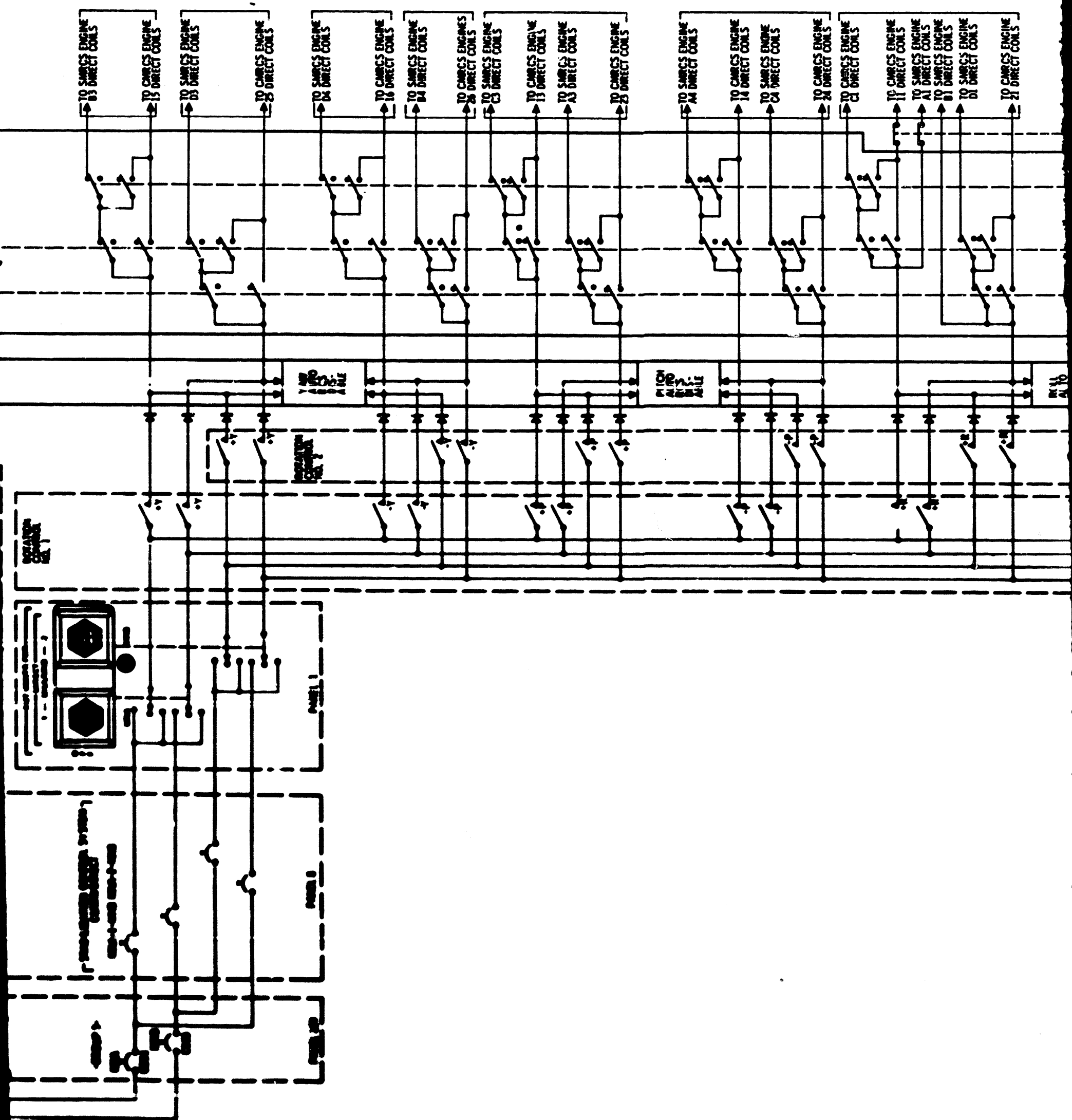
SEP 4
SEP 4
SEP 4

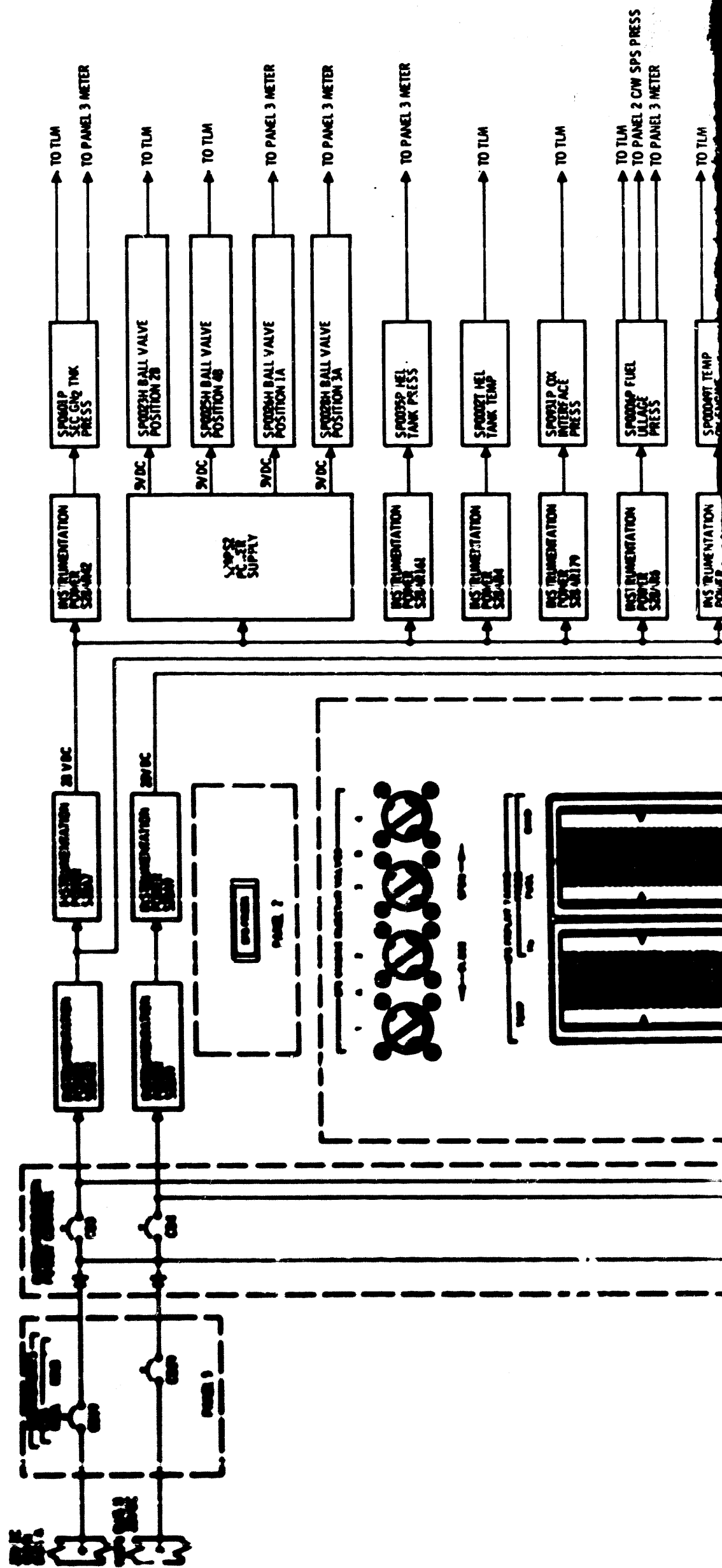


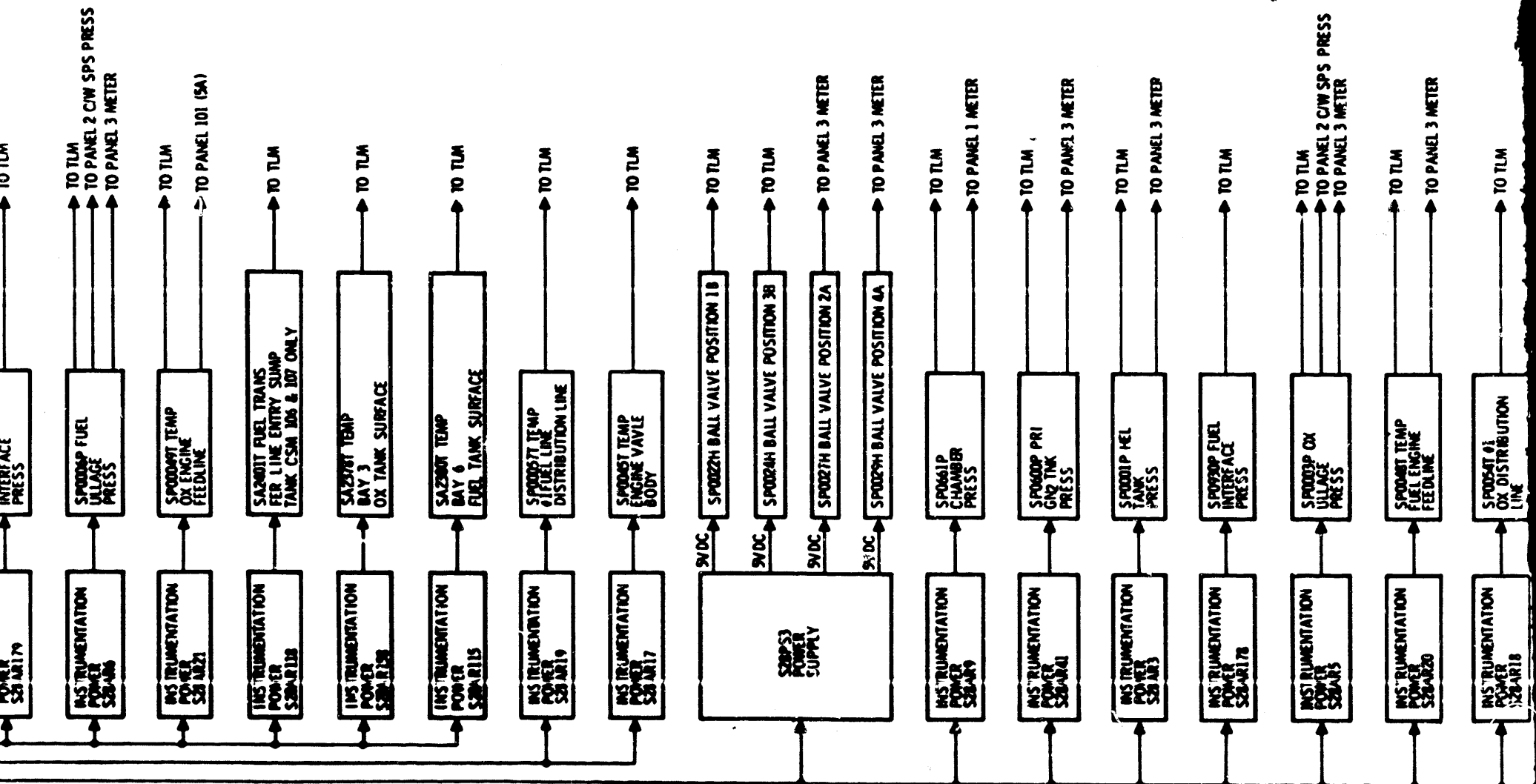
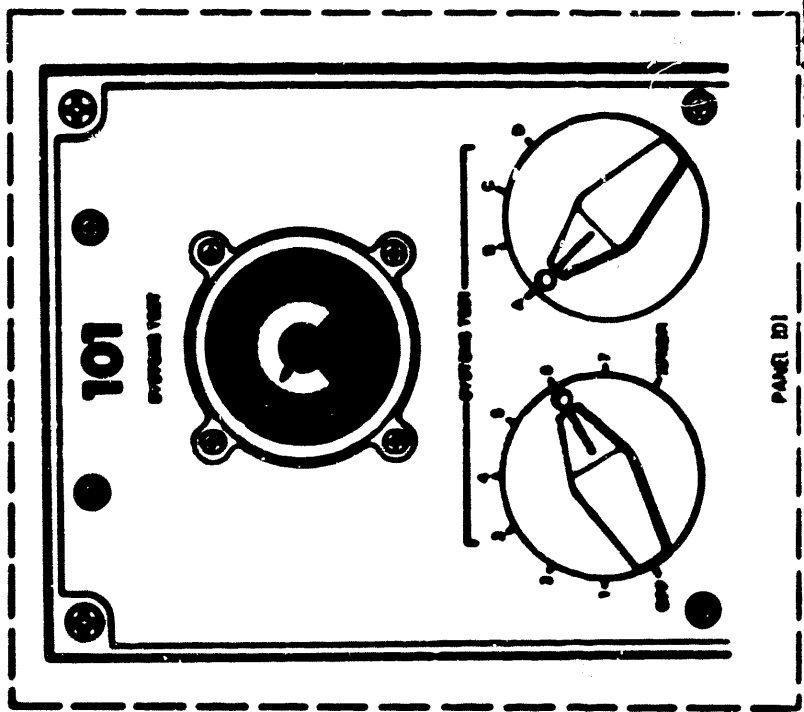
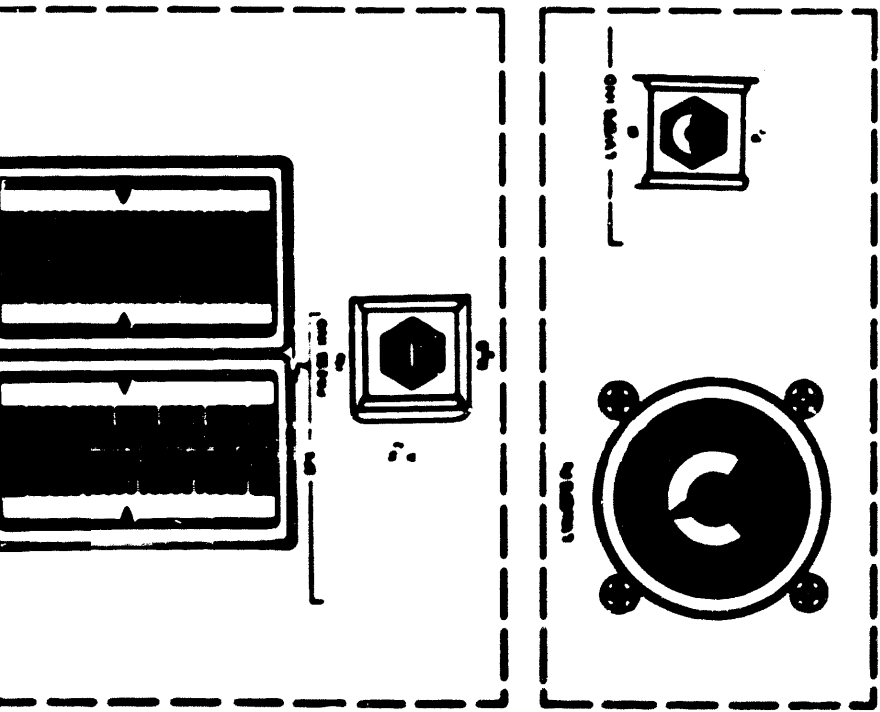


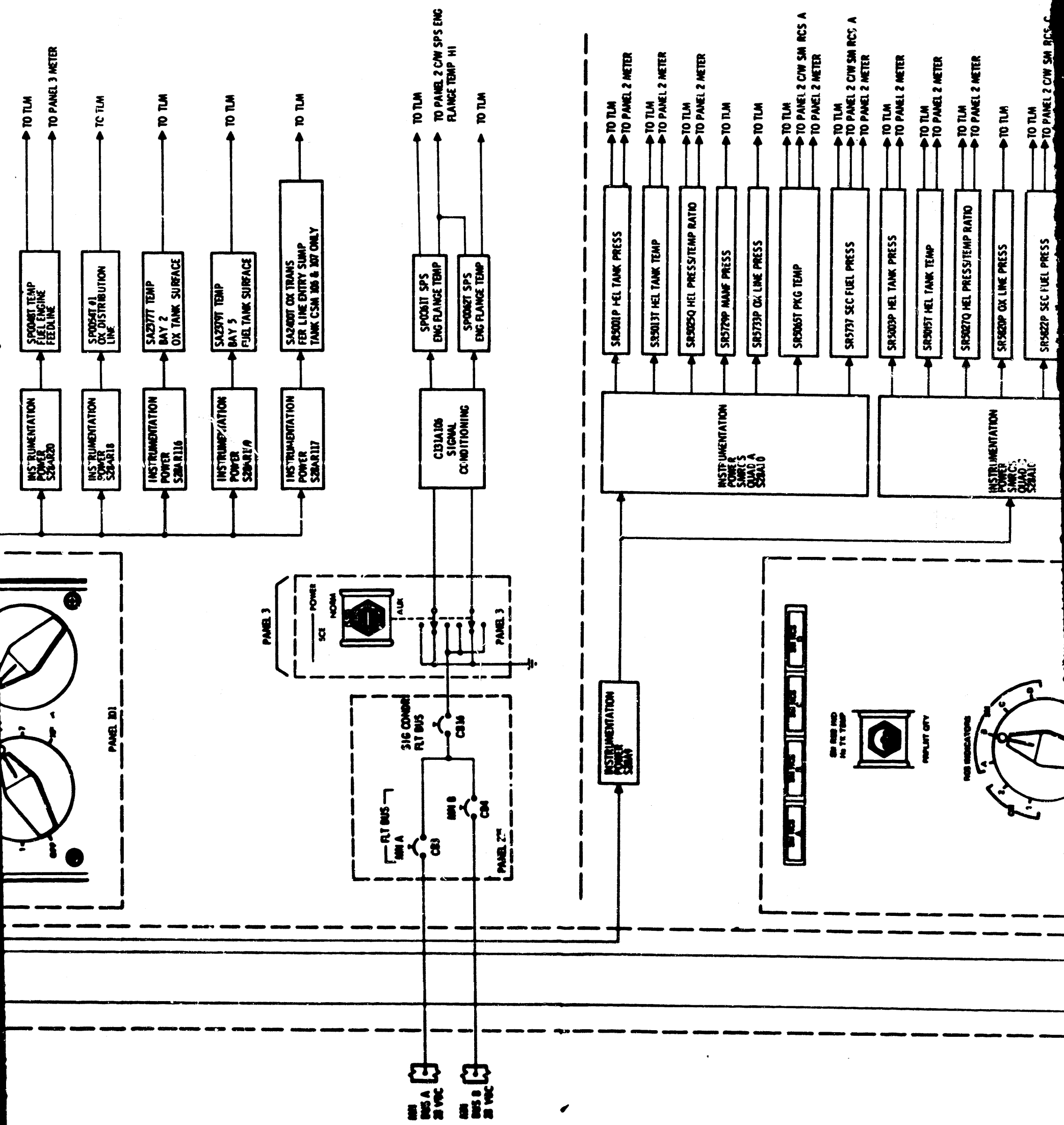


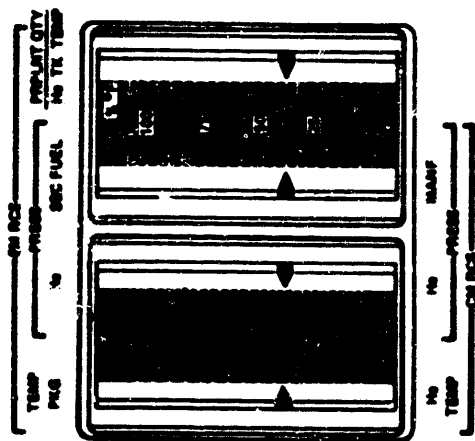
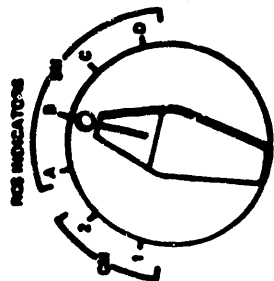




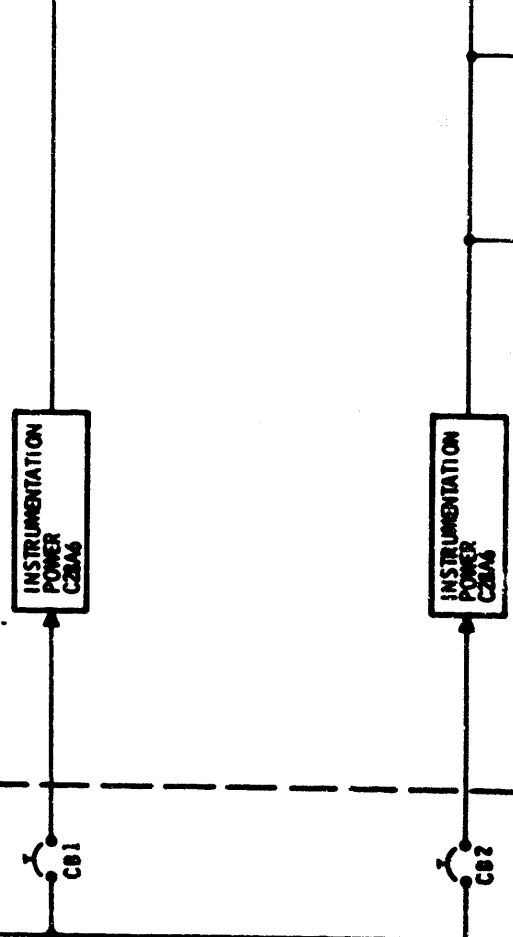
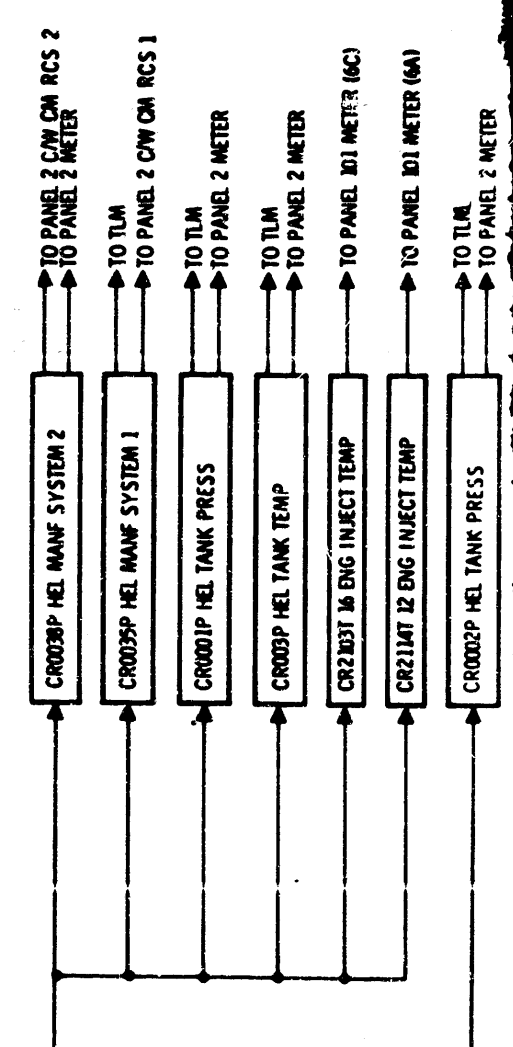
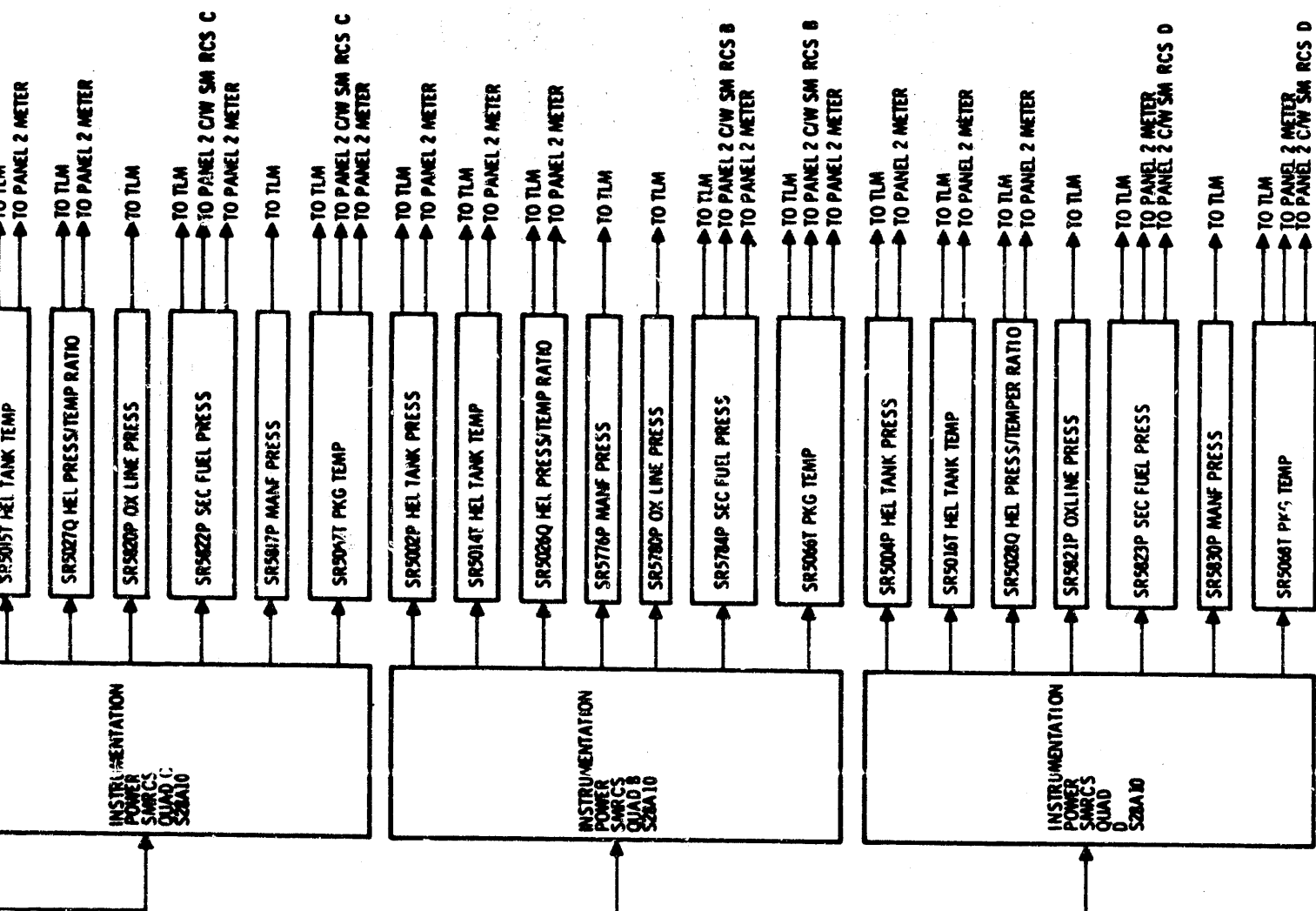


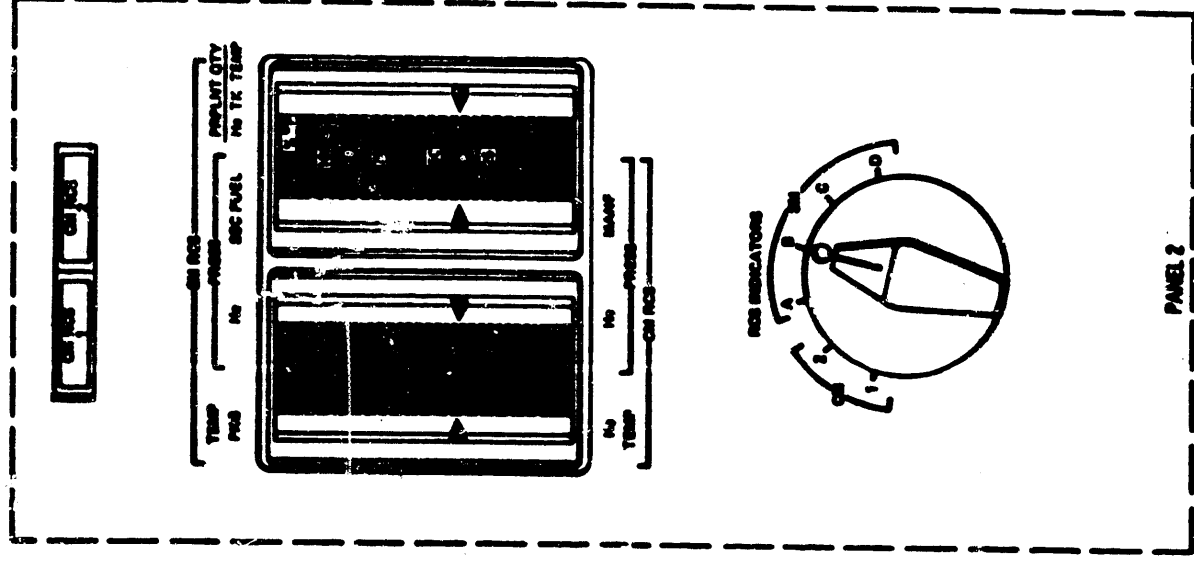
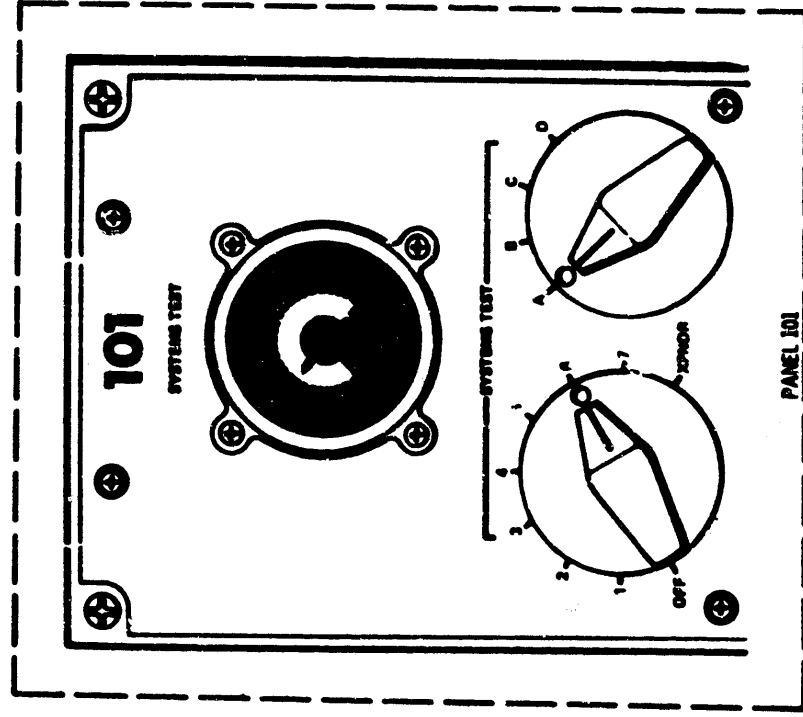
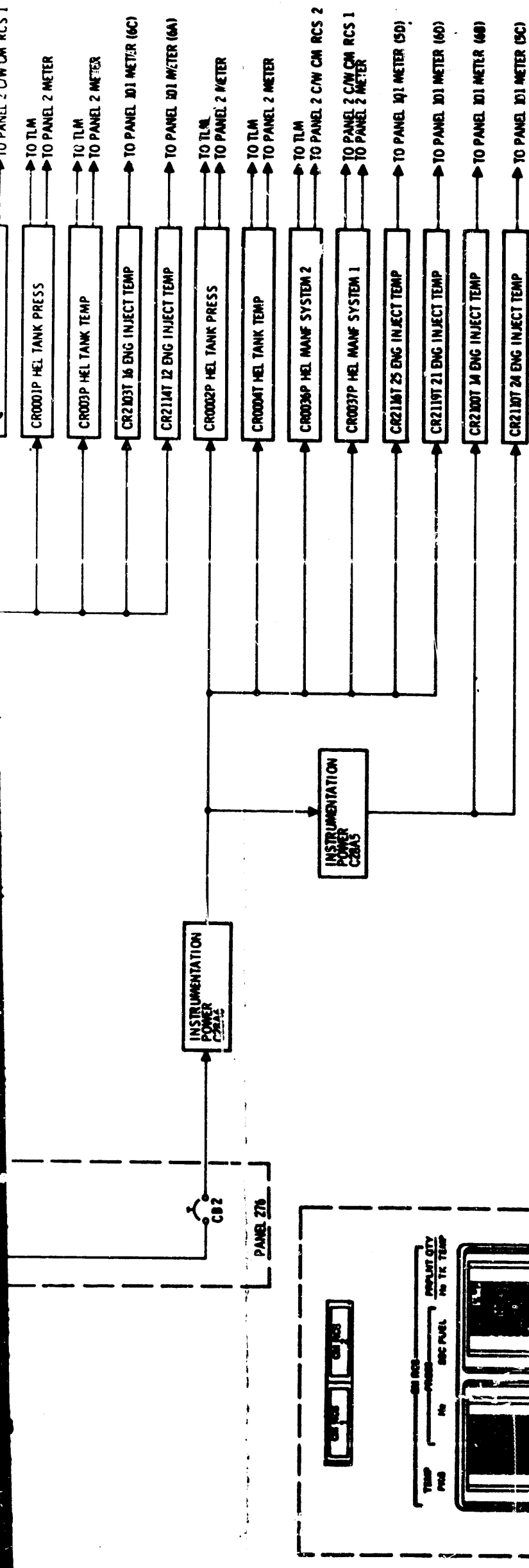






PANEL 2





CSM PROPULSION POWER DISTRIBUTION

(CSM106 THROUGH CSM111)